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NUMERICAL EVALUATION OF THE WAKE-SURVEY EQUATIONS

FOR SUBSONIC FLOW INCLUDING THE EFFECT

OF ENERGY ADDITION

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#### NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

### ADVANCE RESTRICTED REPORT

NUMERICAL EVALUATION OF THE WAKE-SURVEY EQUATIONS

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#### SUMMARY

Direct-reading tables and charts are presented for determining the drag or thrust coefficients from wakesurvey measurements in the subsonic speed range. For flows wherein no energy is added, the point drag coefficient is shown to be an explicit function of the stream Mach number  $M_0$ , the static-pressure coefficient at the wake station  $P_1$ , and the total-pressure-loss coefficient  $\Delta H/q_0$ , where  $\Delta H$  is the total-pressure loss and  $q_0$  is the stream dynamic pressure. Values of the point drag coefficient are tabulated for a wide range of values of these parameters. Inasmuch as the tabulated coefficients (either drag or thrust) represent the point values, which are independent of the integration of the wake, the charts or tables in the form presented are general in application.

For flows wherein energy is added, such as flows behind propellers or heated radiators, an additional parameter, which is a function of the stagnation-temperature rise, must be considered. Values of the point drag coefficient that include the effects of the addition of energy are tabulated.

#### INTRODUCTION

In the field of aeronautical research, wake pressure surveys have been used increasingly for determining profile drag, internal drag, jet thrust, and related factors. The evaluation of the drag or thrust from pressure surveys is based on the solution of the momentum equation. For high-speed flows, in which the air must be considered compressible, the evaluation of the drag or thrust must include the variation in the density of the air; this variation in the density necessarily complicates the solution of the momentum equation. For flows wherein energy has been added, the density variation is of importance even for the low-speed conditions.

Because a large number of pressure readings are usually required to make a complete wake survey, it is essential that solutions of the wake-survey equations for all types of flow be presented in a form to permit rapid and accurate computation. Various simplifying techniques have been developed for evaluation of the drag coefficient from wake pressure measurements by means of charts or tables, but these methods have required excessive interpolation or computation due to the form of the parameters employed. When simplifying assumptions as to the shape of the wake profile and the constancy of the static pressure across the wake are made, integration techniques may be combined with the solution of the wakesurvey equations (references 1 and 2) to reduce greatly the required computation. This method has been found useful in determining the section profile-drag coefficients of airfoils; however, such a method is not general in application and is limited by the original assumptions of uniform static pressure and a definite wake profile.

The method found to be most generally applicable at the Langley 8-foot high-speed tunnel consists in the point-by-point evaluation of the drag coefficient by means of direct-reading charts with the final integration performed in a separate step. Facility of evaluation of the drag coefficient is provided in that the point-by-point values can be determined from simple The charts require no assumptions as to wake charts. profiles or pressure gradients. The flexibility required for application to both two- and three-dimensional flows is provided by performing the integration in a separate In this report the solution of the wake equations has been developed for both isoenergic flows (constant total energy) and flows wherein energy has been added: thus the equations are generally applicable to flows behind airfoils, propellers, and radiators. The fundamental principles may be applied to the determination of the thrust from jet units; however, consideration must be given to the change in the value of the ratio of specific heats and the momentum of the fuel.

#### SYMBOLS

```
speed of sound, feet per second
a
A
         area, square feet
         reference area, square feet
C^{\mathbb{D}}
         drag coefficient
         chord, feet
C
cd
         section profile-drag coefficient
c<sub>d</sub> '
         point drag coefficient
         specific heat at constant pressure (for air,
cp
            0.24 Btu/lb/OF)
D
         drag, pounds
         energy added, foot-pounds per second
E
         compressibility factor
\mathbf{F_c}
         acceleration of gravity (32.2 ft/sec2)
g
H
         total pressure, pounds per square foot
         total-pressure-loss coefficient
qo
         mechanical equivalent of heat (778 ft-lb/Btu)
J
         energy-input factor
K
         Mach number \left(\frac{V}{2}\right)
M
         mass flow rate, slugs per second (ρΑV)
m
         static pressure, pounds per square foot absolute
p
         static-pressure coefficient
P_1
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q dynamic pressu	re $\left(\frac{1}{2}\rho V^2\right)$
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T static temperature, OF absolute

T' stagnation temperature, OF absolute

 $\Delta T'$  stagnation-temperature rise,  $^{O}F$   $(T_1' - T_0')$ 

V velocity, feet per second

y distance across wake, feet

ρ density, slugs per cubic foot

γ ratio of specific heats (for air, 1.400)

Subscripts o, 1, and 2 refer to the flow stations designated in figure 1.

#### DISCUSSION OF THEORY AND METHODS

#### Fundamental Relations

The detailed solutions of the basic wake-survey equations for flows with and without the addition of energy have been presented in reference 3. A theoretical analysis of the basic assumptions required for the solution of the wake-survey equations for isoenergic flows has been presented in reference 4.

The basic form of the drag equation as derived from the momentum relation is

$$D = \int_{\text{wake}} \rho_1 V_1 (v_0 - v_2) dA_1$$

Then

$$\begin{split} c_{D} &= \frac{2}{A_{R}} \int_{wake}^{\rho_{1} v_{1}} \left(v_{o} - v_{2}\right) dA_{1} \\ &= \frac{2}{A_{R}} \int_{wake}^{\rho_{1} v_{1}} \left(1 - \frac{v_{2}}{v_{o}}\right) dA_{1} \\ &= \frac{2}{A_{R}} \int_{wake}^{\rho_{1} v_{1}} \left(\frac{1}{\rho_{o}} + \frac{v_{2}}{v_{o}}\right) dA_{1} \\ &= \frac{2}{A_{R}} \int_{wake}^{\rho_{1} v_{1}} \left(\frac{q_{1}}{\rho_{o}}\right)^{1/2} \left(\frac{\rho_{1}}{\rho_{o}}\right)^{1/2} \left(\frac{\rho_{2}}{\rho_{o}}\right)^{1/2} - \left(\frac{\rho_{2}}{\rho_{o}}\right)^{1/2} \frac{v_{2}}{v_{o}} dA_{1} \\ &= \frac{2}{A_{R}} \int_{wake}^{\rho_{1} v_{1}} \left(\frac{\rho_{1}}{\rho_{2}}\right)^{1/2} \left(\frac{q_{1}}{q_{0}}\right)^{1/2} \left(\frac{\rho_{2}}{\rho_{o}}\right)^{1/2} - \left(\frac{q_{2}}{q_{0}}\right)^{1/2} dA_{1} \end{split}$$

$$(1)$$

With the usual assumption that the total pressure at station 2 (where  $p_2 = p_0$ ) is equal to the total pressure at station 1, the drag coefficient can be evaluated as indicated in the appendix. The numerical solution of equation (1), however, is difficult and requires excessive computation for direct use.

For convenience in presenting and discussing the solution of equation (1) the point drag coefficient is defined as

$$c_{d'} = 2 \left(\frac{\rho_1}{\rho_2}\right)^{1/2} \left(\frac{q_1}{q_0}\right)^{1/2} \left[\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}\right]$$
 (2)

Negative values of the point drag coefficient indicate that the system is producing thrust.

# Isoenergic Flows

Isoenergic flows are found over aerodynamic bodies or through internal-flow systems when no appreciable change in stagnation temperature occurs. An analysis of the terms constituting the point drag coefficient for iscenergic flows indicates that the point drag coefficient is an explicit function of the free-stream Mach number Mo, the static-pressure coefficient in the and the total-pressure-loss coefficient  $\Delta H/q_0$ . The pressure coefficients used are merely an expression of the measured data in coefficient form. Because the values of these parameters can be easily determined from the test data, their use provides a convenient and direct method for evaluation of the point drag coefficient. explicit expression for the point arag coefficient in terms of the parameters  $M_0$ ,  $P_1$ , and  $\Delta H/q_0$ plicated; but the coefficient can be easily determined by usual methods for given values of these parameters (see appendix).

Values of the point drag coefficient for a wide range of values of pressure coefficients  $P_1$  and  $\Delta H/q_0$  and for given values of stream Mach number  $M_0$  are presented in table I. The range of total-pressure-loss coefficients has been extended into the negative region (which indicates a total-pressure increase) to permit evaluation under conditions of low energy input approaching isoenergic flow. The application of table I to flows wherein energy has been added will be discussed in a later section. A large range of positive values of  $P_1$  has been included to permit computation of internal duct flows for which the static-pressure coefficient may approach unity.

Other forms of the parameters and other methods of presentation than those used in table I may be more suitable for certain applications. Instead of Mach number Mo, the pressure ratios  $\frac{H_O-p_O}{p_O}, \frac{H_O-p_O}{H_O},$  or  $\frac{p_O}{H_O}$  may be used; the pressure coefficients P<sub>1</sub> and  $\frac{\Delta H}{q_O}$  may be replaced by  $\frac{p_1-p_O}{H_O-p_O}$  and  $\frac{H_O-H_1}{H_O-p_O},$ 

respectively. The two parameters  $\frac{p_1-p_0}{H_0-p_0}$  and  $\frac{H_0-H_1}{H_0-p_0}$  have an advantage when they can be determined directly from the measured wake and stream pressures. Tables and charts of the point drag coefficient for isoenergic flow in terms of these two parameters are presented in reference 5 with charts and tables for rapid evaluation by an approximate method.

### Flows Wherein Energy Is Added

The fundamental relations expressed in equation (1) are correct for the evaluation of the drag or thrust coefficient for flows wherein energy has been added, such as flows through radiators or propellers. The evaluation of the density ratio  $\rho_2/\rho_0$ , however, involves an added parameter that is a function of the energy input. The evaluation of the density ratio  $\rho_2/\rho_0$  as developed in appendix B of reference 3 becomes

$$\frac{\rho_2}{\rho_0} = \frac{1 + \frac{\gamma - 1}{2} M_0^2 \frac{q_2}{q_0}}{1 + \frac{\gamma - 1}{2} M_0^2 + \frac{E}{Jc_p gmT_0}}$$

In order to determine the drag coefficient from equation (1) for flow conditions wherein energy has been added, an additional parameter  $\frac{E}{Jc_{p}gmT_{o}}$  must be determined; this parameter is designated K.

An inspection of the terms constituting the point drag coefficient (equation (2)) indicates that the quantity  $2\left(\frac{\rho_1}{\rho_2}\right)^{1/2}\left(\frac{q_1}{q_0}\right)^{1/2}$  can be expressed as a function of  $M_0$ ,  $P_1$ , and  $\frac{\Delta H}{q_0}$ . The last part of equation (2)

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$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}$$
 can be expressed as a function of M<sub>o</sub>,  $\frac{\Delta H}{q_0}$ , and K. The two quantities of which the product is

the point drag coefficient can therefore be determined independently. Values of the two parts of equation (2) required to evaluate the point drag coefficient for flows wherein energy has been added are presented in tables II and III.

The value of the energy parameter K can be computed directly for flow conditions where the energy input and mass flow are measurable and are uniform across the survey plane. Such a condition might be attained for flow through an efficient radiator installation. For flow conditions as found behind a propeller, however, where the energy input and elemental mass flow are not uniform and are difficult to measure, an evaluation of the energy parameter can better be made by experimental methods.

The energy equation from the free stream to the wake station becomes

$$\frac{{v_o}^2}{2} + Jc_p gT_o + \frac{E}{m} = \frac{{v_1}^2}{2} + Jc_p gT_1$$

For stagnation conditions at the free-stream and wake stations, the energy parameter becomes

$$K = \frac{E}{J_{c_pgmT_o}}$$

$$= \frac{T_1' - T_o'}{T_o}$$

$$= \frac{\Delta T'}{T_o}$$

where the prime refers to the stagnation temperature. The energy parameter becomes, then, merely the ratio of the stagnation-temperature rise to the absolute stream static temperature.

Inasmuch as the stagnation-temperature difference  $T_1' - T_0'$  is small and must be measured accurately, stagnation-temperature thermocouples or resistance thermometers may be connected and calibrated to read the stagnation-temperature difference directly. The free-stream stagnation-temperature reference may be obtained by installing one of the temperature-measuring elements outside the wake. The determination of the absolute free-stream static temperature does not require great accuracy; therefore, conventional methods may be used.

#### USE OF TABLES AND CHARTS

The values presented in tables I to III for evaluation of the point drag coefficient are difficult to apply directly because interpolation is required. The tabulated values have therefore been plotted in the form of direct-reading charts.

# Isoenergic Flows

Examples of the types of plot found useful for evaluation of the point drag coefficient for isoenergic flow are presented in figure 2 for M<sub>o</sub> = 0.20 and 0.70, respectively. A representative cross plot of the tabular data of table I is presented in figure 3. It should be pointed out that the scale of the plots of point drag coefficient presented in this report is too small for accurate work. These plots have been included, however, to indicate the general form of the curves and to provide a basis for more accurate large-scale plots based on the original tabular data.

The charts in their present form are especially applicable for use in high-speed wind tunnels, where tests are usually run at fixed values of stream Mach number, and the value of the static-pressure and total-pressure-loss coefficients can be determined directly from the pressure records and wind-tunnel calibration. For use at values of stream Mach numbers different from those tabulated, the chart for the nearest tabulated Mach number may be used with but small error (\*12 percent). For greater accuracy at the intermediate Mach numbers a linear interpolation may be assumed (fig. 3).

The presentation of the tabulated results may be changed from that used in figure 2 to fit various conditions. For example, the ratio  $\frac{c_d!}{\Delta H/q_o}$  may be plotted

instead of  $c_d$ ' in order to attain greater accuracy on a small-scale plot. For many tests the wake static-pressure coefficient remains essentially constant with changes in  $M_O$ ; therefore the point drag coefficient may be plotted against  $\Delta H/q_O$  for a range of values of  $M_O$  at given values of  $P_1$ . The use of the tabulated data in this form may be applicable to flight tests.

# Flows Wherein Energy Is Added

In figure 4 are presented representative plots of tables II and III for evaluating the point drag coefficient for flows wherein energy is added. This figure shows that an increase in stagnation temperature of the order of 1° F (K = 0.002) can produce a significant effect on the evaluation of the drag coefficient at low values of  $\Delta H/q_0$ . For many cases, however, the energy effect may be considered negligible - as, for example, the flow behind a lightly loaded propeller - and the tabulated results for isoenergic flow (table I) may be used directly. A comparison of the numerical result for the experimental value of K with that for K = 0 indicates the magnitude of the energy effect and which form of the equations should be used. The iscenergic values of cd' presented in table I correspond to the  $c_d$ ' for K = 0 presented in tables II and III. values of Because the solutions of the equations for isoenergic flow can be determined from table I directly in one step, table I should be used whenever the energy effect can be neglected.

It should be pointed out that for flows wherein energy is added, certain combinations of the parameters  $M_0$ ,  $P_1$ ,  $\frac{\Delta H}{q_0}$ , and K specify supersonic flow in the

wake. For such flows the measurement of the wake pressures is subject to corrections that are dependent on the intensity of the shock at the local supersonic speed. No attempt has been made in this report to evaluate these shock corrections to the measured pressures.

# Integration Techniques

The evaluation of the total-drag coefficient involves the integration of the point-drag-coefficient profile. Inasmuch as the evaluation of cd' is independent of the integration process, great flexibility in the application of the proposed wake-survey techniques to various types of flow can be attained. The flows behind airfeils, within internal-flow systems, or through propellers may therefore be calculated by similar methods that vary only in the integration technique employed. The wake profile and the manner in which the wake is surveyed will determine the optimum type of integration.

With the values of the point drag coefficient determined from the charts or tables presented in this report, the value of the total-drag coefficient becomes

$$c_{D} = \frac{1}{A_{R}} \int_{wake}^{c_{d}} dA_{1}$$

where  $A_R$  is the reference area upon which  $C_D$  is to be based. The integration technique employed should be determined from considerations of expediency and the accuracy of integration desired.

For two-dimensional flows behind airfoils the airfoil-section drag coefficient becomes

$$c_d = \frac{1}{c} \int_{\text{wake}}^{c_d} c_d! dy$$

where the integral represents the area under the  $c_d$ '-curve when expressed in the same units as the chord  $c_{\bullet}$ 

#### NUMERICAL EXAMPLES

#### Isoenergic Flow

Assume the following set of conditions, which might correspond to the center of the wake of an airfoil at high speed:

$$M_0 = 0.70$$
 $P_1 = 0.12$ 
 $\frac{\Delta H}{q_0} = 0.3$ 

An interpolation of table I or the use of the largescale plots of table I (fig. 2(b), for example) gives the value of the point drag coefficient cd' of 0.1964. The resulting positive value indicates drag.

# Flows Wherein Energy Is Added

(1) Assume the following set of conditions, which might correspond to the flow behind a highly loaded propeller for the climb condition:

$$M_0 = 0.20$$
 $P_1 = 0.25$ 
 $\frac{\Delta H}{q_0} = -0.50$ 

(The negative value of  $\Delta H/q_0$  indicates total-pressure increase.)

$$\Delta T' = 4.0^{\circ}$$
 $T_o = 500^{\circ}$  F abs.

From the value of absolute static temperature and stagnation-temperature rise,

$$\kappa = \frac{4.0}{500}$$
$$= 0.008$$

An interpolation of tables II and III or the use of the large-scale plots of tables II and III (fig. 4, for example)

gives the following values of the parts of equation (2) that constitute the point drag coefficient:

$$2\left(\frac{\rho_1}{\rho_2}\right)^{1/2}\left(\frac{q_1}{q_0}\right)^{1/2} = 2.2368$$

$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2} = -0.2217$$

The value of the point drag coefficient thus becomes

$$c_{d}' = (2.2368)(-0.2217)$$

The resulting negative value indicates thrust.

(2) Assume the following set of conditions, which might correspond to the flow behind a radiator measured near the duct outlet:

$$H = \frac{E}{J}$$
  
= 600 Btu/sec  
gm = 30 lb/sec  
 $T_0 = 430^{\circ}$  F abs.  
 $M_0 = 0.7$   
 $\frac{\Delta H}{Q_0} = 0.4$   
 $P_1 = 0.1$ 

Now

$$K = \frac{E}{Jc_{p}gmT_{o}}$$

$$= \frac{600}{0.24 \times 30 \times 430}$$

$$= 0.194$$

From these values of the parameters,

$$2\left(\frac{p_1}{p_2}\right)^{1/2}\left(\frac{q_1}{q_0}\right)^{1/2} = 1.5503$$

and

$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2} = 0.0889$$

The value of the point drag coefficient thus becomes

$$c_{d}' = 1.5503 \times 0.0889$$
  
= 0.1378

The resulting positive value indicates drag.

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#### APPENDIX

#### EQUATIONS FOR COMPUTING TABULATED VALUES

# OF DRAG COEFFICIENT

The various terms of the drag-coefficient equation (equation (1)) can be expressed by the following relations. It should be pointed out that by definition  $p_2 = p_0$  and by assumption  $H_1 = H_2$ . (A detailed derivation of the following terms can be found in appendix B of reference 3.)

Where K is the energy-input factor. (For isoenergic flow K = 0.)

For the use of the equations in the foregoing form, some initial value of one of the pressures must be assumed. Because the values of the various parts of equation (2) are determined fundamentally by the pressure ratio, the numerical value of a pressure has no significance; therefore any positive value may be used.

The detailed computing procedure is as follows:

- (1) Given  $N_0$  and a range of values of  $P_1$  and  $\Delta H/q_0$ . (The value of K must also be considered for flows wherein energy is added.)
- (2) In order to determine the free-stream pressures corresponding to  $M_0$ , assume any convenient value of stream static pressure  $p_0$  (1000 lb/sq ft is assumed here) and use the following relations:

$$q_o = \frac{r}{2}p_o n_o^2$$

$$H_0 = P_0 + F_{\dot{c}_0} q_0$$

where  $\gamma = 1.400$  and  $\Gamma_{\rm C_O}$  is obtained from table IV for the value of  $\rm M_O$ . The value of  $\rm H_O$  can also be determined for given values of  $\rm M_O$  and  $\rm p_O$  from the following equation:

$$M^{2} = \frac{2}{\gamma - 1} \left[ \left( \frac{\underline{H}}{\underline{p}} \right)^{\frac{\gamma - 1}{\gamma}} - 1 \right]$$

(3) The step-by-step computing procedure is as follows:

. ...... .....

Column	Relation
(1)	$\Delta H = \frac{\Delta H}{q_0} q_0.$
(2)	$\Delta p = P_1 q_0$
- (3)	$p_1 = p_0 + \Delta p = p_0 + (2)$
(4)	$H_1 = H_2 = H_0 - \Delta H_0 = H_0 - (1)$
(5)	$H_1 - p_1 = (4) - (3)$
(6)	$\frac{H_1 - p_1}{p_1} = \frac{(5)}{(3)}$
(7)	$F_{c_{1}}$ from a plot of table IV for value of (6)
(8)	$H_2 - p_2 = (4) - p_0$
(9)	$\frac{H_2 - p_2}{p_2} = \frac{(8)}{p_0}$
(10)	$F_{c_2}$ from a plot of table IV for a value of (9)
(11)	$\frac{q_2}{q_0} = \frac{1}{q_0} \frac{H_2 - p_2}{F_{c2}} = \frac{1}{q_0} \frac{(8)}{(10)}$
(12)	$\left(\frac{q_2}{q_0}\right)^{1/2} = (11)^{1/2}$
(13)	$\frac{q_1}{q_0} = \frac{1}{q_0} \frac{H_1 - p_1}{F_{c_1}} = \frac{1}{q_0} \frac{(5)}{(7)}$ $\left(\frac{q_1}{q_0}\right)^{1/2} = (13)^{1/2}$
(1/4)	$\left(\frac{q_1}{q_0}\right)^{1/2} = (13)^{1/2}$

Column	Relation
(15)	$\left(\frac{p_1}{p_0}\right)^{1/2\gamma} = \left(\frac{(3)}{p_0}\right)^{0.357}$
(16)	$\frac{\rho_2}{\rho_0} = \frac{1 + \frac{\gamma - 1}{2} N_0^2 \frac{q_2}{q_0}}{1 + \frac{\gamma - 1}{2} N_0^2 + K}$
	$= \frac{1 + 0.2M_0^2 (11)}{1 + 0.2M_0^2 + K}$
	For isoenergic flow, K = 0
(17)	$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} = (16)^{1/2}$
(18)	$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2} = (17) - (12)$
(19)	$c_{d'} = 2 \left(\frac{p_1}{p_0}\right)^{1/2\gamma} \left(\frac{q_1}{q_0}\right)^{1/2} \left[\left(\frac{p_2}{p_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}\right]$
	= 2 × (15) × (14) × (18)

(4) The numerical solutions of the various steps of the computing equations are given for the following examples:

	<u> </u>		
M <sub>O</sub> P <sub>1</sub> ΔΗ/q <sub>O</sub> K	0.70 0.12 0.3	0.2 0.25 -0.5 0.008	0.70 0.1 0.4 0.194
p q <sub>o</sub> H <sub>o</sub>	1000 343 1387.034	1000 28 1028.28	1000 343 1387.084
(1) (2) (4) (5) (6) (7) (10) (12) (12) (13) (14) (15) (17) (19) (19)	102.9 41.16 1041.16 1284.134 0.2334 0.2334 0.2842 0.2842 0.2860 0.7560 0.8695 0.8695 0.8695 0.8695 0.9890 0.195 0.1964	-14 7 1007 1042.28 35.28 0.0350 1.023 1.0150 1.1377 1.21976 1.1156 1.0025 0.9980 -0.4959	137.2 34.3 1034.884 1249.584 215.084 0.071884 0.071884 0.0824994 0.6715 0.67194 0.76591 0.8249 0.90839 0.0378

Any arbitrary value of  $p_0$  may be assumed without change in the final result. For these examples  $p_0 = 1000$  has been used.

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TABLE I POINT DRAG COEFFICIENT FOR ISCENERGIC FLOW

,						: •	<b>x</b> o = 0	]		•	e	. •		-
AH qo	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .500 .500 .250 .250 .075 .055 .025 .075 .025 .025 .025 .025 .025 .020 .250 .200 .250 .25	89978 699788 6997888 699788 699788 699788 699788 699788 699788 699788 69	0.8914 -7876 -5770 -57710 -5772 -15736 -23762 -23762 -17926 -0598 -0598 -0598 -12150 -37710 -4276 -7784 -778	0.8568 -555826 -555826 -34271 -1719 -08288 -0529 -0529 -0829 -124032 -124032 -124034 -	0.8200 .7292 .5402 .43948 .28164 .1148 .0862 .0576 0 .9984 .0878 .11770 .23766 .11770 .23766 .23602 .48148 .7398 .8700 .10018	0 - 7818 - 7008 - 6150 - 52466 - 27466 - 27466 - 27466 - 27466 - 27466 - 27466 - 27566 - 27566	0.74-16 -67-08 -592-6 -592-6 -31-84 -26-64 -11-00 -08-28 -11-10 -08-28 -11-10 -08-28 -11-10 -28-9 -11-10 -28-9 -11-10 -11	0.69926 .63946 .4932 .49032 .49032 .2112 .1598 .1076 .0810 .05142 .05832 .116808 .22542 .16808 .225434344 .16808 .28434 .16808 .28434 .16808 .28434 .16808 .28434 .16808 .28434 .16808 .28434 .16808 .28434 .16808 .28434 .16808 .16808 .16808	0.65468 .6068 .5450 .37904 .3012 .2058 .1052 .0792 .05308 0 .0268 0 .05642 05142 1648 2792 37762 7008 5772 7008 9540	0.6056 5722 5198 4578 25198 4577 2922 -2170 0206 1026 0774 0518 0262 0 -0264 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0 - 55282 - 53522 - 19344 - 3828 - 23628 - 1948 - 1948 - 1948 - 19506 - 05566 0 - 0558 - 07806 - 1044 - 2692 - 2692 - 4412 - 5596 - 8038 - 9292	0	0. 14282 14348 293444 22348 293434 22348 20149 2	0.3496 4026 37188 25730 21768 21768 06994 04256 0238 20726 2	0 24724 245076 3462422 255076 36462422 26708
AH q <sub>o</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0,80	
0.800 .700 .500 .500 .500 .500 .200 .150 .025 .025 .025 .025 .025 .025 .025 .0	0.2860 .32868 .285286 .285108 .1988 .0858 .0858 .0858 .0858 .0432 0 .0432 .025668 .0432 .025668 .0432 .025668 .0432 .025668 .0432 .0358 .0432 .0458 .0	. 2846 .2950 .2966 .2190 .1896 .1208 .0828 .0424 .0214 .0218 .0428 .0424 .05214	0872 1334 1810 2300 2804 3842 4922 6040	- 1294 - 1760 - 2238 - 2734 - 3756 - 4820 - 5924 - 7060	0 - 0196 - 0398 - 060l - 0816 - 1252 - 1706 - 2176 - 2660 - 366l - 4714 - 6802	16468 1258 09868 05258 05258 0182 0188 -0582 -07866 -1210 -21106 -25772 -46692	0756 1166 1596 2508 3476 4494 5556	0.1008 1.264 1.1198 1.056 0.0668 0.0168 0.0168 0.0169 0.0170 0.0524 1.1208 1.19748 1.1	- 0690 - 1072 - 1478 - 1902 - 2346 - 3278 - 4298 - 5272	0.0730 0.0818 0.0818 0.0514 0.010 0.0152 -0.0152 -0.0152 -1.1828 -2.260 -3.174 -5.164 -5.164 -5.164 -7.326	0618 0970 1350 1750 2172 3066 4020 5026 6076	0472 0494 0398 0320 0218 0 - 0130 - 0270 - 0420 - 0578 - 1280 - 1280	0248 0386 0534 0856 1284 1982 2838 4738 4738 5764	

TABLE I - Continued

[Mo = 0.05]

AH Qo	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 -7000 -5000 -4000 -2500 -2500 -0750 -0500 -1500 -1500 -2500 -1500 -1500 -1500 -1500 -1500 -1500 -1500 -1500 -1500 -1500	8976628 8976628 17740 17740 18061128 17740 18061128 1806112	- 0604 - 0908 - 1214 - 1830 - 2146 - 3074 - 3696 - 4970 - 62566 - 8888	0594 0894 1194 1802 2410 3039 3644 4902 6180	812588646246276 8125886476246276 6553886476246276 655388647748579 6000 60	- 0562 - 01542 - 1174346 - 237364 - 2771646 - 2771646 - 28574	6698 59148 59148 51480 67622 116008 60527 60 008148 11712 12880 11712	6368 90 4468 40 4168 4	664428 644789 644789 654789 65478 65578 65778 65778 65778 65778 65778 65778 65778 65778 65778 65778 65	9286602668 5157626602768 157626602768 15762	- 0518 - 0786 - 1586 - 1586 - 2240 - 245840 - 55792 - 802	0248 0 02562 - 05662 - 10588 - 15588 - 26888 - 3184 - 43292 - 6686	- 0494 - 0744 - 1000 - 1518 - 2044 - 2584 - 3126 - 3248 - 5482 - 7799	0.35024 40244 36984 2530 217668 11768 06964 11768 00236 00236 00236 00236 00236 00236 00236 00236 0036 00	- 0468 - 07528 - 09548 - 1954 - 24988 - 249881 - 249881 - 25360

										<del></del>	,		
P <sub>1</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
40 V													
0.800													
700	0.2864												
.600	.3288	0.2848		0.1650									
.500	320L	.2926	.2618		0.1854	0 _ , ,							
.400	.2848	.2664	-2468	•225 <u>4</u>	.2016			0.1012					
.300	.2310	.2192	.2066	1934	.1790	1634	.1462		0.1036			}	
.250	.1984	.1892	.1796	.1694	.1584	.1466	.1340	.1198			0.0602		
.200	.1636	.1566	.1494	.1416	.1336	.1250	.1158	.1056	.0946	.0820		0.0474	
.150	.1254	.1206	.1154	.1100	·1044	.09,84	.0920	.0852		.0696			0.0350
.100		.0828	.0796	.0762		.0688	.0650	.0608		.0514			.0326
.075	.0650	.0628	.0604	.0580	.0554	.0526		.0468	.0436	.0400	.0362		.0270
.050	.0438	.0424	8040 <b>.</b>	0392	.0376	•0358		.0320		.0278			.0196
.025	.0222	.0214	.0208	.0200	.0192	.0182		.0164		.0144	.0132		.0106
0	0	0	lo	0	0	0	0	0	0	0	0	10	0
025	0226	0218	0212	0204	0196	0188	0180	0170	0162	0152	0142	0130	0118
050	0456	0442	ໄ ∧ໄ∙ລΩ	1 _ 01:31:	I N2QX	I 🗕 . 03X2 I	L D366	0350	0332	0312	0292	0270	0549
075	10688	10668	0648	0626	0604	0582	0558	0534	0508 0690	0489	10450	0420	0206
100	0926	0900			0816	0788	0756	0724	0690	0656	0618	0578	10226
150	1	1 - 137	155):	129L	1254	1212	1168	1122	1074	J- • 102 <u>4</u>	0972	10916	10056
200	Anor.	- 1858	1808	11758	1706	1652	1596	1558	1478	J-• <del>!\}\</del>	1350	1280	1200
250	1 ລ1.⊤Ω	- 2220	2300	I _ 223H	1 - 2176	121101	20hh	11974	1902	11828	1752	1670	11284
300	2932	2866	2796	2726 3748	J2654	2578	2502	2422	- • < 340	2250	2168	2076	1350
400	4006	3920	3836	3748	3656	3564	13470	<b>~•</b> 3372	3272	12168	- 3060	12828	2024
500		5016	_:₽03Jı	14812	I <b> a</b> 11706	I <b></b> 4590:	- 44400	142/4	<b>-</b> •4250	14130	4014		- • 2 720
600		[ 293.1.	6020	L 5012	1 - 5701	1 5672	I = . 55/L8	15020	1 - 4 5290	15458	5018	140/0	- 4732
700		<b></b> 7304	7776	701.8	arpa _ i	1		1 6500	1 - D 2 D 1	0210	6068	12910	- • 7 (30)
	8622	8488	8350	8210	8068	7922	7776	7624	7472	7314	/154	6990	0022
1						•							

NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

TABLE I - Continued

	[N <sub>0</sub> = 0.10]													
P1	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700 .500 .500 .200 .250 .250 .075 .050 .025 .050 .075 .150 .025 .050 .075 .075		- 0602 - 0902 - 1204 - 1814 - 2428 - 3648 - 3670 - 4932 - 6210 - 8810	7.5538866865020 7.5538866865020 7.5538866865020 7.5538866865020 7.558866865020 7.558866865020 7.558866865020 7.558866865020	- 0582 - 0872 - 01166 - 1758 - 295668 - 356668 - 67520 - 8606	-0570 -0856 -1146 -1728 -2516 -2912 -3512 -4730 -5728 -8502	68942 58942 11628 11628 10822 005276 008252 0092 108252 0092 1	- 0550 - 0826 - 1104 - 1668 - 2238 - 2818 - 3402 - 4592 - 7038 - 8290	0810 1084 1638 2198 2770 3346 4520 5720 6940	5700 -5174 -5750 -2750 -2452 -2950 -1020 -0766 -0762 -1068 -1068 -27208 -1068 -27208 -27208 -3248 -3448	0516 0776 1040 1574 2670 32574 45746 5742	1931476868284 1941176868284 1941177868284 1961147407866 1961147407866 1961147407866 1961147407866 1961147407866	- 0492 - 0742 - 0994 - 1508 - 2566 - 3108 - 4222 - 6538 - 7734	10182 10182 10182 10182 10182 10182 10182 10183 10182	

P <sub>1</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
0.800 .700 .500 .400 .300 .250 .075 .025 .025 .025 .025 .025 .025 .025 .02	- 0454 - 0686 - 0920 - 1402 - 1894 - 2402 - 2918 - 3984 - 50816	2854 2926 21886 21886 1560 1202 0626 0422 0216 0 0216 0 0216 13686 12852 13986 12852 13986 16966 17966	2060 1790 11488 1152 0602 0406 0 02106 0 02166 0 02166 0 08688 117986 11	0.1676 .2272.1928 .19288 .1498 .05578 .05578 .0599 .0019 .00198	2018 1786 1580 1532 1014 07552 0374 0190 0 0194 - 0398 - 06014 - 1248 - 1248	1750 16324 1246 0686 0526 0526 0526 0526 0526 0526 0526 052	1462 1338 11538 10922 0648 0340 0172 0 0178 6 0754 1162 12034 1466 1534 1466 1534 1466 1556 1666 1666 1666 1666 1666 166	0.1028 .1279 .11798 .0854 .0854 .0854 .0320 .0164 .0350 .0170 .0170 .0752 .1116 .1536 .2414 .3554 .35592 .56172	1040 0780 0780 0562 0360 0154 0 0162 - 0532 - 05688 - 1072 - 1894 - 12534 - 1258 - 1258 - 1258 - 1258 - 1258	0.0744 .08540 .08540 .08598 .05140 .02784 .01522 .0380 .05542 .06542 .1020 .18200 .23156 .14120 .5136	.0254 .0132 00140 0292 0450 0616 0968 1744 2162 3050	0482 0498 0320 0228 0120 0 0170 0 0170 - 0276 - 09120 - 1276 - 1276 - 22940 - 2872 - 2872 - 4888	021986 011886 011886 011886 011886 011886 011886 011986 01986 01986 01986 01986 01986 01986 01986 01986 01986 0198

WATIONAL ADVISORY COMMITTEE FOR AEROWAUTICS

TABLE I - Continued

[Mo = 0.15]

P P		Γ	<del></del>				T	r						
AH Qo	-0.50	-0.45	-0.4 <b>0</b>	-0.35	-0.30	<b>-0.2</b> 5	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700 .500 .300 .250 .250 .100 .075 .050 .075 .050 .075 .200 .370 .250 .300 .300 .300 .300 .300 .300 .300 .3	0.9112 .7966 .6860 .5760 .4644 .2938 .2362 .1192 .0894 .0500 0 .0602 .0902 .0902 .1818 .2434 .30578 .4938 .4938 .4938 .4938 .4938 .4938 .4938 .4938 .4938 .4938 .4938 .4938	0.8788 .7718 .6674 .5614 .5142 .2316 .2316 .1140 .0586 .0294 .1170 .0892 .0892 .1290 .1792 .2400 .3628 .4866 .6124	0.8450 .7462 .6476 .5470 .4434 .3368 .2820 .2272 .1170 .0862 .0576 .0290 0 .0590 .0582 .1182 .1764 .2364 .2366 .2966 .4802 .4968 .49	0.8096 -7174 -5324 -5324 -5324 -5224 -5224 -0864 -0284 -0572 -0864 -1738 -2324 -23526 -4758 -72488 -72488 -72488 -72488	0.7726 6966 6966 4212 32176 16166 08356 00556 00556 00556 1708 1708 - 2878 - 1708 - 2878 - 2878 - 4672 - 58358 - 9658	0 7328 6685046 580046 580046 16082 16081444 16082 0 0 20752 168544 0 0 20752 168544 168544 16804 16804 16804 16805 1680	0.6926 .6926 .5624 .3976 .3976 .2568 .2568 .2568 .0796 .0538	.6010 .53690 .4852 .2970 .25026 .1036 .0778 .0522 .00262 .1082 .16206 .21740 .2	0.6018 .5672 .5144 .3724 .2432 .1974 .1496 .0760 .0256 0.0256 0.0256 .0786 .1596 .1596 .2695 .2695 .4396 .57972 .79204	53848 43848 4359942 19488 279962 19488 007498 0057700 005700 005	0.4940 4928 44612 247088 1862 14152 22882 14160 0724 0024 1484 09724 00498 10498 105056 11526 11	0.4300 .1501 .1501 .38908 .2812 .1804 .0704 .0704 .0023 .004	0.3544 -4044 -40676 -3158 -25134 -1744 -1334 -0684 -0482 -0478 -0478 -0979 -1460 -1970 -1460 -1970 -1460 -1970 -19	0.2572 .35666 .31998 .23052 .2052 .16888 .2056646 .004666 .004
AH qo	0.20	0.25	0.30	0.35	0.140	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80	
0.800 .700 .500 .100 .250 .250 .250 .075 .050 .075 .050 .075 .150 .250 .250 .250 .250 .250 .250 .250 .2	0.0816 .2902 .3288 .3194 .2830 .2290 .1966 .1618 .0843 .0218 0.0218 0.0218 0.03	0 2106 2862 2962 21650 21676 11876 11876 11920 06418 06418 0 2166 06418 0 2166 0 2166	0.0668 .23622 .2160 .2052 .1782 .1180 .0788 .0404 .0208 .0204 .0208 .0322 .0427 .042	0.1712 2282 2250 1922 1682 11496 1075 1496 03788 0196 00462 1478 00462 00	0.05144 .2018 .1776 .1326 .17326 .17326 .0738 .0742 .0190 0 -0394 .03794 .0394	0.1366 .1756 .1650 .1650 .1650 .0552 .0552 .0186 .0552 .0186 .0576 .1654 .1654 .1654 .1654 .1654 .1654 .1654 .1659	14352864482 143528649337 764955554 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0346]	.105448 .1054480 .07563484 .07563984 .01530 .01530 .01530 .01530 .01530 .01530 .01530 .01530 .01530 .01530 .01530 .01530 .01530	1402 1810 2238 3134 4088 5092 6134	.0678 .0460 .0460 .0362 .0252 .0132 0 -0140 -0450 -0564 -1734 -1734 -33968 -4958	.0502 .0400 .0320 .0226 .0120 0 0130 0580 0580 1272 1656 2062 2922 3844 48820	0.0156 .0364 .0350 .0272 .0198 .0106 0 -0118 -0246 -0588 -0852 -1272 -1968 -1572 -1968 -25716 -4682 -5688	

TABLE I - Continued

 $\left[\mathbf{x_o} = 0.2\overline{0}\right]$ 

型 q <sub>o</sub>	<del>-</del> 0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700 .500 .200 .150 .250 .150 .025 .025 .100 .250 .150 .250 .250 .250 .250 .250 .250 .250 .2	0.9006 -7870 -67644 -5684 -2326 -1752 -1170 -0880 -0294 -0592 -0890 -1788 -2392 -1788 -2392 -16084 -73642 -73664 -9876	0.8690 .7628 .5594 .4480 .3394 .2842 .1720 .0866 .0582 .0288 0.0288 0.0292 .0872 .1762 .2958 .37564 .4782 .7252 .7252 .9778	7380 6402 5402 4376 3322 2784 2240 0850 0850 0850 09568 0286 - 0574 - 0862 - 1154 - 1154	.6206 .5256 .4270 .3250 .2726 .2194 .1656 .1110 .0836	0 7660 6850 6850 5104 14160 3174 21148 1682 1088 0820 0514 0 0274 11168 - 0554 11682 - 2254 - 2832 - 3459 - 7900 - 9468	0.7276 6568 -5768 -1948 -3096 -2100 -1588 -0804 -05370 0 -08298 -2788 -37560 -2714 -2718 -37560 -375	•55786 •47930 •35942 •2554 •075264 •075264 •075264	0.6446 .5962 .59618 .3810 .2958 .2002 .1020 .0770 .0258 .0	0 598342 596343 596363 196852 196852 19582 0075952 00557 00557 005773866 151062	142666 142666 142666 1426 1426 1426 1426	.4070 .34674 .2266 .18424 .1948 .07180 .0249	0 1496 1496 1496 3868 2580 2580 21786 17864 1922 0698 0234 0 0240 - 07274 - 1988 - 2508 - 1988 - 1988 - 1988 - 2528 - 1988 - 2528 - 252	0468 0708 0950 1442 1944	0.2644 .35532 .3658 .3126 .2978 .2036 .1668 .1668 .0658 .0458 .0458 .0222 0 .0226 .06908 .1408 .1908
AH qo	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80	
700 6000 700 70	2272 1952 16034 06346 0428 0216 0 0220 109748 109748 123594 123594 14964	3816 4872 5956	1766 2242 2730 3736 4778	.0384 .0194 0 .0200 .0104 .0614 .0828 .1264 .1718 .2184 .26654	0390 0592 0800 1226 1668 2126	0 0186 0376 0572 0772 1186 2064 25482	1464 1434 11484 1642 1642 1642 1642 1642 1642 1643 1643 1644 1644 1644 1644 1644 1644	1936 2378 3300 4272	0326 0500 0680 1056 1452 1868 2360 3206 4162 5158	.0870 .0828 .0700 .0512 .0406 .0142 .01508 -0474 -1068 -1392 -1796 -12218 -3108 -3108 -3108 -3108	- 2134 - 3006 - 3932 - 4904 - 5924	0.0262 .0508 .0508 .0508 .0508 .0508 .0508 .0266 .0120 0.0130 .0268 .0166 .0574 .0906 .1264 .2900 .3812 .4772 .5780 .6816		

TABLE I - Continued

 $M_0 = 0.25$ 

						Γα	·	<u> </u>						<del></del>
<u>Ан</u> <del>дн</del>	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 0	75588 75588	7516 64964 44030988 65404 2268682 87004 1185682 0 05681 128982 1458 128982 1458 1458	.0278 0 .0280 .0562 .0846 .1126 .1700 .2276 .3438 .4616 .5804 .5804 .8214	7026 61208 4202 32154 10208 22154 10208 05474 10208 05474 0 05574 11742	0 - 0272 - 05144 - 0820 - 1092 - 1648 - 2210 - 2774 - 3349 - 5664 - 68141 - 8038	64720 54882 54882 54884 54884 5488 5	6250666668282501868825018688250186882501868825018688250186882501868825018688250186854660668546606685466666854666666854666666854666666854666666685466666666	5908620 590862	5505408 5505408 5505408 5505408 19508 1990	00216 0191 0746 0996 1510 2030 2558 4182 5290	1878762 1878762 1878762 1878762 1878762 1878762 1878762 1878762 19798 197982 197982 197982 197982 197982 197982 197982 197982 197982 197982 197982 197982 197982 197982 197982 197982 197982 197982 19	1480 14874466 1784668 1784106 1784106 1784106 1784106 188410 1884	1914 2416 2928 3976 5050	548 548 5495 5495 522 522 522 522 522 522 523 523

Р <sub>1</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
0.800 .700 .600 .100 .300 .250 .250 .150 .075 .025 0	.3290 .3192 .2248 .1932 .1218 .0828 .0212 .0212 .0212 .0212 .0213 .0212 .0213	0.2240 .2840 .2910 .2622 .2140 .1846 .1572 .0802 .0410 .0206 0 .0212 .0410 .0212 .0410	- 2440 - 26440 - 26440 - 1756 - 1454 - 1758 - 0589 - 0589 - 0412 - 0626 - 08166 - 17408 - 2688 - 3670 - 1752	0.1824 .2240 .1962 .13662 .13662 .13662 .13662 .0562 .0562 .01908	.1926 .2026 .1768 .1508 .1508 .0798 .05466 .0184 .0798 .0188 .0798 .0188 .0792 .1218 .20560 .3588 .358	0.14574 .16742.1252 .12520 .06744 .05506 .05766 .057662 -117938 -241349 -3414349	1484 1464 11344 10918 10938 10038 10	.0460 .0316 .0160 00168 0340 0522 0704 1912 2348 4214 5208	.1076 .1056 .0948 .0776 .0430 .0296 .0150 00158 0498 0672 1046	0.0812 .0884 .0832 .0700 .0512 .0276 .0142 00150 0306 1380 1778 2194 3072 3978	.0696 .0614 .0460 .0362 .0130 0 -01484 -0606 -07318 -1706 -21712 -3886 -4844	0.0326 .0524 .0404 .0328 .0118 0 - 0130 0268 0916 0568 0956 1632 2872 2872 24716	.0278 .0200 .0106 0 -0118 -0216 -0528 -0816 -11552 -1938 -2764 -3618 -4582

NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

TABLE I - Continued

						[Mo :	= 0.30]							
AH P1	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700 .500 .400 .250 .250 .100 .075 .05 .055 0 055 0 075 150 250 250 300 400 500 500	7604 -6542 -5470 -22730 -22730 -22730 -2284 -05650 -08550 -17028 -17	1118 1678 2250 2816 3390 4542	192 -6192 -6192 -6192 -5216 -4216 -3192 -21508 -1082 -0542 -0542 -0542 -0642 -0642 -1656 -2218 -2788 -33488 -34480	0.77912 6012 5012 5128 21128 2110 1580 0.537 0.704 -0532 0.705 -0532 0.704 -0532 0.704 -0532 0.704 -0532 -16	-5824 -4944 -4905 -3058 -2568 -2568 -1054 -0786 -0266 -0532 -0800 -1608 -1608 -2158 -25512	- 5628 - 4798 - 3916 - 2988 - 2536 - 2026 - 1024 - 0772 - 0514 - 0260 0 - 0262 - 1758 - 2126 - 2666 - 3216 - 3216 - 3216 - 1324 - 5146	- 124 - 5424 - 54648 - 38194 - 2914 - 1980 - 1496 - 1004 - 0756 - 0256 - 0256 - 0774 - 1558 - 2626 - 31706 - 31706 - 53770	-5252 -5252	55288 19302 25562 1888 11962 19624 0 0484 0 0494 0 0494 1505 1005 1005 1005 1005 1005 1005 100	- 14750 - 3462 - 2268 - 1840 - 0798 - 0476 - 0798 - 0486 - 0798 - 0798 - 0798 - 1478 - 1478 - 1576 - 1478 -	.4894 .49980 .39980 .25998 .25998 .0916 .0916 .0236 .04692 .0474 .04716	- 4462 - 42792 - 379206 - 25100 - 21322 - 1736 - 1736 - 10450 - 0252 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	10346 139492 139492 139492 14682 128686 108656 10922 109494 109492 10949	356248 356248 252846 1626 1626 1626 1626 1626 1626 1626 16

AH P1	0.20	0.25	0.30	0.35	о.40	0.45	0.50	0.55	0:60	0.65	0.70	0.75	0.80
0.800 .700 .600 .900 .400 .200 .150 .075 .050 .025 -055 055 055 150 250 250 250 250 200 200	.3288 .3148 .3148 .2768 .2268 .1908 .0818 .0618 .0618 .0212 	2900 29004 21626 11526 11526 11580 0590 0590 000 000 000 000 000 1284 1282 12702 127	.2624 .2426 .2006 .1740 .1440 .1112 .0760 .0578 .0388	0.1888 .28314 .18848 .16493 .10752 .0075742 .0075742 .007594	2022 17650 1550 15101 0700 0532 00184 00188 -0780 -1192 -20780 -1192 -25456 -4430 -5464 -5464	0.1766 .17840 .1620 .1444 .1222 .0960 .0666 .0594 .0176 0.0182 -0754 -1156 -2452 -2452 -23338 -5352	1506 11460 11370 11470 10902 10630 10484 10167 10752 10752 10772 10726 1116 10726 1116 1116 1116 1116 1116 1116 1116 1	0-1162 1292 12048 .0848 .0594 .0456 .0310 .0160 0 -0358 -0516 -0698 -1076 -1888 -2316 -1414 -5114 -5114	1092 1062 0950 07752 0429 0152 0152 - 0492 - 0492 - 0668 - 1042 - 1824 - 2120 - 1942 - 2120 - 1992 - 5990 - 5990	0.0842 .0896 .0838 .0700 .0510 .0272 .0142 0.0150 .0636 .0936 .1368 .3028 .3028 .3938 .4887 .5870	0692 .0708 .0460 .0460 .0250 .0132 0 .0140 -0602 -0602 -1588 -2084 -2934 -3836 -3836 -57740	.0522 .0406 .0324 .0226 .0120 0 -0130 -0268 -0414 -0566 -2834 -3718 -3718 -3644	0.0306 .0402 .0304 .0280 .0200 .0108 0 .0118 0528 0810 1538 1538 1538 2732 3602 5470

TABLE I - Continued

 $\left[\mathbf{M}_{o}=0.35\right]$ 

								_				_		
AH Qo	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700 .600 .500 .400 .300 .250 .150 .025 .025 .025 .025 .025 .025 .025 .0	0.8518 743840 53490 21628 10818 10818 10819 1081	0 0270 0538 0812 1082 1632 2166 2728 3282	0.7952 .7002 .6060 .51106 .3116 .2696 .0528 .0266 0.0266 0.0266 .0532 -0800 .0266 -16140 -21494 -3454 -7698 -8820	6789726 549026 44005554 2005554 1505788 1005050 100505 100		0.6992 .6288 .5524 .4704 .2922 .2952 .1492 .0554 .0752 .0554 .0762 .0762 .1044 .25902 .31194 .525902 .5367	25630 48 25630 48 25630 48 25630 49 26630 48 26630 48 266	-1414 -3626 -2782 -23490 -1432 -09482 -0244 -0722 -0244 -07489 -1490 -1496 -1496 -1251	0482 0726 0970 1470 1960 2476 2992 4030 5084	.4696 .3490 .2632 .2832 .1800 .1366 .0952 .0464 .0234 .0234 .0472 -14428 -14428 -24376 -23946			.1030 .39162 .355282 .2026 .1652 .10652 .08544 .0218 .0218 .0470 .0218 .0470	0.2924 .3576 .3616 .3384 .2292 .1954 .1222 .0832 .0626 .0420 .0212 0 0 0216 -0432 -17386 -2268 -2750 -37750 -4774 -58682 -7858
AH P1	0.20	0.25	0.30	0.35	0.10	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0,80	
0.800 .700 .500 .500 .300 .150 .150 .055 .055 .055 .055 .055 .0	0.1814 .3048 .31266 .31266 .31266 .1820 .0120 .0608 .0408 .0208 0.0210 .0628 .0628 .0636	0.2402 2914 2878 2878 20794 1140 07588 0396 0202 0 0201 0 0204 12710 0 0204 127176 26428 - 1276 2654 - 1276 - 1276	.2624 .2410 .1986 .1718 .1420 .0752 .0568 .0382 .0194 .0696 .0812 .1670 .2126 .2558 .4500 .5528 .4500 .5528 .4500	1206 1628 2076 2528 3458 4416 5396	0.1216 .1984 .1536 .1284 .1002 .1284 .1002 .0354 .0182 0 .0374 .0776 .1794 .2024 .2024 .2438 .2438 .2438 .25	1540 1970 2406 3308 4244 5204 6208	6098	0158 0 1638 0 1638 0 1638 0 1644 1856 11446 1225 1225 13060 1466 15060	1110 1068 09772 0752 0290 0150 0156 - 0316 - 0658 - 1022 - 1796 - 2208 - 3966 - 4897 - 5872	0300 0462 0628 0980 1340 1732 2134 2978 3866	0718 0728 0624 0462 0253 0130 0 0138 0288 09364 - 0936 - 12866 - 20868 - 2762 - 4672 - 4672	0260 0410 0562	0 0118 0244 0382 0524 1162 1522 1528 2694 3544	

TABLE I - Continued

POINT DRAG COEFFICIENT FOR ISOENERGIC FLOW - Continued

**M**<sub>0</sub> = 0.40

						<b>L</b>								
AH P	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
100 150 200 250 300 400 500 600	.7240 .6210 .6210 .4168 .3134 .2616 .2098 .1052 .0796 .0796 .0796 .0796 .0796 .1060 .1060 .1060 .1266 .2126	7062 5088 5088 5087 2066 1556 0780 0525 0780 0525 0784 15702	.5966 .4966 .4066 .30528 .1020 .07612 .0254 .0256 .057666 .05766 .05766 .05766 .05766 .05766 .05766 .05766 .05766 .057666 .05766 .05766 .05766 .05766 .05766 .05766 .05766 .05766 .057666 .05766 .05766 .05766 .05766 .05766 .05766 .05766 .05766 .0576	- 626 - 5448 - 5926 - 3926 - 3926 - 1956 - 1956 - 1957 - 1957	6450 5580 1624 38306 1960 1476 0988 0744 00252 00	- 6161 - 5161 - 5162 - 3782 - 2382 - 1950 - 0772 - 0782 - 0784 - 0784 - 0784 - 1998 -	5.24.24.24.25.26.26.26.26.26.26.26.26.26.26.26.26.26.	- 5650 - 5754142 - 5754142 - 18444 - 19544 - 19554 - 1	1176 3144 2628 1808 1364 0916 09230 00230 00236	- 0696 -	75814 143582 20112 1730762 20112 1730762 1730762 1730762 1730762 173762 173762 173762 173762 173762 173762 173762 173762 173762	11000 11000		5588 55910 19250 1928 11968 11968 11969 00 00 10214 00 10456 11970 10456 11970 10456 11970 10456 11970 10456 11970 10456 11970 10456 11970

ΔH Q <sub>o</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	a.65	0.70	0.75	0.80
.700 .500 .400 .300 .250 .150 .025 .025 .025 .025 .025 .025 .025 .0	.18\u00e48 .1514 .1160 .0786 .0596 .0400 .0200 0 .0006 .0111 .0838 .1272 .1716 .2168 .2628 .25566 .4558	2676 2675 2054 2064 11458 11462 0756 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	.2620 .23920 .16920 .16920 .057600 .05760 .05760 .05760 .05760 .05760 .05760 .05760 .05760 .057600 .05760 .05760 .05760 .05760 .05760 .05760 .05760 .05760 .057600 .05760	.0182 0 0190 0380 0578 0774 1182	2016 2024 17340 12788 06880 0550 0550 00176 003680 0550 00176 003680 003	0.1622 .1804 .1604 .1940 .0940 .0650 .0170 0.0176 .0736 .0176 .07428 .1510 .19324 .19324 .19324 .19324 .29364 .19324 .29364	.1466 .1318 .0888 .0618 .0474 .0322 .0162 0 -07522 -07522 -1080 -1174 -1878 -11878 -11878 -11878 -11878	0.12528 1.120849 1.120849 0.0154 0.01	-1126 -10790 -0750 -07546 -0422 -0288 -05946 -0314 -0450 -11766 -11766 -11766 -11766 -11766 -11766 -11766 -11766 -11766 -11766 -11766 -11766 -11766 -11766 -11766 -11766 -11766 -11766 -11766	0.0908 .0926 .0848 .07026 .05708 .05708 .0138 .0298 .04520 .1708 .1708 .2098 .2098 .23788	0.0464.008 0.0464.008 0.0464.00 0.04	05412 05412 05419 0 012 0 012	0244 0380 0522 0826 1156 1504 1872 2652 3484 5260

MATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

TABLE I - Continued

# $\left[\mathbf{x}_{\mathbf{0}} = 0.45\right]$

$\frac{\Delta H}{q_o}$	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.1 <b>0</b>	-0.05	0	0.05	0.10	0.15
0.800 .700 .600 .500 .100 .250 .150 .075 .025 0.025 0.075 .100 -1200 -1200 -1200 -15	1016 1528 2040 2552 3064 4088 5128 6144 7170	684820 59820 399766 19982 1998 10050 0 0250 0 0250 0 02546 - 10012 - 250362 - 71086 - 71086	6540 6540	.2874 .2404 .1952 .0728 .0488 .0242 .0728 .0494 0494 0494 1474 2474 2974 2974 3978 4998	6545920666666666666666666666666666666666666	6022 52774 3634 23163 1864 1942 0796 0 0236 0 0480 0 0716 1928 1949 1928 1949 1928 1949 1949 1949 1949 1949 1949 1949 194	579028 5514020308 5514020308 5514020308 18380346 00406 00406 00902 14308656 143086 143	5,42 4,9218 3,45,4 2,220 1,792 1,3508 0,4528 0 02,528 0 02,528 0 02,528 1,40,76 1,40,7	- 0458 - 0684 - 0914 - 1380 - 1850 - 2324 - 2802 - 3766 - 4752	4998 459366 25116 211128 21116 211128 21116 211128 21128 211	0682 0884 1336 1794 2256 2726 3670 4642 5604	1312 1764 2222 2684 3620 4584 5538 6514	1,403 3,452 2,294 1,582 1,204 1,582 1,204 1,582 1,204 1,082 1,204	- 0416 - 0622 - 0834 - 1264 - 12148 - 2148 - 2598 - 3516 - 4462 - 6366

$\frac{\Delta E}{q_o}$ P1	0.26	0.25	0 <b>.30</b>	0,35	0.4 <b>0</b>	0.45	0 <b>.</b> 5 <b>0</b>	0.55	0.60	0.65	0.70	0.75	0.80
0.800 .700 .600 .500 .2500 .2500 .1500 .075 .055 0.055	3128 3266 2666 2660 21814 1184 11770 0582 0392 0196 0 - 0406 - 0406 - 1238 - 1278 - 1278 - 2554 - 34508 - 43520 - 35320	.098 .0748 .0566 .0382 .0192 .0196 0396 0594 1212 .16368 2508 3408 3408	2551466 22354896 137588 137588 1072476 00186	2300 23400 18388 13168 105288 005288 005288 00756 - 00574 - 1056 - 11564 - 115	20160 17204	1680 18198 11598 11186 0928 0648 09326 016 093530 017 12350 11488 23150 11491 11491	.1314 .0878 .0610 .0466 .0318 .0160 0 .0168 .0168 .0169 .0168 .0532 .0690 .11848 .3966 .3986 .3986 .4856	1296 1216 1202 1036 0824 0578 04302 0152 0 0326 - 0326 - 13986 - 1386 - 2386 - 23884 - 38846	0146 0 01514 - 01512 - 0472 - 06388 - 17324 - 2938 - 38604 - 5580	0940 0940 0850 0750 05968 05968 00136 00292 004512	0.0672 .07742 .0630 .0462 .0360 .0128 0 .0138 0582 0598 1614 1920 2780 2780 4476	.0118 00128 0264 0404 0550 0864 1200 1552 1552 1920 2694	0.0100 0.136 0.0450 .0362 .0296 .0108 0 - 0118 - 0518 - 0518 - 0518 - 0518 - 0518 - 11846 - 11846

TABLE I - Continued

											T	1-1-1-1		
						м	o = 0.5	]						
AH qo	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.80 .70 .60 .50 .20 .15 .10 .07 .02 .05 .07 .10 .15 .07 .02 .05 .07 .05 .05 .07 .05 .05 .07 .05 .05 .07 .05 .05 .05 .05 .05 .05 .05 .05 .05 .05	00000000000000000000000000000000000000	6567426426426426426464264264642642644264264426426	655554888824 657554888824 67656888824 67656888824 67658486 67658486 676584 67658 676584 67658 676584 67658	231608 18608 109342 070468 0 07047 0 07048 0 07048 0 112848 1 12884	78000000000000000000000000000000000000	1850	5.652 4.926 3.442 2.1618 1.1339 0.6456 0.045	.4106 .3356 .2152 .1734 .1308 .0878 .0660 .0442 .0224 0 0672 0900 1352 1806	0 0220 0440 0662 0888 1332 1782 2236	2174 2440 2058 1662 1258 0846 0426 0426 0 216 0 - 0450 - 0874	0 0212 0426 0640 0860 1294 1732	2355842 2355842 152800 06100 06108 0 04684706 0 04684706 0 04684706 0 0 04684706 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0204 0410 0618 0830 1250 1678 2112 2554 3446 4354	0402 0604 0814 1228
$\frac{\Delta H}{q_o}$	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80	
0.800 .700 .500 .400 .250 .250 .100 .071 .052 -075 -100	3166 .32514 .3250 .2620 .2080 .1776 .1450 .0752 .0568 .0194	2826 2484 11910 11074 10730 10730 105378 0 01880 -036780 -11880 -11880	0186 0374 0564 0762 1154 1556	0742 1128 1522	0176 0352 0534 0722 1098 1486	.1586 .1398 .10914 .0630 .01480 .0166 .0166 .0166 .01748 .07078 .07078	.1462 .1304 .1102 .0866 .0602 .01460 .0312 .0160 0164 0502 0660 1038 1103	.1322 .1208 .0816 .0570 .0438 .0152 .0156 03184 0656 1368	.1162 .1086 .0762 .07538 0414 0282 0146 00150 0464 04632 0972	09754 08750 070288 070288 001490	.0800 .0752 .0652 .0360 .0248 .0128 .0128 .01276 01276 01278 01278 01278 01278	0630 .0556 .0418 .0328 .0120 .0126 .0126 .0260 -0260 -0260 -08180 -1524 -15886 -1886	2556 3344 4166	

TABLE I - Continued

 $\left[\mathbf{x}_{0}=0.55\right]$ 

P <sub>1</sub>	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700 .500 .500 .200 .200 .100 .075 .025 .025 .025 .150 .150 .250 .360 .360 .360 .360 .360 .360 .360 .36	0 . 75.82 . 55.82 . 57.82 . 57.81 . 69.62 . 18.90 . 06.62 . 18.90 . 06.63 . 18.90 . 06.63 . 18.90 . 06.63 . 06	36748 22994 18276 09168 09250 00250 00260 00250 002600 00260	5562 549968 55968 55967528 1855967528 22564 0 00452 0 00454	604342 5239342 523935686666 173494708 004570 180580	5886 5104 4298 3468 2624 2192 1750 1322 0884 0662 0142 0 0 222 0 0222	- 0440 - 0662 - 0880 - 1326 - 1774 - 2216 - 35664 - 3566 - 3568 - 6272	51246 51246 5125 5126 5126 5126 5126 5126 5126 512	5288 4670 3250 2478 2478 21666 12662 1264 0624 0624 0624 0636 0636 0636	5062 4506 38768 31626 2036 1634 1240 08214 00210	4818 43746 3746 3746 2768 1992 1600 1216 0612 0410 0208 0 0208 0418 0418 0428 1694 1694 1694 1694 1694 1695 1694 1694 1694 1695 16	151462 151462 1514662 1514662 1514662 1514662 1514662 16642	4266 -3944 -2896 -2896 -2246 -1898 -1166 -0796 -0200 -0408 -0810 -0810 -0810 -1226	0 3848 37728	.3596 .3492 .31686 .2110 .1790 .1450 .0750 .0564 .0388 .0388
AH q <sub>o</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80	
0.800 .700 .500 .500 .300 .250 .025 .025 .025 .025 .025 .025 .0	0.2584 .3194 .3232 .2988 .2576 .17408 .07570 .0188 .07576 .0768 .11640 .15774 .15774 .2329	2724 27948 27948 19572 1362 10710 00736 00182 00752 -00752 -11440	0364 0550 0734 1118 1512 1906	2300 231649 17836	2002 2002 11408 0014 1294 00168 00168 00168 00168 00168 00168 00168 00168 00168 00168 00168 00168 00168 00168 00168 00168 00168	1786 1822 1578 1380 1148 0618 0618 0162 0 0518 0162 0 0508 1042 1412 1779	1208545066 005555066 00560615 0056015	- 0512 - 0476 - 0638 - 0982 - 1738 - 1702	0300 0458 0616 0950 1298 1654 2026	0.0652 1004 0.0856 0.0700 0.0458 0.0136 0.0136 0.0136 0.0138 0.0136 0.01	0824 07636 06456 06456 06456 0656 0656 0656 0656	0258 0396 0538 0842 1496 1848	0240 0372 0508 0800 1112 1438	

NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

TABLE I - Continued

 $[M_0 = 0.60]$ 

$\frac{\Delta H}{\overline{q}_o}^{P_1}$	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .500 .500 .500 .2500 .150 .075 .025 0 .025 075 1500 2500 2500 300	- 6062 - 5348 - 3544 - 2652 - 1762 - 1762 - 1762 - 0660 - 0222 - 0660 - 0876 - 17546 - 17546 - 2620	61326 513796 613796 613796 613796 613797 613	0862 1298 1732 2160	\$65880404 \$5021953042404 \$5021953042404 \$50217485484 \$6642 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$66888047 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$668880447 \$66888047 \$668880447 \$66888047 \$6688047 \$66888047 \$66888047 \$66888047 \$66888047 \$66888047 \$66888047 \$66888047 \$66888047 \$66888047 \$66888047 \$66888047 \$66888047 \$66888047 \$66888047 \$66888047 \$66888047 \$6688047 \$66888047 \$66888047 \$6688047 \$66888047 \$66888047 \$66888047 \$66888047 \$66888047 \$668	2514 21682 21682 1266 00431 0042 12766 0042 0042 00431	5-18-0-8 5-18-0	3956 3208 3232 1634 16334 16334 16334 1634 1634 1640 1640 1640 1640 1640 1640 1640 164	-51460 -3856 -3138 -2386 -2000 -1608 -2010 -0408 -0204 -0618 -0618 -0822 -1660 -124076 -20764	4936 43750 3066 1960 1960 10600 0600 0000 0000 0000	47120 3638 2988 1924 1572 0786 0596 0 0200 0 0402 - 0602 - 0602 - 1614 - 2038	.4470 .40508 .2954 .1884 .1520 .0774 .0580 .0396 .0396 .0396 .0396 .0792 .1198 .1196 .1196 .2012		.3662 .3252 .27218 .1792 .11450 .1102 .0558 .0376 .0376 .0376 .0376 .0576 .0576 .1162 .1162 .1562 .1562	3586 31402 3

$\frac{\Delta H}{q_o}$	4	0.25	0.30	0.35	o.4 <b>o</b>	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
0.800 .700 .600 .500 .400 .250 .250 .150 .075 .055 .055 075 075 075 150 250 250 300 400	. 3216 .32942 .2942 .2518 .1984 .1050 .0710 .0754 .0360 .0182 .00184 .0376 .00742 .00742 .1126 .1126 .1126 .1126 .1126 .1126 .1126	2940 2762 12402 19532 1030 0522 0552 00178 00178 00178 00178 11490 11490 11490 11268	0.2626 .26362. .2576. .1834. .1288. .0508. .0507. 0.017. 0	2284 23590 1748 15090 10956 00532 00168 0 - 01746 0 - 05420 0 - 0168 0 - 0168 0 - 1063 10634 11810 11810	0.1854 .11142 .1958 .14388 .0940 .06376 .0164 0 013680 0 - 013680 0 - 013680 - 10380 - 11774 - 17774	183249 183249 11562 11562 1136	0 11488 116748 114808 110748 10054308 0 15160 0 15180 0 0 15180 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	1328 11910 1010 0798 0551 0288 0148 0 152 - 01506 - 0461 - 09506 - 1306 - 1602	0940 0750 0524 0402 0274 0140 0 - 0146 - 0294 - 0448 - 0928 - 1268 - 1614	1030 0978 0896 0696 04980 00134 0 01480 0 015888 - 104888 - 11988	0638 0458 0354 0126 0 132 0 1370 - 0410 - 0556 - 1180	0688 0668 0572 0428 0226 0218 0 - 0124 - 0256 - 0538 - 1148 - 11488	0496 0376 0298 0108 0 0116 0 0368 - 0502 - 07994 - 1412

TABLE I - Continued

# $\left[\mathbf{M}_{\mathbf{0}} = 0.65\right]$

$\frac{\Delta H}{\overline{q_0}}$	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700 .600 .500 .400 .300 .250 .100 .075 .055 0 .025 075 150 150 250 300	0.6892 5976 5976 1,292 3366 2516 2094 1672 1252 0620 0 0832 0416 0208 0 0416 - 0418 - 08208 - 1240 - 1240 - 2060 - 2468 - 3276	0 0206 0416 0618 0824 1234 1644 2052	-5744 -4924 -4104 -3286 -2464 -1642 -1250 -0820 -0614 -0920 -0614 -0820 -1636 -1636 -2042	.5616 .4830 .4938 .3240 .2034 .1626 .1218 .0812 .0608 .0406 .0204 0 -0410 -0410 -1820 -1626	.4766 .47266 .37966 .3192 .2006 .1608 .1204 .0602 .0202 0 0406 0406 0406 1212 1616 2022	.4620 .3890 .3140 .23780 .1588 .1190 .0796 .0596 .0200 0 - 0198 - 0404 - 0802 - 1606 - 2006	151644 45808 36824 19566 115766 115768 00594 00594 00594 11592 11592 11592 11592 11592	.4980 .3720 .3022 .2294 .1920 .0778 .0596 .0196 .0588 -0788 -0788 -1182 -1580 -1182	.4244 .2958 .2258 .1888 .1524 .0766 .0384 .0194 .0582 .0582 .0770 .11566 .11666	-47902 -47902 -2888 -2288 -1854 -1126 -07566 -0756 -0758 0 -05866 -07586 -11550 -11550 -11550 -12338	.42/96 .39/16 .39/16 .21/60 .11/60 .07/58 .03/74 .05/62 .05/62 .05/62 .05/62 .05/62 .05/62 .05/62 .05/62 .05/62 .05/62	4134 3278 22736 21100 1778 1438 1038 00548 00186 00186 00186 00186 00186 00186 00186 00186 00186 00186 00186 00186 00186 00186	.0538 .0362 .0182 0 0182 0570 0552 0740 1116 1498	- 5568 - 5594 - 5596 - 5560 - 19690 - 11672 - 07528 - 07528 - 07528 - 01864 - 05544 - 05544 - 0187302 - 11478 - 11478 - 11478

$\frac{\Delta H}{\overline{q_o}}$ P <sub>1</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0 <b>.50</b>	0.55	0.60	0.65	0.70	0.75	0.80
0.800 .700 .600 .500 .400 .250 .250 .150 .075 .025 .050 .075 .075 .075 .075 .075 .075 .07		0526 0704 1066 1434 1808	2374 2652 2259 22496 17538 .09652 .0492 .0368 0 .0168 05914 06948 1042 1780 2152	2332 2352 2352 2114 1718 11214 0934 0478 0324 0164 0 - 0150 1 - 0550 1 - 0150 1 - 1028 1 - 1028	2128 1976 1632 1412 1466 0898 0614 0464 0514 0158 0	0480 0646 0984 1330 1684 2046	1564 1650 1140 1266 10626 0570 0439 0148 0 0 1 0152 - 0310 - 0466 - 1606 - 1606	0452 0610 0934 1268 1612	.1210 .1202 .1092 .0936 .0740 .0518 .0270 .0138 00140 0290 0436 1572 1518	0.0906 .0988 .0862 .0692 .0488 .0374 .0256 .0132 0 - 0136 0278 0570 0878 11580 11580	0882 0868 0780 0638 0456 0352 0124 0128 - 0128 - 0266 - 0402 - 0548 - 0846 - 1160 - 1188	.0576 .0420 .0326 .0226 .0116 0 0122 03524 0524 0812 1118 1436	.0538 .0572 .0580 .0298 .0108 0 .0114 036498 0376498 07784 10744

TABLE I - Continued

AH Qo	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
075 100 150 200 250	-5680 -4830 -4830 -3180 -2374 -1576 -1576 -0782 -0588 -0592 -0780 -0780 -0780 -11648	5586 4764 -3956 -3954 -3956 -1566 -1778 -0786 -0792 -0786 -0786 -1760 -1760 -1160 -1160 -1160 -1160	4694 49018 31184 1164 11782 0782 0192 0 0196 0 0196 0 0774 - 1156 - 1156 - 1156 - 1156 - 1156 - 1156	5.4658208066884 5.4658208066884 6.57589 6.57589 6.577524	0766 11146 1526 1908	5124 -372988 -29988 -29988 -1850 -17568 -03768 -03769 -0762 -15188	1982 19508 19508 1944 11128 10766 103	0372 0562 0650 1124 1500 1878	441-94-94-94-94-94-94-94-94-94-94-94-94-94-	- 59 76 6 - 59 46 6 - 78 20 17 80 0 - 10 80 6 - 10 95 64 0 - 10 95 75 66 0 - 10 95 75 66 0 - 10 95 75 66 0 - 11 14 75 75 66 0	0360 0546 0730 1096 1464 1836	.368i .360i .203i .1016 .0730 .0354 .0176 .0356 .0536	.3812 .3518 .2568 .1984 .1356 .0520 .0348 .0520 .0174 0 -0178 -0732 -0712 -1074 -11434	0524 0702 1058 1418 1782

AH qo	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
150 200 250	35256 31832 2398 18764 1594 00662 05366 0168 0 0 10168 0 10169 10169 10169	.2912 .2684 .2302 .1816 .1546 .1260 .0960 .0328 .0164 0 -0170 -0336 -0508 -1028 -1338	.2658 .2520 .1748 .1496 .1292 .0934 .0632 .0160 0 .0166 0328 0498 0698 1010 1752	2370 2312 2082 1680 11482 1156 00156 00156 0 01622 0 01646 0 01646 0 01656 0 01656 0 01656 0 01656 0 01656 0 01656 0 01656	2029 21406 11602 11956 11408 00596 0	0306 0466 0628 0954 1288 1630	1658 11658 11250 1042 005524 005524 0014 00298 00150 00298 0038 0038 0038 0038 0038 0038 0038 00	0.1284 11476 11374	1208 10908 07308 07308 07508 07508 07508 07508 07508 07508 07508 07508 07508 07508 07508 07508 07508 07508 07508	0 090764 098688 098688 098752 0 0 102455558 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	.0920 .0886 .0788 .0452 .0350 .0258 .0130 .0130 .0258 .0596 .0598 .0138	0700 0582 0426 0324 0114 0 0124 0 0246 - 0346 - 0516 - 0798 - 1094	.0596 .0598 .0518 .0382 .0300 .0208 .0106 .0234 .0340 .0492 .0764 .1054

MATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

TABLE I - Continued

M<sub>o</sub> = 0.75

AH Qo	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700 .600 .500 .100 .250 .150 .075 .050 .025 -050 -075 050 075 050	.1478 .2234 .1478 .1478 .1478 .1478 .1478 .0734 .0364 .0182	.5298 .4502 .37736 .2974 .2220 .1844 .1472 .11472 .07548 .0364 .0364 .0546 .0728 .0728 .0728 .0728 .0728		1,382 2,3656 2,21902 1,1090 1,1090 0,0562 0,0180 0,				.4052 .3430 .2772 .2098 .1752 .1406 .0706 .0530 .0352 .0176 .0356 -0576	.7550 .3758 .2068 .1728 .1386 .0698 .0518 .0318 .0318 .0318 .0550	. 13h2 .3836 .2672 .2034 .1704 .1570 .0518 .0346 .0172 0 -0172 0 -0172 0 -0172	.3196 .2616 .1998 .1676 .1350 .0682 .0512 .0170 0 .0174 0346 0520 0694 1394	.3966 .3578 .2554 .1960 .1648 .1004 .0506 .0168 0	.3748 .3428 .3002 .2486 .1918 .1614 .1308 .0498 .0498 .0366 0.0166 0.0170 0538 0510	0 0168 0334 0502 0672 1012

AH Qo	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
0.800 .700 .600 .500 .400 .250 .250 .150 .075 .050 .025 0 025 050 150 150 150	0.3050 .3288 .3088 .2768 .2768 .2334 .1822 .1540 .0950 .0460 .0950 .0164 .0160 .0164 .0160 .0164 .0166 .0166 .0166 .0166 .0166	.2914 .2882 .2636 .2246 .1766 .1498 .1220 .0928 .0472 .0158 0 .0158 .0158 .0058	.2654 .2490 .1708 .1454 .1906 .0612 .0310 .0156 .0310 .0156 .0310 .0156 .0460 .0510	0.20926 -2394 -2048 -1049 -1152 -0852 -0152 0 -0152 -0452 -0452 -0452 -0958 -0958	2192 1932 1932 11574 1352 0580 0440 00148 0 0158 0 0158 0 01680 0 0680 0 0680 0 0680 0 0680 0 0680 0 0680 0 0680 0 0680 0 0 0680	0.1716 .1934 .1198 .1294 .1070 .0826 .0564 .0128 .0286 .0144 0 .0300 .0452	.1662 .1412 .1232 .1024 .0794 .0544 .0278 .0140 .0278 .0140	0.1384 .1498 .1162 .09740 .0760 .0268 .0136 .0136 .0136 .0136 .0136 .0136 .0136	.1214 .1084 .0918 .0722 .0500 .0382 .0258 .0130 0 0136	0.1074 .1094 .0998 .0858 .0476 .0364 .0246 .0126 .0132 01604 0546	.0954 .09508 .0636 .0448 .0346 .0120 .0126 .0126 .0526 .0526	.0784 .0712 .0586 .0418 .0324 .0222 .0114 .0120 0120 0506 0780	0644 0620 0528 0586 0500 0006 0 0114 - 0256 - 0484 - 0750 - 1055

TABLE I - Continued

 $\left[\mathbf{M}_{\mathbf{0}} = 0.80\right]$ 

P <sub>1</sub> -0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 0.5851	5002 502 502 502 502 502 502 503 503 503 503 503 503 503 503 503 503	1938 19410 19480 19780 19780 19780 19780 197900 19790 19700 19700 19700 19700 19700	14168 14168 27604 17174 1026 05140 0168 0168 0168 0172 0168 0173 0173 0173 0173 0173	0336 0506 0676 1012	0 0170 0334 0506 0674 1010	0 0170 0334 0504 0672 1008	3827-1668 119688 119688 119688 10660346 0 01350706 0 01350706 0 01350706	195344 139428 139669326 00155660 00155660 00155660	0 0166 0328 0496 0664 0996	1388 1398 1398 1398 1396 1396 1396 1396 1396 1396 1396 1396	0322 0488 0654 0582	57466800464446800464476864768887688476884	3458 348010 328386 1152320 1152320 006272 0015180 0015180 0015180 0015180 0015180 0015180 0015180 0015180

AH Qo	<b>0.</b> 20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
0.800 .700 .600 .500 .400 .250 .250 .150 .050 .025 -050 -075 -150 -150	.5212 .50762 .27682 .17692 .12084 .0614 .0310 .0154 .0310 .0158 .0474 .0956	.2950 .2856 .2582 .2190 .1714 .1182 .0896 .0450 .0152 .0358 .0358 .0358 .0462 .0964	2650 2450 2450 24160 141576 251498 2015 20150 20	.2216 .2418 .23010 .1604 .1370 .0854 .0578 .0292 .0146 0 .0150 .0292 .01452 .0608	- 0148 - 0292 - 0444 - 0598 - 0904	1826 1979 1979 1140 180 14178 001 198 198 001 198 198 001 198 001 198 001 198 001 198 001 198 001 198 001 198 001 198 001 198 001 198 001 198 001 198 001 001 001 001 001 001 001 001 001 00	0 11408 1752 1662 17512 1006 10776 00136 0 01266 0 01266 0 0577	1516 1310 10960 0744 0512 03962 0138 0138 - 0276 - 0416 - 0854	0.1134 1346 1214 1080 0910 0710 0490 0374 0252 0128	11106 10054 10054 00458 003242 0 12592 0 12592 0 005532	0.0880 .0982 .0912 .0792 .0644 .0312 .0232 .0118 .0246 .0378 .0788	0834 0810 0722 0586 0416 0322 0220 0112 0 0118 - 0364 - 0494 - 0762	0.0612 .0686 .06140 .0534 .0386 .0306 .0106 0 0112 0348 0474

TABLE I - Continued

$\frac{\Delta H}{q_0}$	-0.50	-0,45	-0.40	-0.35	-0.30	-0,25	-0.20	-0.15	-0.10	-0,05	0	0.05	0.10	0.15
0.800 .700 .600 .500 .400 .250 .250 .150 .150 .075 .050 .025	4732 4008 3302 2616 1944 1612 1284 0958 0636 0476 0316 0	4698 3988 3292 2612 1944 1614 1286 0960 0638 0476 0318 0156 0	.3962 .3278 .2606 .1942 .1614 .1288 .0962 .0640 .0478 .0318 .0156	.4600 .3928 .3258 .2596 .1938 .1610 .1286 .0962 .0640 .0478 .0318 .0156 0	.4538 .3888 .3232 .2580 .1930 .1606 .1284 .0950 .0478 .0318 .0156	.4464 .3838 .3202 .2562 .1920 .1600 .1280 .0538 .0478 .0156	.4380 .3782 .3166 .2540 .1908 .1590 .1274 .0954 .0476 .0318 .0156 0	.4284 .3718 .3122 .2514 .1894 .1580 .1266 .0950 .0634 .0474 .0316 .0156 .0	.4174 .3644 .3074 .2482 .1876 .1256 .0472 .0630 .0472 .0314 .0154 0	4052 3560 3018 2448 1854 1550 1246 0936 0468 0312 0154 0	3916 3468 2958 2400 1830 1532 1232 0928 0620 0464 0310 0152 0	.3764 .3364 .2888 .2362 .1802 .1512 .1218 .0918 .0614 .0460 .0308 .0152	.5592 .3248 .2810 .2312 .1772 .1488 .1202 .0906 .0608 .0456 .0304 .0150	.3400 .3120 .2724 .2256 .1738 .1464 .1182 .0894 .0600 .0450 .0302 .0148 0
050 075 100 150 200	0472 0628 0934	0474 0630 0940	0318 0476 0632 0944 1254	0476 0634 0946	0476 0634 -:0948	0476 0636 0950	0476 0634 0950	0476 0634 0948	0474 0632 0946	0472 0630 0946	0470 0528 0940	0468 0624 0936	0464 0620 0930	0460 0614 0922

P <sub>1</sub> Δ <u>H</u> q <sub>0</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
0.800 .700	.3182	.2936		.2308	0.1866			******					
.600 .500	.2976 .2630			.2428	.2190 .2128	.1908 .1960		0.1076 1538	0.1256	0.0868			
400	2194	2126	2052	.1968	.1874	.1772	.1656	.1524				0.0674	
.300	.1700	.1658	.1612	.1560	.1504	.1442		.1298					0.0714
.250	.1434	.1402	.1368	.1328	.1286	.1240	.1188						
.200	.1162	.1138	.1112	.1084	.1054	.1020		.0942					.0656
.150	.0878	.0864	.0845	.0826	.0806	.0782	.0758	.0730		.0666			
.100	.0592	.0582		.0560	.0546	.0532	.0518	.0500		.0462			.0388
.075	.0444			.0422	.0412		.0392	.0380					.0298
•050	.0298	.0292	.0288	.0282	.0276			.0256					.0204
.025	.0146	.0144											.0104
0	0	0	0		0		0		0	0	0	0	0
025			0148										
050			0296										
075			0444										
100	0608	0602	0594	0586	0576	0566	-,0556	0544	0532	0518	0502	0484	0456
150			0896										
200	1226	1214	1202	-•1188	1174	1158	1138	-•1118	1096	1072	1046	1018	0988

TABLE I - Continued

w<sub>o</sub> = 0.90

AH qo		-0.45	-0.4 <b>0</b>	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	~0.05	0	0.05	0.10	0.15
0.800 .700 .600 .500 .400 .250 .250 .150 .075 .050 .025 050 075 150 150	1408 37066 24726 24726 17992 11990 05888 05490 0 1146 0 04378 0 04578	37722 37070 21806 1500 11982 0592 0144 0 1148 0 - 01438	13682 1376632 1370431632 137043163 137043 13704	36550 NG 16666 0664 0664 0664 0664 0664 0664 066	. 1290 .3666 .31,266 .1810 .1506 .1506 .0598 .0148 .0298 .0146 0 .0150 - 0150 -	423424 • 50241466 • 1520098 • 1520098 • 1520094 • 1520098 • 1520094 •		90 90 90 90 90 90 90 90 90 90 90 90 90 9	349376 3493778 3493778 3493778 3493778 3493778 3493778 349378 349	3900 3484 23762 111888 217474 1085 0014 00159 001059 001059	33340 1767622 117888424 118884944 0 01594 0 01594 0 01594	3526-642 3624-6642 3722-11164-7-64 3722-1164-7-64 3722-11	35026 37140 371402 1140266 114026 114	30334 20364 201668 114046 114046 114046 10856 0054388 00144 0054388 00144 0054388 00145 00

AH qo	0.20	0.25	0 <b>.</b> 3 <b>0</b>	0:35	0.40	0.45	0 <b>.</b> 5 <b>0</b>	0.55	0.60	0.65	0.70	0.75	0.80
0.800 .700 .600 .500 .400 .250 .250 .100 .075 .050 .025 -050 -075 -100 -150	0290 0434 0580 0874	2928 27662 24662 16600 13598 085520 001338 0 01886 0 01886 0 0577	26586 26596	.2382 .2424 .1922 .1514 .1290 .0506 .0272 .0134 .0472 .0134 .0472 .0420 .0420 .0420	.2026 .2216 .2116 .1840 .1466 .1252 .0528 .0528 .0368 .0132 0 .0138 -0276 -0414 -0840	0 1562 1972 1962 1748 1410 1210 0760 0516 0362 0130 0 0136 - 0270 - 0408 - 0538	16762 17946 1131644 1131644 1131644 113164 1	0 1292 15928 1582 1582 15814 1007 1582 1007 1007 1007 1007 1007 1007 1007 100	135966 135968 112058866 00455420 00455420 00455420 00455420 00455420 00556 00556	0 1040 1040 10940 00856 00456 0011 003708 0 1055	1052 1022 10790 0622 0432 0332 02112 0 0118 - 0368	0.0810 0.0908 0.0584 0.0516 0.0116 0.0116 0.0116 0.0150 0.0250 0.0250 0.0250	0746 0668 05482 0386 0298 0204 0102 0 - 0110 - 0222 - 0338 - 0458 - 0706

TABLE I - Continued

 $\left[\mathbf{M_o} = 0.95\right]$ 

AH P1	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700 .600 .500 .400 .250 .250 .150 .100 .075 .050 .025 -050 -075 -100 -1150	0.4752 -\$1086 -\$14,686 -\$28,54 -226,58 -13,74 -1090 -013,4 -0266 -013,4 -0266 -03,98 -07,99	4090 34458 24562 1672 11900 0824 0408 0270 0136 00270 0 0136 0 0136 0 04532			2856 2270 1694 1120 0840 0556 0418 0276 0138 0 0140 - 0276 - 0414 - 0544	.4002 .3244 .2268 .1696 .1410 .0558 .0426 .0558 .0426 .0138 0 - 0140 - 0278	39546 3959328244 39593282641244 39593282641244 39593282 395932 39592 39592 39592 39592 39592 39592 39592 39592 39592 39592 39592 39592 39592 39592 39592	38550 38550 38050 11600	38214 3272862 114040 118140 11	3740 372578 272578 2727278 2727278 272727 27272	3644 3196 2710 2194 1660 1388 11110 0836 0556 0428 0138 0 0148 - 0148 - 0420	3124 2662 21642 1642 1376 1102 0554 0418 0276 0138 0 - 0140 - 0420 - 0556	3406 34068 32630 213624 13662 13662 05174 00174 00174 00174 0178 00178 00178	.3260 .2934 .25346 .2090 .1600 .10816 .0546 .0546 .0136 .0136 .0276 .0276 .0276

$\frac{\Delta H}{q_0}$	0.20	0.25	0.30	0.35	o.4 <b>o</b>	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
0.800 .700 .600 .500 .400 .250 .250 .100 .075 .050 .025 0	.3096 .2834 .2474 .24046 .1574 .1326 .0808 .0540 .0270 .0136 .0270 .0138 .0274 .0408	27944 1996 1996 1996 10596 00596 0134 012710 0 012710 0 054	2688 25704 1938 15086 12762 0784 05264 0136 0 0136 0 0136 0 0136 0 0150 0 0150 0 0150	2434 2414 2202 1874 1478 10770 0518 0770 0130 0130 0 01368 - 0402	2132 2232 2088 1802 11416 10988 0758 0558 01266 0 01266 - 0353	2018 1958 1728 1738 1180 0952 0758 0498 03572 0126 0 - 0130 - 0392	1766 1808 1630 1324 1140 0972 0718 0486 0370 02146 0 0128 - 0256 - 0514	0.1452 1636 1528 1264 1094 0900 0696 0472 0340 0122 0 - 0126 - 0250 - 0378	1430 1410 1196 1044 0672 0458 0350 001 0 0124 0 02476 0 0496	0.1174 .1274 .1120 .0988 .0826 .0614 .0142 .0338 .0216 .0114 0 .0240 .0362 .0484	1034 0926 0780 0614 0424 0218 0110 0 - 0116 - 0232 - 0352 - 0472	0 0914 09550 097582 09550 0950 0950 09500 09500 09500 09500 09500 09500 09500 09500	.0816 .0770 .06714 .05114 .0382 .0296 .0200 .0102 0 0108 0218 0330

TABLE I - Concluded

[N<sub>o</sub> = 1.00]

P <sub>1</sub>	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	.0.10	0.15
0.800 .600 .500 .400 .250 .150 .075 .050 0.025 0.025 0.025 0.025 075 150	.3762 .3170 .2602 .1520 .1258 .1000 .0714 .0194 .0120 .0216 .0120 .0120 .0361 .0482	779.84 779.86 779.86 779.86 779.86 779.87 77	37914 379146 379146 379168 379168 379168 37916 3	3796 3702 3656 3700 3701 3701 3701 3701 3701 3701 3701	\$6.488 \$7.488.88 \$7.488.88 \$7.488.88 \$1.45.49 \$7.488.88 \$1.45.89 \$	57688 77168466 77168466 77168466 77178270 771782	56024600446 700646004460 700646000000000000000000000000000000000	369540 36159218 361592 36159 361	334028 344028	3516108400 5516108400 5516108400 5516108752986 5516100 551610 551	0264 0398 0528	.3406 .2998 .2964 .1564 .1568 .1566 .0786 .0596 .0130 .0130 .0130 .0398	35500 25508 15506 12042	0.3396 .3176 .2850 .2452 .2006 .1532 .1284 .0776 .0522 .0392 .0262 .0130 0 -0130 0 -0130 -0262 .0396 -0566 -0786

P <sub>1</sub>	.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
.700 .600 .500 .400 .300 .250 .200 .150 .100	.5034 .2756 .2390 .1968 .1510 .1268 .0768 .0518 .0260 .0130 .0130 .0262 .0394 .0524	72.4466 65.4466 11445.100 1145.100	.0506 .0382 .0254 .0126 0 0128 0258 0390 0520	25050000000000000000000000000000000000	233580666686278844 455820 233580666686278844 455820 233580666666866278844 455820 0 003510 0 003510 0 003510 0 003510 0 003510	189549 195549 196549 19	1836 1814 1610 1298 1110 0694 0478 0359 0120 0	15649 116649 11670 116876 00455548 00118 00118 00118 00118 00118 00118 00118	111085548 111085548 111085548 0013141 1108568 0013141 110868 0013141 110868 0013141 110868 0013141 110868 0013141 1108568 0013141 0013	1274 137018 119714 1197	1162 1042 0924 0776 0604 0418 0318 0216 0 0108 0 - 0126 - 0344 - 03462	0954 0954 09559 0774 0406 0204 0 0108 0 0108 0 0203 0 0203 0 0450	.0852 .0786 .0680 .0546 .0382 .0198 .0100 0 0104 0212 0324 0436

TABLE II

values of 2 
$$\left(\frac{\rho_1}{\rho_2}\right)^{1/2}$$
 for determining point drag coefficient for flows wherein energy is added

 $\left[\mathbf{m}_{\mathbf{o}}=\mathbf{0}\right]$ 

P <sub>1</sub>	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700 .400 .300 .200 .150 .075 .050 .075 .050 .075 .050 .075 .050 .025 .075 .050 .050 .050 .050 .050 .050 .05	1 1 1 1 1 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2	1.6124 1.7320 1.8440 1.9494 2.0494 2.1448 2.1360 2.3874 2.3874 2.3874 2.4290 2.4490 2.5298 2.4490 2.5298 2.4490 2.5298 2.4490 2.5298 2.4490 2.5298 2.4490 2.5298 2.7228 2.7228 2.7228 2.7228 2.7228	1117807688 5478874068860428 547887406888 547887406888 547887406888 547887406888 547887406888 547887406888 5478874088 54788748 5478	111111200974408025674494008060808080808080808080808080808080808	1.5492.1 1.5492.1 1.67888 1.6976.8 1.6976.8 1.6976.8 1.6976.8 1.6976.8 1.9036.8 1.90	1.4824 1.	1.1.5478.8 1.54	1.1832 1.3416 1.16124 1.76320 1.8974 1.7320 1.8974 1.7320 1.8974 1.8974 1.8974 1.8974 1.8988 1.22360 1.8988 1.22360 1.8988 1.22360 1.8990 1.8900 1.89	1.2650 1.	1.184524 1.184524 1.184524 1.18524 1.1	0.04544 0.2654 1.2673 1.5673 1	0.77\6 1.0000 1.0832 1.3\1612\1.4832 1.7320 1.7320 1.7320 1.87\8 1.89\6 1.92\6 1.92\6 2.02\6 2.02\6 2.02\6 2.02\6 2.32\6 2.32\6 2.32\6 2.569\6	0.6324 1.0954 1.2454 1.2454 1.4142 1.6734 1.6734 1.7320 1.7888 1.8140 1.8764 1.9236 1.9236 1.9748 2.0004 2.0148 2.0148 2.0004 2.0148 2.0004 2.0148 2.0004 2.0148 2.0004 2.0148 2.0004 2.0148 2.	0.4472 1.0002 1.18416 1.3416 1.4832 1.5492 1.5720 1.7888 1.8740 1.7888 1.8140 1.89236 1.89236 1.9906 2.23284 2.23284 2.24090
800	3.0332	3.0000	2.9664	2.9326	2.8982	2,8636	2.8284	2.7928	12.7560	1	}	1	1	7
4° TH	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80	-
0.800 .700 .500 .500 .250 .250 .150 .071 .051 .021 0 .021 .051 .051 .051 .051 .051 .051 .051 .05	.632 .894 1.095 1.265	7746	.6324 .894 1.095	7740 4 1 0000	6 .632 894	111.000	894	4 774	0 6321 7714 0 951 1 1095 1	0 1472 6774 1 0048 2 1 1048 2 1 1048 3 1 1245 3 1 1 245 3 1	7.72.4 6.77.4 7.72.4 9.40.8 9.40.8 9.40.8 1.1.8 1.8	2 0 14 6 6 7 7 4 6 6 6 7 7 3 6 6 6 7 7 3 6 6 6 7 7 3 6 6 1 2 6 7 1 1 1 6 7 2 1 1 1 1 6 7 2 1 1 1 1 6 7 2 1 1 1 1 6 7 2 1 1 1 1 6 7 2 1 1 1 1 6 7 2 1 1 1 1 6 7 2 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	20 1, 1, 1, 1, 2, 2, 3, 4, 1, 1, 1, 1, 1, 1, 1, 1, 1, 1, 1, 1, 1,	04:064:0004:0004:0004:0004:0004:0004:00

TABLE II - Continued

VALUES OF  $2\left(\frac{\rho_1}{\rho_2}\right)^{1/2}\left(\frac{q_1}{q_0}\right)^{1/2}$ 

FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $M_0 = 0.05$ 

						l	$M_0 = 0$	05]						
AH q <sub>o</sub>	-0.50	"							-0.10	-0.05	0	0.05	0.10	0.15
.400 .3250 .1260 .150 .075 .075 .025 0 .025 050 075	2.3860 2.4066 2.4272 2.4480 2.4682 2.4884 2.5082	1568860 1110000000000000000000000000000000	2.3008 2.3224 2.3438 2.3652 2.4066 2.4274 2.1182	2.2574 2.2792 2.3010 2.3228 2.3438 2.3652 2.3860	2.2128 2.2352 2.2574 2.2796 2.3010 2.3224 2.3438 2.3654	2.1672 2.1902 2.2126 2.2574 2.2574 2.2792 2.3008	2.1206 2.1140 2.1672 2.1906 2.2128 2.2352 2.2574	2.0730 2.0970 2.1206 2.11111 2.1672 2.1902 2.2126	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	1.349494 1.949494 1.949494 1.97994 1.9794 1.9794 1.9794 1.9794 1.9794 1.9794 1.9794 1.9794 1.9794 1.9794 1.9794 1.9794 1.9794 1.97	1	1.0012 1.1812 1.1814 1.1813 1.1813 1.1813 1.1813 1.1813 1.1813 1.1813 1.1913 1.	8958 1.0966 1.2658 1.4150 1.5500 1.6124 1.6738 1.7316	1.3426 1.4842 1.56130 1.7304 1.7304 1.7884 1.8162 1.8968 1.8968 1.8968
AH qo	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80	
.400 .300 .250 .200 .150 .075 .050 .025 0	.6344 .8960 1.9660 1.4152 1.4836 1.5500 1.6124 1.6738 1.7026 1.7604 1.7604 1.7892 1.8136 1.8704 1.8704	1.1844 1.3426 1.4146 1.4842 1.5494 1.6132 1.6430 1.7026	1.2660 1.3420 1.4152 1.4826 1.5502 1.5812 1.6124 1.6430 1.6740 1.7026 1.7316 1.7316 1.7604 1.7894	1.2654 1.3426 1.4146 1.5168 1.5194 1.5812 1.6134 1.6430 1.7026 1.7326	8960 1 0968 1 1838 1 2662 1 3420 1 4154 1 4492 1 15502 1 5812 1 6428 1 6740	1.0014 1.0962 1.1846 1.2654 1.37788 1.4146 1.4494 1.5494 1.5494 1.5494	.8960 1.0008 1.0970 1.1838 1.2662 1.3042 1.3786 1.4154 1.4500 1.4834 1.4834 1.5502	. 7764 . 8956 1.0016 1.0962 1.2852 1.2654 1.34408 1.3786 1.3446 1.4446 1.4444	6346 .7760 .8962 1.0008 1.1414 1.1870 1.2254 1.2662 1.3042 1.3788 1.4156	.7766 .8956 1.0016 1.0496 1.0960 1.1846 1.2252 1.2524 1.3042	.7760 .8962 .9498 1.0008 1.0496 1.1406 1.1836 1.2252 1.2664	7766 8376 8954 9498 1.0006 1.0494 1.0960 1.1408	0.14196 6348 7786 7786 8962 9498 1.0008 1.0972 1.18364 1.15864 1.5764 1.7898 1.7898 1.7898 1.7898	

TABLE II - Continued

VALUES OF 2 
$$\left(\frac{\rho_1}{\rho_2}\right)^{1/2} \left(\frac{q_1}{q_0}\right)^{1/2}$$

FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[\mathbf{M}_{o}=0.1\overline{0}\right]$ 

N =				····										
AH Qo	<b>-0.50</b>	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20			-0.05	0	0-05	0.10	0.15
.025 0 025 050 075 100 150	2.4234 2.4438 2.4640 2.4838 2.5036 2.5236 2.5622	2.3826 2.4032 2.4238 2.4440 2.4644 2.5238	2.3408 2.3620 2.3828 2.4036 2.4242 2.4446 2.4848	2.2984 2.3200 2.3416 2.3626 2.4038 2.4038 2.4038	2.2554 2.2770 2.2986 2.3202 2.3414 2.3626 2.4042	2.2110 2.2334 2.2556 2.2774 2.2990 2.3206 2.3630	2.1660 2.1888 2.2112 2.2336 2.2560 2.2776 2.3208	2.1200 2.1432 2.1662 2.1890 2.2116 2.2340 2.2780	2.0726 2.0966 2.1202 2.1434 2.1664 2.1892 2.2342	2.0244 2.0490 2.0730 2.0970 2.1238 2.1438 2.1896	1.9750 2.0000 2.0246 2.0492 2.0732 2.0972 2.1440	1.9242 1.9498 1.9752 2.0000 2.0248 2.0494 2.0974	1.8718 1.8982 1.9244 1.9500 1.9754 2.0006 2.0498	0.45842 7812 1.08742 1.18452 1.14528 1.14528 1.14528 1.16154 1.1615
P <sub>1</sub>	0.20	0.25	0.30	0:35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80	
.075 .050 0 .025 050 075 100 150 200 300 400	.6408 9002 1.1000 1.2688 1.4176 1.4864 1.5522 1.6756 1.7052 1.7342 1.7342 1.7342 1.7342 1.7342 1.7342 1.7342 1.7342 1.7342 1.7342 1.8456 1.8456 1.8756 1.8756 1.8756 2.0010 2.0502 2.0982 2.1908	1.05h 1.18784 1.41642 1.41644 1.647584 1.647584 1.73428 1.73428 1.73984 1.8158 1.7984 2.00104 2.00104	.408 1.1002 1.262 1.3456 1.4178 1.4188 1.5242 1.6166 1.6760 1.7348 1.7632 1.7348 1.7632 1.7912 1.910 1.8992 1.9510 2.0018	0.1586 .7816 1.0058 1.189 1.3458 1.4180 1.5202 1.5526 1.6158 1.6158 1.6764 1.7914 1.9914 1.9514	9006 1 1004 1 1082 1 2694 1 3460 1 4182 1 4530 1 4530 1 5530 1 5646 1 6164 1 7752 1 7916 1 8996 2 0020	1.0060	-6412 -9010 1.0060 1.1088 1.2698 1.3086 1.4534 1.4534 1.5206 1.5534 1.6768 1.7520	0.4588 .7818 .9010 1.0062 1.1062 1.1086 1.2302 1.2708 1.3464 1.4574 1.4574 1.4574 1.4574 1.6166 1.4574 1.6166	.7820 .9012 1.0064 1.1010 1.1458 1.1888 1.2702 1.3702 1.3852 1.1190 1.4878 1.5538 1.6774 1.7924	.6414 .7820 .9012 1.0064 1.0548 1.1014 1.1458 1.1888 1.2304 1.2702 1.3090 1.3468 1.4190 1.4880 1.6170 1.7362		0 .4590 .6416 .7822 .8440 .9016 .9556 1.0068 1.0552 1.1016 1.1462 1.2706 1.3472 1.41884 1.6174	7156 7824 8916 9016 9558 1.0056 1.1016 1.2708 1.3472 1.4196	

TABLE II - Continued

VALUES OF  $2\left(\frac{\rho_1}{\rho_2}\right)^{1/2}\left(\frac{q_1}{q_0}\right)^{1/2}$ 

# FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $M_0 = 0.15$ 

						M	o = 0.1	5]						
AH Qo	-0.50	-0.45	-0.′.	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700 .500 .500 .500 .250 .250 .250 .255 .055 .0	1.99464 999464 999464 99964 99646 99646 99646 99646 99646 99646 99	2.3766 2.4376 4.4376 4.55746 2.445746 2.555324 2.2222 2.2222	111828282828458882 1118882828458882 11188827914588 11188827914588 11188827914588 111888774888 111888774888 111888774888 111888774888 111888774888 111888774888 111888774888 111888774888 111888774888 111888774888 111888774888 111888774888 111888774888 111888774888 111888774888 11188877488 111888774888 1118887748 11188877488 111888 11888 11888 11888 11888 11888 118	13440082888028240824089666666666666666666666666666666666666	11111100010000000000000000000000000000	11.184240 11.1947700434000 11.19477000465000 11.1000765000000600000000000000000000000000	1-1-1-1-1-1-1-1-1-1-1-1-1-1-1-1-1-1-1-	2.0948 2.1184 2.1412 2.1868 2.2314 2.2315 2.2315 2.3160 2.3600 2.401	1.279498 1.45756 1.457	11.10000 11.40000 11.	1.148 1.12724 1.15736 1.15736 1.57350 1.7915 1.7915 1.995 1.	1.4888 1.617546 6.77546 8.7754	11111111111111111111111111111111111111	1.5558 1.617848 1.7548 1.7548 1.7548 1.7548 1.7548 1.7548 1.7548 1.7548 1.7548 1.75518
P <sub>1</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80	
0.800 .700 .600 .500 .400 .250 .250 .150 .075 .050 .025	9740 9740 9740 9740 9740 9740 9740 9740	0.47898 7013002 11135022 11135022 11135022 11135022 1113502 11	1.9016 1.9534 2.0034 2.1000 2.1922 2.2806 2.3660	1.8492 1.9022 1.9536 2.0528 2.1472 2.2376	9084424596909999999999999999999999999999999999	1.7392 1.7954 1.8502 1.9548 2.0540 2.1484 2.2388	1.6812 1.7398 1.7960 1.9038 2.0056 2.1022 2.1948	1.6214 1.6820 1.7402 1.8514 1.9558 2.0552	1.5590 1.6220 1.6824 1.7970 1.9048 2.0068	0.4732 6524 790942 1.0142 1.1088 1.11532 1.2370 1.237524 1	1.4942 1.4948 1.5598 1.7980 1.7958 2.0078	1.4262 1.4262 1.4946 1.6232 1.7420 1.8580		

TABLE II - Continued

values of 2  $\left(\frac{\rho_1}{\rho_2}\right)^{1/2} \left(\frac{q_1}{q_0}\right)^{1/2}$  for determining point drag coefficient - continued

						[M <sub>o</sub>	= 0.20							
AH q <sub>o</sub>	-0.50	-0.45	-0.Ho	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700 .500 .100 .250 .150 .050 .025 .075 .100 .250 .250 .250 .250 .250 .250 .250 .2	1.6788968 1.783968 1.1988968 2.17586 2.17586 2.2305648 2.242620 2.3358070 2.2450568 2.245068 2.2	220 220 220 220 220 220 220 220 220 220	1.678566 607856660 7.67856660 9.9873780 9.9873780 1.112222288748 9.9873788 9.987378	1.48740 1.6140 1.7310 1.94312 2.03440 2.13349 2.12349 2.12349 2.12349 2.26676 2.26676 2.3550 1,47086 6,6763 2.26676 2.	1.4212 1.55424 1.55424 1.78940 2.08952 2.18040 2.224267466 4.674666 2.224267466 2.224267466 2.224267466 2.224267466 2.224267466 2.224267466 2.224267466 2.224267466 2.224267466 2.224267466 2.2242674666 2.2242674666	1.4526 5.508 6.1526 1.61526	11.45758 2.25758 21.57	11111111111111111111111111111111111111	1.1108 1.2770 1.4234 1.5558 1.7970 1.89782 1.99824 1.99824 1.99824 2.07936 2.07936 2.07936 2.13942 2.13942 2.22142 2.22142 2.22142 2.23356 2.2356 2.2356 2.3556 2.3556 2.3556 2.3556 2.3556 2.3556 2.3556 2.3556 2.3556 2.3556	1.0180 1.1972 1.39218 1.49188 1.73268 1.73258 1.73458 1.9744 1.9744 1.9934 1.9934 1.9746 1.23736 1.237	0.9156 1.1120 1.2748 1.55792 1.55792 1.673792 1.673792 1.89250 1.97590 1.97590 1.97590 1.97590 2.0722 2.0722 2.1418 2.27584 2.27584 2.25936 2.2670	0.7998 1.01984 1.35934 1.35934 1.68006 1.68006 1.68006 1.79368 1.8742 2.02524 1.9968 1.9968 1.9968 2.02524 2.14806 2.25318	0.6636 9168 1.11794 1.4262 1.5512 1.6810 1.7390 1.8220 1.87590 1.8452 1.9012 1.9772 2.0508 2.1492 2.1492 2.25062 2.55062	0.4906 1.0206 1.0296 1.34948 1.55220 1.7455920 1.779530 1.782396 1.782396 1.782396 1.782396 1.9923314 1.99234 1.9924 1.99234
P <sub>1</sub> ΔH <b>Q</b> <sub>0</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0,50	0.55	0.60	0.65	0.70	0.75	0.80	
050 075 100 150 250 300 500	.6644 .9174	0. 1910 1.0212 1.2008 1.3564 1.15612 1.6236 1.6540 1.6136 1.7114 1.7972 1.8248 1.7912 2.0050 2.0556 2.11768	6,650 9,154 1,152 1,3574 1,4290 1,4970 1,5936 1,6844 1,7124 1,7704 1,7982 1,9562 2,1936 2,1936 2,2314 2,2314 2,2314	0.4916 .80222 1.20222 1.2026 1.2826 1.2582 1.4298 1.55028 1.55028 1.55028 1.6252 1.6252 1.7146 1.7452 1.7452 1.7452 1.7452 1.7558 2.1594 2.2558 2.2558 2.2558	6656 9192 1.1164 1.2026 1.2834 1.4504 1.4504 1.4986 1.5314 1.6566 1.7442 1.6666 1.7442 1.8650 1.8650 1.9670 2.1042 2.1042	0. 1920 .8030 1.1273 1.1273 1.2840 1.3596 1.4312 1.4856 1.4794 1.5322 1.5640 1.8554 1.8554 1.8554 2.2556 2.2556 2.2556 2.2556	9202 1.0236 1.2038 1.2846 1.3246 1.32604 1.3364 1.4864 1.5330 1.5330 1.6672 1.6276 1.6878 1.7458 1.7458 1.9088 2.0102 2.1062	1.4326 1.4672 1.5008 1.5660 1.6286 1.6886 1.7468 1.9612 2.05988	.6670 .8042 .9210 1.0246 1.1186 1.1624 1.2052 1.2858 1.3242 1.3616 1.3978 1.4334 1.5016 1.6296 1.6898 1.9108 2.0182	0.4670 .6674 .9216 .9216 1.0252 1.0752 1.11632 1.2468 1.3250 1.2468 1.3250 1.5676 1.6302 1.7484 1.8590 1.9630	1.2064 1.2474 1.2870 1.3630 1.4348 1.5682 1.6912 1.8052	1.3638 1.4354 1.5038 1.6318 1.7502 1.8608	0.2016 .4936 .6682 .7404 .8058 .8664 .9229 1.0266 1.0746 1.1208 1.2084 1.3642 1.4362 1.4362 1.6300 1.6300	

TABLE II - Continued

VALUES OF  $2\left(\frac{\rho_1}{\rho_2}\right)^{1/2}\left(\frac{q_1}{q_0}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

0.80
0.2538
.5190 .6882
.7586
.8232
.8830
.8830 .9392
.9918
ไว ผู้ผู้เวลิไ
1.0894
1.1350
1.0894 1.1350 1.2212 1.3014
1.3014
1.3770
1.5816
1.7038
1.8176
1.9214
2.0252
+ 0:0:0

TABLE II - Continued

VALUES OF  $2\left(\frac{\rho_1}{\rho_2}\right)^{1/2}\left(\frac{q_1}{q_0}\right)^{1/2}$ 

FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $M_0 = 0.30$ 

						בי	,	J						
P <sub>1</sub>	-0.50	-0.45	<b>-</b> 0.40		-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700 .500 .400 .250 .250 .250 .250 .075 .050 .075 .075 .0750 .0750 .0750 .0750 .0750 .0750 .0750 .0750 .0750 .0750 .0750	119.05686488041060268876514886488041060268876514876987651548769876515487698765148769876514876987651487698765148769876514876987651487698765148769876514876987651487698765148769876514876987651487698765148769876514876987651487698765148769876514876987651487687651487687651487687651487687676876768767687676876768767687676	119215894430008 8331689443008 11921589443008 1192158944502 1192154949 129212 12921 129212 129	11122222222222222222222222222222222222	1.4920 1.6150 1.7356 1.83558 2.0362 2.2280 2	111.75550 5650 5650 5650 5650 5650 5650 5650	1111190081546 11111190081546 11111190081546 11111190081546 11111190081546 11111190081546 11111190081546 11111190081546 11111190081546 1111119081546 1111119081546 1111119081546 1111190816 11111908 11111908 11111908 11111908 11111908 11111908	11111111111111111111111111111111111111	11111111111111111111111111111111111111	11111111111111111111111111111111111111	14644415647924673888848879245888879299999999999999999999999999999999	11111111111111111111111111111111111111	11111111111111111111111111111111111111	11111111111111111111111111111111111111	250060478864888888888888888888888888888888
P <sub>1</sub>	0.20	0.25	0.30	0.35	0.1.0	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80	]

P <sub>1</sub>	.20 0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
0.800 0.700	7028 7030 0.541; 9146 1834 1370 1.047; 3000 1.222; 1442 1.375; 5716 1.512; 5716 1.512; 66944 1.637; 77231 1.667; 77231 1.667; 77231 1.696; 87326 1.753; 8853 1.834; 8850 1.834; 9606 1.892; 9092 1.963; 9106 1.861; 9606 1.892; 9092 2.1963; 9778 2.258; 9778 2.258; 9778 2.258; 9778 2.258; 9778 2.258;	70418 914766 91476 914766 9147	1.2243 2.5436 2.5436 2.5490 2.5555 2.5555 2.5556 2.5556 2.5570 2.	70498288 7949288 7949288 794928 794928 794929 79492 7	0 111111111111111111111111111111111111	7618 08 66 20 07 09 50 50 50 50 50 50 50 50 50 50 50 50 50	0.54546081208465544674806556605592667556605566055660556605566055660556605	79266 2 57226 78457572682 57214 78457572682 57214 7845757682 5721 784575778 5721 78457578 5721 784578 5721 7	9494555389696996669979666997966699796669979666997966699796669979666997966699796669979669979699796999999	784508 794508 90508 1149345 1149345 1149345 1149345 114934 114	0.54742 -547321 -547321 -577336 -9577356 -9577356 -105590 -	54.83 -78.120 -84.43 -95.80 -95.80 -95.80 -95.80 -10.15.87 -10.123

TABLE II - Continued

VALUES OF  $2 \left(\frac{\rho_1}{\rho_2}\right)^{1/2} \left(\frac{q_1}{q_0}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

						[x	o = 0.3!	5]				. •		
AH qo	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	o ´	0.05	0.10	0.15
700 -500 -500 -500 -500 -500 -250 -050 -0	2.118526 2.118526 2.12656 3.22656 3.22656 3.22656 3.2265 3.3566 3.2266 3.3566 3.3666 3	1.826408 1.826408 1.826408 1.826408 1.82696 1.	11.7788 6761088 11.9741682 11.904	11.6762 6762 6762 6762 6762 6762 6762 6762	9460888884888888989898989898989898989898989	1119963222211111111111111111111111111111	11.5638 82 11.56308 82 11.56308 82 11.56308 83 11.56308 83 11.5630	111785420 111785420	1116876 111	256902 2579902 2579902 2579902 2579902 2579902 25799602 257	1545847088880440044440460 154584708888044003449460 1565740858090500340905091400334946 00000000000000000000000000000000000	11111111111111122222222222222222222222	1.4504 1.45788 1.6588 1.6968 1.8066 1.8088 1.8888	8526 11.23384 11.5884 11.5884 11.5884 11.5884 11.5884 11.5884 11.5884 11.5884 11.8874
P <sub>1</sub>														7

P <sub>1</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
0.800 .700 .500 .400 .250 .250 .075 .025 .050 .050 .150 .250 .150 .250 .250 .250 .250 .250 .250 .250 .2	9.15.24.66.60.22.66.60.22.66.60.22.66.60.22.66.60.22.66.60.22.66.60.22.60.60.22.60.60.22.60.60.22.60.60.22.60.60.22.60.60.60.60.60.60.60.60.60.60.60.60.60.	0.5730	7.968,1468,279,151698,279,151698,279,1698,2798,2798,2798,2798,2798,2798,2798,27	0.5466 1.02486 1.024986 1.024986 1.024986 1.02498 1.02	9688460 9154204434489 154204434499 1542043346667 162336667 162336667 1623267 1	0 111111111111111111111111111111111111	79768264884884848489676482648888844848968687677883448888896648874883448888448888888888888888888888	0.586724 9.774280 9.774280 1.21880	8640 10752 107660 112500 112500 112500 113660 114762 115628 11562	86548 876848 9768866 1012676488 10126910488 10126910488 10126910488 1012691048 101	57868 78684 972886 972886 11112288 972886 11111288 972886 11111288 97288 97288 11111111111111111111111111111111111	0.146879888 358786257888820045454807645750014124548074547678096745457809680	0.3606 57398 80792 983121 983212 1.175766 1.175766 1.175766 1.175766 1.175766 1.175766 1.175766 1.175766 1.175766

TABLE II - Continued

VALUES OF 2  $\left(\frac{\rho_1}{\rho_2}\right)^{1/2} \left(\frac{q_1}{q_0}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

[Mo = 0.40]

AH									-						
0.800	AH .			•		_									
	0.800 .700 .600 .500 .100 .250 .150 .150 .075 .050 .075 .050 .075 .050 .075 .050 .075 .050 .075 .050 .075 .050 .075 .050 .075 .050 .075 .050 .075 .050 .075 .050 .075 .050 .075 .050 .075 .050 .075 .050 .075 .050 .075 .050 .050	1. 7656 1. 8642 1. 9568 2. 01486 2. 1286 2. 1286 2. 1297 2. 2287 2. 23406 2. 3038 2. 3588 2. 3770 2. 4980 2. 5512 2. 4980 2. 5512 2. 5644 2. 5690 2. 2590 2. 2	1.6198 1.9158 2.90920 2.1736 2.2136 2.2136 2.22714 2.22714 2.2346 2.2346 2.2346 2.2346 2.2346 2.2346 2.236 2	1.7732 1.8732 1.96538 2.0538 2.0538 2.1388 2.1378 2.2376 2.2376 2.2376 2.2351 2	1 . 8278 1 . 9242 2 . 01508 2 . 10150 2 . 1030 2 . 1030 2 . 1234 2 . 2234 2 . 2234 2 . 2334 2 . 2334 2 . 2334 2 . 2334 2 . 2343 2 . 2334 2	11.7814 1.7814 1.7814 1.9714 12.0154 12.0166 12.1066 12.2286 22.11684 22.1286 22.2286	1.6226 1.7330 1.9326 1.9326 2.02478 2.10324 2.11316 2.11324 2.11334 2.12334 2.22334 2.2344 2.234 2.234 2.2344 2.23	1.6824 1.6824 1.7898 1.9382 2.0724 2.1362 2.13762 2.13598	1.52982 562982 562982 57435 57435 59324	1.1,944,6,6,6,7,9,1,1,1,1,1,1,1,1,1,1,1,1,1,1,1,1,1,1	1.53464666 .551646666 .561646666 .561646666 .5916266666666666666666666666666666666666	1.1650 1.45808 1.75808 1.75914 1.8559 1.8559 1.8959 223768 20348 20348 22121778 22121778 22121778 22121778 22121778 2212178 2212178 2212178 2212178 2212178 2212178 2212178	1.2492 1.5292 1.5292 1.6400 1.7508 1.8592 1.8092 1.8092 1.8093 1.9094 1.9040 2.0944 2.1370 2.1370 2.1370 2.1370 2.1370 2.1370 2.1370 2.15550	1.5426 1.4610 1.5466 1.6466 1.7036 1.8628 1.8880 1.9512 1.9612 1.	1.2480 1.3960 1.5906 1.5906 1.6502 1.7072 1.7624 1.7890 1.8156 1.8672 1.8920 1.9170 1.9414 1.9652 2.0584 2.1470 2.2316 2.3120 2.3896
													7	т	7

P <sub>1</sub>			•										
Δ <del>I</del>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
.700 .600 .500 .400 .300 .250 .075 .050 .025 .025 .025 .025 .025 .025 .02	9884 11.3268 11.3268 11.55918 11.55918 11.57184 11.7766 11.77692 11.78192 11.8760 11.8960 11.90168 11.90168 12.00168 12.00168 12.00168 12.00168	1 .55768 1 .55768 1 .55768 1 .55768 1 .55768 1 .75769 1 .7769 1 .7769	7,5682 7,9873244280462804280428042804280428042804280800000000	0.60922 1.257492 1.2574972 1.4775 2.57492 1.4775 1.4775 1.4775 1.4777 1.4777 1.4777 1.48822 1.4981 1.4981 1.2498 1.2498 1.2498	.79928 1.1782 1.2674 1.263740 1.41780 1.41780 1.514608 1.514608 1.7834 1	1.0914 1.1804 1.2628 1.3426 1.4476 1.4476 1.5448 1.554788 1.6404 1.7636 1.7636 1.8896 1.8896 1.8896 1.8896 1.8896 1.8896 1.8896 1.8896 1.8896 1.8896 1.8896 1.8896 1.8896 1.8896 1.8896 1.8896	.9960 1.0934 1.1826 1.2626 1.2626 1.4754 1.4754 1.5502 1.6732 1.6732 1.7872 1.6132 2.0414 2.1334	88490 9978 1.0850 1.38678 1.3678 1.38828 1.4874 1.4874 1.5253 1.46766 1.6766 1.6766 1.6974 1.9974	. 9966 1.0976 1.1872 1.2794 1.3096 1.33848 1.4560 1.4560 1.4560 1.4560 1.4560 1.4560 1.4504 1.61800 1.4504 1.61800 1.4504 1.9524 1.9524 2.0194	0.6162 .767246 1.0916 1.09986 1.1894 1.1894 1.2328 1.35276 1.4232 1.55224 1.55224 1.68378 1.79046 2.0052	6174 76840 1.00348 1.00348 1.1078 1.1018 1.12746 1.27546 1.3146 1.35260 1.5625 1.74458 1.74458 1.74458 1.755942	0.4148 6186 7698 .9520 1.0054 1.00558 1.1040 1.1236 1.1498 1.2774 1.3562 1.4988 1.5652 1.6896 1.8046 1.9120	0.4156 .6196 .7192 .8368 .8972 .9538 1.0072 1.1060 1.1522 1.2798 1.3582 1.4518 1.4518 1.5018

TABLE II - Continued

Values of  $2\left(\frac{\rho_1}{\rho_2}\right)^{1/2}\left(\frac{q_1}{q_0}\right)^{1/2}$  for determining point drag coefficient - continued

 $\left[\mathbf{x}_{0}=0.45\right]$ 

AH Q <sub>0</sub>	-0.50										_	0.05	0.10	0.15
.500 .400 .250 .250 .100 .075 .050 .025 .050 .075 .050 .025 .050 .025 .050 .025 .050 .025 .050 .025 .050 .025 .050 .025 .050 .025 .050 .025 .050 .025 .050 .050	1.6562 1.7586 1.8586 1.8546 2.0300 2.1596 2.1596 2.2268 2.2268 2.2268 2.2334 2.3340 2.33520 2.33520 2.35520 2.	1 1 9058 1 1 9054 1 1 9054 1 1 1 1 568 2 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	1.8652 1.9558 2.0420 2.0420 2.1240 2.1240 2.1240 2.2208 2.2208 2.2258 2.	1 2288 1 9166 1 9166 2 00478 2 0894 1 1694 1 1695 1 1695 1 2084 2 224 2 244 2 254 2	1.6756 1.9756 1.9670 2.01360 2.0536 2.1360 2	1.625 1.7356 1.7376 1.9726 2.0166 2.1016 2.1016 2.1026 2.1622 2.1622 2.1622 2.2506 2.2536	1 1 68 62 1 1 68 62 1 1 78 66 1 1 9 7 62 1 1 9 7 62 1 1 9 7 62 1 1 9 7 62 1 1 1 88 62 1 1 1 88 62 1 1 1 88 62 1 1 1 8 62 1 8 62 1 8 62 1 8 62 1 8 62 1 8 62 1 8 6	1.5388 1.7428 1.8916 1.8938 1.8938 1.8938 1.9288 2.0493 2.1546 2.2720 2.1546 2.	11578 115836 116936 118946 118946 119931 11990 11900 1	1.3956 1.5248 1.5424 1.6424 1.8554 1.8554 1.9018 1.9490 1.9746 2.0392 2.0392 2.0392 2.0392 2.0392 2.2462 2.2462 2.2462 2.2462	1.3292 1.4654 1.7026 1.7026 1.7026 1.8080 1.8080 1.8582 1.9506 1.9540 1.9772 2.0002 2.0002 2.0002 2.0002 2.0008 2.	1.2582 1.4026 1.5326 1.5326 1.7670 1.7608 1.7608 1.8832 1.9358 1.9358 1.9352 2.0278 2.0278 2.0278 2.2378 2.2378	1.022 1.18262 1.47376 1.5958 1.79558 1.7165 1.76558 1.7166 1.8834 1.88370 1.9446 1.9972 1.997	.9006 1.0988 1.2646 1.4100 1.5101 1.5014 1.6014 1.7702 1.7766 1.7702 1.7766 1.7702 1.7966 1.8978 1.9222 1.9496 2.0610 2.1050 2.1178 2.2306

AH Qo	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
.700 .500 .300 .250 .250 .100 .050 .050 .050 .050 .050 .050 .0	7836 1.0078 1.0078 1.1878	1.1052 1.2710 1.4172 1.5486 1.5486 1.6686 1.6686 1.7248 1.7241 1.7251	0.4606 .7878 1.0122 1.1936 1.4208 1.4208 1.4882 1.5526 1.6728 1.6728 1.7014 1.7704	9994 112774 112774 1132530 114920 115874 115874 116478 116766 117614 117688 117614 117688 117614 118926 119902 21902	0.4634 1.79172 1.19808 1.1,2808 1.1,2938 1.1,2938 1.1,2938 1.5218 1.5218 1.5218 1.5218 1.5218 1.5218 1.5218 1.5218 1.5218 1.7002 1.7036 1.70	6504 91466 111528388 11283898 11283898 11283898 11283898 11283899 1128389 1128	1.0222 1.1178 1.2056 1.2870 1.3632 1.4550 1.4550 1.5030 1.5030 1.5030 1.5030 1.6896 1.6896 1.6856 1.6856 1.6856	.6536 .9184 1.0214 1.1208 1.1208 1.2909 1.3288 1.3684 1.4030 1.13884 1.5654 1.5718 1.5718 1.5719 1.5	0.4672 .7992 1.0270 1.1230 1.2212 1.2232 1.2232 1.3528 1.3528 1.3568 1.4766 1.5778 1.5778 1.5778 1.5868 1.5868 1.9668 1.9668	.6566 .8010 .9226 1.0296 1.1258 1.1710 1.2112 1.2560 1.2560 1.3352 1.3730 1.4054 1.5102 1.5102 1.5102 1.7020 1.8152	10.4601.1	. 4764 . 6596 . 9268 . 9268 . 9268 . 9268 . 1034 1 . 1764 1 . 1720 1 . 2606 1 . 2616 1 . 3794 1 . 4522 1 . 5870 1 . 7100 1 . 8240	6610 8066 8790 9844 10366 11336 1136 11366

TABLE II - Continued

VALUES OF 2  $\left(\frac{p_1}{p_2}\right)^{1/2} \left(\frac{q_1}{q_0}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[\mathbf{M}_{0} = 0.50\right]$ 

P <sub>1</sub>	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700	1.6504 1.7498	1.6040 1.7072 1.8038	1.5546 1.6626 1.7626	1.5026 1.6154 1.7198	1.4474 1.5658 1.6744	1.3884 1.5130 1.6268	1.3258 1.4574 1.5766	1.2584 1.3978 1.5232	1.1856 1.3348 1.4670	1.1066, 1.2668 1.4072	1.0194 1.1934 1.3434	0.9222 1.1136 1.2750	0.8110 1.0260 1.2010	0.6796 .9280 1.1208
.500 .400	2.0130	1.9796	1.9446 2.0280	1.9080	1.8700 1.9586	1.8298	1.7878	1.7438 1.8424	1.6976	1.6490	1.5978	1.5436 1.6600	1.4864	1.4258
.200 .150 .100 .075	2.1666 2.2030 2.2384 2.2558	2.1376 2.1748 2.2112 2.2292	2.1072 2.1456 2.1830 2.2014	2.0356 2.0758 2.1152 2.1536 2.1726 2.1910 2.2094	2.0428 2.0842 2.1228 2.1422	2.0082 2.0504 2.0910 2.1108	1.9726 2.0156 2.0576 2.0780	1.9352 1.9796 2.0228 2.0438	1.9418 1.9864 2.0082	1.9026 1.9484 1.9710	1.8614 1.9090 1.9322	1.8182 1.8676 1.8916	1.7732 1.8242 1.8492	1.7260 1.7788 1.8048
.050 .025 0	2.2730 2.2900 2.3068 2.3236	2.2468 2.2644 2.2818 2.2990	2.2194 2.2374 2.2552 2.2728	2.1910 2.2094 2.2278 2.2460	2.1614 2.1804 2.1990 2.2174	2.1506 2.1500 2.1692 2.1984	2.1186 2.1384 2.1578	2.0856 2.1058 2.1258	2.0514 2.0720 2.0926	2.0150 2.0368 2.0580 2.0790	1.9776 2.0000 2.0220	1.9386 1.9616 1.9844 2.0066	1.8978 1.9218 1.9450 1.9684	1.8552 1.8800 1.9042 1.9280
050 075 100 150	2.3568	2.3326	2.3076	2.2814 2.2990 2.3334 2.3672	2.2540 2.2720 2.3072	2.2258 2.2440 2.2802 2.3160	2.1962 2.2152 2.2524 2.2888	2.1654 2.1848 2.2230 2.2604	2.1332 2.1532 2.1924 2.2306	2.0998 2.1204 2.1608 2.2000	2.0650 2.0862 2.1278 2.1682	2.0288 2.0506 2.0932 2.1348	1.9906 2.0134 2.0574 2.1002	1.8552 1.8800 1.9042 1.9280 1.9516 1.9748 2.0200 2.0642 2.1072 2.1494 2.2304 2.3074 2.3818
250 300 400	2.4978	2.4462 2.4772 2.5378	2.4238 2.4554 2.5174	2.4004 2.4328 2.4962 2.5574	2.3760 2.4092 2.4740 2.5364	2.3508 2.3848 2.4510 2.5150	2.3244 2.3592 2.4270 2.4924	2.2968 2.3326 2.4020 2.4690	2.2682 2.3048 2.3762 2.4446	2.2386 2.2762 2.3492 2.4194	2.2076 2.2462 2.3212 2.3928	2.1756 2.2152 2.2918 2.3656	2.1420 2.1828 2.2616 2.3370	2.1072 2.1494 2.2304 2.3074
600 700 800	2.6702 2.7242 2.7766	2.6534 2.7082 2.7618	2.6352 2.6912 2.7456	2.6164 2.6738 2.7294	2.5970 2.6554 2.7120	2.5766 2.6364 2.6942	2.5554 2.6164 2.6754	2.5334 2.5956 2.6560	2.5104 2.5740 2.6356	2.4866 2.5516 2.6146	2.4620 2.5284 2.5926	2.5042 2.5696	2.4092 2.4788 2.5458	2.3818 2.4528 2.5212
P <sub>1</sub>			[				T -	I	[					
ΔH q <sub>o</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80	
0.800		0.2410	0.5144	0.2426										
.600	1.0326	1.1276	0.5144 .8210 1.0386	9394	0.5176	0.2440 6918	0.5206	0.2454	0 5276	0.2468				
.300	1.4960	1.4346	1.3692	1.2992	1.2256	1.1414	1.0512	.9506	8356	1 .6996	0.5262	1 5000	10 01.07	
.200	1.6188	1.5636	1.5052	1.3734 1.4434 1.5100 1.5734 1.6040 1.6542	1.3774	1.3070	1.2308	1.1482	1.0572	.9558 1.0598	.8404 .9588	.7034 .8426	.5292 .7054	
100	1.7316	1.6812	1.6286	1.5734	1.5144	1.4522	1.3858	1.3146	1.2380	1.1548	1.0634	.9614 1.0154	.8448 .9062	
.050	1.7848	1.7368	1.6866	1.6542	1.5780	1.5192	1.4566	1.3900 1.4258	1.3186	1.2410	1.2024	1.0662	1.0178	
0	1.8364	1.7904	1.7422 1.7694	1.6920	1.6682	1.5830	1.5238	1.4610	1.4300	1.3610	1.2864	1.2060	1.1176	
050	1.9106	1.8422	1.8222	1.7480	1.7252	1.6736	1.6186	1.5608	1.4996	1.434	1.3648	1.2902	1.2088	
100	1.9814	1.9406	1.8980	1.6632 1.6920 1.7202 1.7480 1.7750 1.8018 1.8538	1.8070	1.7588	1.7076	1.6538	1.5972	1.5372	1.4738	1.4062	1.3320	
250	2.0714	2.0332	1.9938	1.9530	1.9098	1.8654	1.8184	1.7694	1.7178	1.6636	1.6066	1.5462	1.4816	
400	2.1980	2.1634	2.1278	1.8538 1.9040 1.9530 2.0004 2.0912 2.1774	2.0526 2.1416	2.0130 2.1046	1.9714	1.9278	1.8824	1.8348 1.9394	1.7852 1.8934	1.7328	1.7948	
600	15.5320	2 7076	15.2291	2.2598 2.3382 2.4136	2 3066	2 2713	12.21.02	2.2050	2.168	2.1301	8000.5	2.01.96	12.0061	
000	2.4900	12.4072	2.4420	12.4190	2.,,0,0	>	1-17212	1	1	1	1-1-0-1	1	1	,

TABLE II - Continued

VALUES OF  $2\left(\frac{\rho_1}{\rho_2}\right)^{1/2}\left(\frac{q_1}{q_0}\right)^{1/2}$ 

FOR DETERMINING POINT DRAG COEFFICIENT - Continue

 $\left[\mathbf{M}_{\mathbf{0}} = 0.55\right]$ 

K				<del>,</del>	·							,		
AH Q <sub>o</sub>	-0.50	-0-45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	<b>-0-</b> 05	0	0.05	0.10	0.15
0.800 .700 .600 .500 .400 .350 .250 .250 .075 .050 .075 .050 .075 .050 .075 .075	11.8138 11.9936 10.508 10.5	9310488 9310488 964194008 964194008 964118042 974776	2.2486 2.2654 2.	1-1-90 1-1-90	1.673.2 1.773.2 1.794.2 1.994.0 1.994.0 1.994.0 1.994.0 1.008.	1.6286 1.73258 1.95764	1.587662 1.78662 1.92642 1.926	1.5704 1.7454 1.8474 1.8474 1.9320 1.9750 1.9750 1.9750 2.0376 2.0376 2.13568 2.11746 2.2272 2.22873 2.23878 2.318488	11.79488849420 1.590184949420 1.590184949494 1.5901846666645666 1.000184666664566 1.0001846666645666 1.00018466666645666 1.00018466666645666	1.4200 1.5432 1.6558 1.7598 1.8570	1.2126 1.3589 1.69728 1.7658 1.7658 1.7658 1.9562 1.9562 2.02146 1.9562 2.02146 2.02146 2.2329 2.2329 2.3329	1.5556 1.6694 1.7230 1.7746 1.8724 1.8960 1.9420 1.9644 1.9084 2.05140 2.05140 2.1732 2.2122 2.2866	1.220 1.3694 1.6206 1.6206 1.6258 1.7814 1.8558 1.9268 1.9268 1.9744 1.9744 1.9744 1.9744 2.01596 2.1122 2.1818 2.2588 2.2588	9576 1.3030 1.1,248 1.6264 1.6824 1.6824 1.7884 1.8338 1.8338 1.83374 1.9374 2.0678 2.0678 2.1508 2.2302
P <sub>1</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80	
.500 .400 .250 .250 .100 .075 .050 .025 .025 .075 .100 .150 .200 .200 .200 .200 .200	1.7438 1.7698 1.7698 1.8462 1.8462 1.8708 1.9482 1.9488 1.9488 1.9488 2.03758 2.1198 2.1198 2.2198 2.2198	.9648 11.31334 11.45184 11.45184 15.5899 16.6995 17.77624 17.77624 11.4518 11.	80674904936020 24904936020 24904936020 24904936020 259845402060 279886094 2990948624 2990948624 200948	1.16128 1.3256 1.46146 1.5296 1.52920 1.66514 1.6802 1.6802 1.76362 1.79362 1.79362 1.79362 1.9170 1.9170 2.1008 2.18640	1111577444 34457778448 34457778448 34457778448 34444444444444444444444444444444	.9788 1.1702 1.2542 1.2542 1.4058 1.4058 1.4750 1.55410 1.55410 1.57236 1.66368 1.7268	1.1746 1.255748 1.255748 1.4480 1.4480 1.51484 1.51484 1.5778 1.6490 1.6	9864489489 9874849489 9874849 9874849 9874849 987489 98748 97448 9748 97	1.26 1.26 1.36 1.35 1.35 1.35 1.45 1.45 1.45 1.55 1.68 1.79 1.68 1.79 1.68 1.79 1.68 1.79 1.68 1.79 1.68 1.79	.8784 .9924 1.0940 1.1866 1.2302 1.2718 1.3122 1.3512	1.3560 1.3938 1.4308 1.5686 1.6322 1.6936 1.8088	1.2584 1.2806 1.3206 1.32606 1.450638 1.5507380 1.5507380 1.75698	1.2848 1.3650 1.4406 1.5114 1.5792 1.7054 1.829	

TABLE II - Continued

VALUES OF  $2\left(\frac{p_1}{p_2}\right)^{1/2}\left(\frac{q_1}{q_0}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $M_0 = 0.60$ 

P <sub>1</sub> -0.9	0 -0.45	-0.40	-0.35	-0.30	-0.25	-0.20				0	0.05	0.10	0.15
0.800 1.63 .700 1.72 .600 1.81 .500 1.89 .400 1.97 .300 2.07 .200 2.11 .150 2.11 .005 2.12 .050 2.21 .050 2.21 .050 2.21 .075 2.22 .075 2.21 .075 2.31 .075 2.31 .075 2.31 .075 2.31 .075 2.31	26 2 . 0918 27 2 . 1928 2 2 1 1928 2 3 1 1928 2 4 1 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2	12.0684 2.1556 32.1724 32.1856 32.1872 32.1872 32.2576	1.8014 1.8868 1.9674 2.0436 2.0436 2.1588 2.1358 2.1358 2.238 2.238 2.2508 2.2508 2.3744	2.05/2 2.050/2 2.1970/2 2.1120/2 2.1145/2 2.1870/2 2.1870/2 2.23114 2.	2.0662 2.0662 2.0850 3.2.1216 2.1571 2.1750 2.1750 2.2430 8.2.2430 8.2.2430	1.9996 2.0398 2.0777 2.0960 2.1144 2.151 2.151 2.151 2.152 2.259 2.2385 4.2.259 4.2.259 4.2.259 4.2.259	1.9696 2.0104 2.0302 2.0500 12.0694 13.2.0881 13.2.0881 13.2.1446 14.2.1446 14.2.1446 14.2.1446 14.2.1466 14.2.1469 14.2.1469	1.9378 1.9798 2.00014 2.0210 2.0410 2.0608 2.0996 2.1186 2.1714 2.174 2.2156 2.2456	1,9040 1,9476 1,9688 1,9900 2,0106 2,0512 2,0512 2,0712 2,1484 2,1484 2,1485 2,222 2,2276 2,276 2,276 2,276 2,276 2,276 2,276 2,276 2,276 2,276 2,276 2,276	1.8682 1.9134 1.9574 1.9578 2.0000 2.0614 2.0614 2.0614 2.1214 2.1214 2.1214 2.2374 2.2374 2.2374	1.8508 1.8776 1.909 1.9253 1.9453 1.9453 1.9453 1.988 5.2010 2.0512 2.0312 2.0312 2.052 2.	1.2440 1.3882 1.5164 1.6326 1.6326 1.7398 1.7398 1.7398 1.8630 1.8864 1.9324 1.9984 2.0198 2.0198 2.1822 2.1842 2.1823	1.1704 1.3246 1.4606 1.45832 1.6404 1.6952 1.64952 1.79888 1.84726 1.84726 1.84726 1.89588 1.84726 1.9986 1.9986 1.9986

O.20   O.25   O.30   O.35   O.40   O.45   O.50   O.55   O.60   O.65   O.70   O.75   O.80	P <sub>1</sub>												}	
0.800 0.6190 0.4244		ţ	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
025   1.8810   1.8110   1.7986   1.7361   1.7342   1.6324   1.5778   1.5190   1.4556   1.3550   1.2786   1.0010   1.9016   1.8656   1.8656   1.7346   1.7342   1.6324   1.5778   1.5190   1.4256   1.3550   1.2786   1.0010   1.9016   1.8984   1.8066   1.7616   1.7166   1.7166   1.6324   1.6022   1.5522   1.4618   1.2525   1.4618   1.2526   1.2786   1.7886   1.7418   1.6926   1.6100   1.5846   1.5522   1.4618   1.2786   1.2786   1.7886   1.7418   1.6926   1.6100   1.5846   1.5522   1.4618   1.2786   1.2786   1.7886   1.7886   1.7886   1.7418   1.6928   1.6470   1.5910   1.5312   1.4670   1.3998   1.9958   1.9958   1.9958   1.8920   1.8824   1.8824   1.8824   1.5276   1.4736   1.4736   1.6026   1.4736   1.4736   1.4736   1.6026   1.4736   1.4736   1.4736   1.6026   1.4736   1.4736   1.4736   1.6026   1.4736   1.4736   1.6026   1.4736   1.4736   1.6026   1.4736   1.4736   1.6026   1.4736   1.4736   1.6026   1.4736   1.4736   1.6026   1.4736   1.4736   1.6026   1.60	0.800 -700 -600 -500 -500 -300 -250 -150 -107 -051 -051 -051 -051 -051 -051 -051 -051	0.6190 .8874 1.8890 1.2558 1.4010 1.5504 1.5904 1.7662 11.7762 11.8565 11.8824 1.8824 1.8810 1.9046 51.928 01.995 001.995 01.995	0.4244 .7690 .9982 1.1810 1.34648 1.5368 1.5556 1.712 1.7380 1.7644 1.7764 1.8110 1.8898 1.8110 1.8898 1.8136 3.9363 2.0054	0.6242 8952 1.0964 1.4134 1.4804 1.5444 1.6050 1.6632 1.7760 1.7760 1.7786 1.8444 1.8742 1.8742 1.8742	0.1280 .7756 1.0066 1.910 1.3482 1.1198 1.6120 1.6116 1.6701 1.620 1.7536 1.7806 1.7806 1.8322 1.8322 1.8322 1.8322 1.8322	0.6294 9028 1.1080 1.2778 1.3546 1.4256 1.4936 1.5890 1.6192 1.7616 1.7342 1.7616 1.7866 1.8406 1.8406	0.4314 7820 1.0152 1.2012 1.2832 1.4300 1.4998 1.5262 1.5646 1.5958 1.6262 1.6136 1.7416 1.7416 1.7416 1.8488 1.8488	0.6350 9102 1.1172 1.2062 1.2888 1.4378 1.4728 1.5064 1.50712 1.6032 1.6232 1.6232 1.6232 1.6232 1.6232 1.6332 1.6332 1.6332 1.6332 1.6332 1.6332 1.6332	0.14348 .7884 1.0236 1.1214 1.2112 1.4982 1.5126 1.14788 1.5126 1.5778 1.6092 1.61092 1.61092 1.61092 1.61092 1.61092	0.6400 9176 1.0278 1.2162 1.2162 1.2994 1.3774 1.4500 1.55522 1.4500 1.55522 1.4500 1.55522 1.4500 1.7664	0.4382 .7946 .9212 1.0318 1.2322 1.23646 1.3046 1.3046 1.4558 1.4910 1.5510 1.5538	0.6450 .7978 1.2352 1.1352 1.2686 1.300 1.3496 1.3682 1.4612 1.4612 1.4612	0.1416 6474 8028 1.0400 1.1398 1.2308 1.2378 1.3550 1.3550 1.3550 1.3550 1.3550 1.3550	0.1434 6498 8058 9322 9898 1.0438 1.0438 1.1440 1.2786 1.2786 1.3202 1.3990 1.4736 1.5436

TABLE II - Continued

VALUES OF  $2\left(\frac{\rho_1}{\rho_2}\right)^{1/2}\left(\frac{q_1}{q_0}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued  $\left[m_0 = 0.65\right]$ 

			·											
AH Qo	-0.50	-0-45	<del>-</del> 0•†0	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0+05	0	0.05	0.10	0.15
.700 .500 .400 .300 .250 .250 .075 .025 .025 .050 .025 .050 .025 .025 .02	1.7134 1.7962 1.8740 1.9472 2.0502 2.0502 2.1046 2.1146 2.1612 2.1612 2.2060 2.2206 2.2350 2.2634 2.2914 2.2914	1.6848 11.7648 11.9948	1.6468 1.7372 1.9750 1.	1.9544 1.9584 1.05298 1.05298 1.05298 1.1276 1.1264	1.5668 6682 1.5668 1.56	1.6764 1.7264 1.8786 1.8787 1.9787 1.	1.5896 58928 58928 58928 58928 59928 5	1.67480 1.74881 1.83814 1.9638 8832 2.0221 1.9638 2.0032 2.00970 2.0789 2.11520 2.1251	1.6086 1.80506 1.80506 1.89552 1.93528 1.93558 1.93558 2.05542 2.05542 2.1108 2.1990 2.19928 2.19928 2.26640 2.26640 2.26640 2.26640 2.26640 2.26640	1.9476 1.9476 1.96882 1.96882 1.96882 0.06660 1.96860	1.5160 1.6278 1.7308 1.7308 1.77208 1.8720 1.8261 1.8720 1.9160 1.9794 2.0002 2.0002 2.0002 2.0006 2.1548 2.1548 2.1266 2.2260 2.2260	1.46402 1.58902 1.7900 1.77900 1.8372 1.88534 1.927490 1.9927490 1.97014 1.97014 2.03226 2.05226 2.13680 2.27526 2.22426 2.23407	1.6469 1.	1.1782 78298 1.1782 7829 7829 7829 7829 7829 7829 7829 7
P <sub>1</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80	
0 025 050 075 100 250 250 300	.9264 1.1200 1.2820 1.4232 1.6684 1.6646 1.7184 1.7704 1.7754 1.8204 1.8146	1.6776 1.6776 1.7274 1.7800 1.8054 1.8854 1.8788 1.90584 1.90584 1.90584 2.01584 2.01584 2.01584 2.01584 2.01584	1.2354 1.1564648 1.1562648 1.562648 1.56264 1.77685 1.88648 1.77685 1.88648 1.	8226 1.0438 1.2223 1.34516 1.5744 1.5744 1.6628 1.7192 1.7729 1.7729 1.79246 1.9926 1.9927 1.	1.1428 1.3828 1.3828 1.45392 1.51322 1.6126 1.6716 1.6798 1.7278 1.7824 1.7829 1.90590 1.90590 1.90592 2.0728	1.3898 1.4602 1.5270 1.55904 1.6208 1.6800 1.7086 1.7642 1.8174 1.8174 1.8698 1.9670	1.3968 1.4674 1.5346 1.55862 1.5668 1.5686 1.6886 1.7776 1.8278 1.8268 1.9290 2.1188	1.1596 1.24276 1.46374 1.45944 1.45944 1.50470 1.5070 1.5070 1.6570 1.7636 1.7636 1.7636 1.7636 1.8882 1.9880 1.98	7002 70630 106524 1.25344 1.4162 1.4162 1.51916 1.	1.0742 1.1702 1.2504 1.3402 1.34902 1.4566 1.4536 1.5534 1.6832 1.6832 1.7992 1.7992	8500 1.0790 1.1756 1.2210 1.2642 1.3462 1.3462 1.4538 1.4956 1.5640 1.6294 1.7508 1.8628	1.1334 1.1808 1.2262 1.2700 1.3526 1.3916 1.5024 1.5024 1.6990 1.8164	.5276 .7266. .9876. .9876. .9876. .9876. .9876. .1.860. .1.275. .1.375	

TABLE II - Continued

values of 2  $\binom{\rho_1}{\rho_2}$   $\binom{q_1}{q_0}$  for determining point drag coefficient - **Continued** 

•							$M_0 = 0$	.70]						
P <sub>1</sub>		-0.45	-0.40	-0.35	-0.30	-0.25	]	-0.15	-0.10	-0.05		0.05	0.10	0.15
- 075 - 100 - 150 - 250 - 300 - 400 - 500	2.2068 2.2204 2.2470 2.2730 2.2984 2.3234 2.3716	2.1930 2.2676 2.2676 2.2938 2.3196 2.3196 2.1166	2.1904 2.2326 2.2604 2.2874 2.3149 2.3658 2.4163	1.5078 1.6058 1.7806 1.7806 1.9536 1.9536 1.9536 1.9536 2.0374 2.0374 2.0374 2.1080 2.1180 2.	2.1822 2.2124 2.2120 2.2708 2.2990 2.3542 2.4068 2.1578	1.4208 1.5308 1.7288 88.968 1.905366 1.905366 1.905366 1.905366 1.90536 1.9056	1.3724 1.4884 1.5902 1.7808 1.9809 1.9800 1.9800 2.0346 2.0696 2.1046 2.1538 2.2176 2.12784 2.2176 2	1.3188 1.4420 1.5548 1.5548 1.7488 1.87768 1.99570 2.0520 2.05502 2.05502 2.1270 2.127	1.2602 1.3694 1.6166 1.7166 1.	1. 1958 1. 3370 1. 4750 1. 5750 1. 6769 1. 8192 1. 8050 1. 9462 1. 9660 1. 9462 1. 9660 2. 0054 2. 0054 2. 0620 2. 0890 2. 1346 2. 2366 2. 2366 2. 2366 2. 2366 2. 2536 2. 5326	1.1244 1.2770 1.4106 1.5302 1.6390 1.7862 1.8758 1.9186 1.9802 2.0198 1.9802 2.0582 2.0790 2.1502 2.1140 2.1852 2.21946 2.2224 4.6602	1.0148 1.2146 1.4816 1.4816 1.5962 1.7508 1.7508 1.9528 1.9528 1.9528 1.9528 2.0536 2.0536 2.2694 2.2694 2.2694 2.25150	0.9550 0.9550 1.1388 1.2936 1.4290 1.6560 1.7624 1.7624 1.8792 1.9234 1.9234 1.9250 1.9870 2.00676 2.10676 2.10676 2.11438 2.1804 2.1804 2.1804 2.25060	0.8516 1.05786 1.05786 1.2708 1.5008 1.50724 1.50724 1.50724 1.8915 1.8915 1.8915 1.9580 1.95
AH P1	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	<b>0.7</b> 5	0.80	
.600 .500 .400 .250 .250 .1500 .0750 .025 .025 0750	1.1526 1.3096 1.4472 1.5762 1.6276 1.6822 1.7854 1.7854 1.8100 1.8342 1.8812 1.9048 1.9488 1.9488	1.3888 1.5196 1.58748 1.6928 1.7458 1.7714 1.7968 1.8458 1.8698 1.8932 1.9392	9772 1.16234 1.4634 1.52882 1.5882 1.7286 1.7286 1.7850 1.8556 1.8756 1.87934	1.475554 1.4757792654 1.475779266 1.47579568 1.47579568 11.776999	1.6086 1.6384 1.6672 1.6960 1.7238 1.7510 1.7780 1.8042 1.8300	1.0952 1.2704 1.4216 1.4904 1.5558 1.5587 1.6480 1.6772 1.7758 1.77684	1.0000 1.1930 1.27562 1.4296 1.4988 1.55646 1.56270 1.6574 1.6856 1.7440	1.1908 1.1946 1.2846 1.3636 1.4772 1.5475 1.5736 1.6364 1.6666 1.6964	1.0108 1.1132 1.20614 1.29710 1.4052 1.4450 1.54810 1.54820 1.6454	.9000 1.0160 1.1190 1.2184 1.2984 1.3390 1.3782 1.4162 1.4532 1.4532 1.5236 1.5236	1.1248 1.2188 1.2626 1.3052 1.3460 1.4238 1.4608 1.4968 1.5318	1.0264 1.1306 1.1786 1.2250 1.2259 1.3118 1.3530 1.3530 1.4312	.9136 1.0316 1.0852 1.1846 1.2308 1.2740 1.3186 1.3598	

TABLE II - Continued

VALUES OF  $2\left(\frac{p_1}{p_2}\right)^{1/2}\left(\frac{q_1}{q_0}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[\mathbf{w}_{0}=0.75\right]$ 

AH P1	-0.50	-0.45	-0.40	-0.35	-0.30	-0,25	-0.20	-0.15	-0.10	-0.05	o	0.05	0.10	0.15
.050 .025 0 .025 050 075 160 250 250 300 400	2.0682 2.0816 2.0816 2.1088 2.1218 2.1250 2.1608 2.1608 2.17608 2.2470 2.2470 2.3606 2.3606 2.3606	1.6;5;1 1.8070 1.8070 1.9;726 1.9;726 2.0334 2.0339 2.0320	1.7888 1.8608 1.9612 2.02[0.0 2.02[0.0 2.05[0.0 2.05[0.0 2.05[0.0 2.05[0.0 2.05]0.0 2.05[0.0 2.05[0.0 2.05[0.0 2.05[0.0]0.0 2.05[0.0 2.05[0.0]0.0 2.	1.7678 1.84,32 1.9140 1.94,78 1.9810 2.0128 3.0444 2.05,98 2.05,98 2.05,98 2.1046 2.1192 2.1388 2.1482 2.1482 2.2182 2.2182 2.2182 2.2182 2.2182 2.2182	1.000 1.000 1.000 1.000 2.000	1 - 5702 1 - 5702 1 - 5702 1 - 97502 1 - 97502 1 - 97502 1 - 97502 2 - 05502 2 - 05602 2 -	1.59700 1.69700 1.7762 1.89526 1.9526 1.9526 1.9588 2.0212 2.0538 2.0512 2.0548 2.1058 2.112 2.13560 2.13560 2.13560 2.13560 2.13560 2.13560 2.13560 2.13560	1.5606 1.6586 1.7492 1.8732 1.9722 1.9122 1.9122 1.9864 1.2004 1.9864 1.2004 1.2004 1.2004 1.2006 1.	2.268 2.422	1.7786 1.824 1.824 1.9056 1.9450 1.9450 1.9836 2.00210	1.54592 1.54592 1.54592 1.67473 1.8793 1.8793 1.8793 1.9412 1.9808 1.990	1.2750 1.5702 1.5702 1.6132 1.6132 1.8518 1.8518 1.9572 1.	1.1766 1.4512 1.56762 1.7664 1.7768 1.7748 1.7748 1.8214 1.8214 1.8214 1.8214 1.9520 1	1.0846 1.25726 1.5256 1.5256 1.5256 1.5256 1.5256 1.5256 1.5256 1.5256 1.5256 1.5256 1.5256 1.5256 1.5256 1.5266 1

AH Qo	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
400 500 600	1. 4722 1. 5918 1. 6478 1. 7512 1. 7522 1. 8014 1. 8252 1. 8948 1. 8720 1. 8720 1. 8948 1. 9392 1. 9392 2. 0638 2. 0638 2. 1116 2. 22550 2. 2118 2. 2550 2.	91098 1107474 127474 15432 11744 117	1.2036 1.2036 1.3574 1.45552 1.65148 1.6724 1.67254 1.67254 1.6726 1.8726 1.8726 1.8726 1.8726 1.8726 1.8726 1.8726 1.8726 1.9226 1.7728 1.9226 1.9226 1.9226 1.7728 1.9226 1.922	0.6538 9226 1.2242 1.2918 1.4368 1.5030 1.5660 1.6652 1.7760 1.7668 1.7760 1.7648 1.8648 1.8648 1.8848 1.8848 1.8848 1.8848 1.8848 1.8848 1.9858 2.0684 2.0684	1.0354 1.2192 1.3756 1.4430 1.5764 1.5766 1.6656 1.6656 1.6656 1.722 1.7	0.6618 93408 1.3886 1.3886 1.4558 1.5228 1.5228 1.5228 1.6176 1.6472 1.6472 1.7054 1.77054 1.7878 1.7878 1.7878 1.7878 1.9630 1.9630 1.9630 1.9630 1.9630 1.9630 1.9630 1.9630 1.9630 1.9630 1.9630 1.9630 1.9630 1.9630 1.9630	1	0.6696 1.1530 1.2422 1.4020 1.4740 1.5086 1.51422 1.5086 1.5070 1.6382 1.6382 1.6382 1.6388 1.78380 1.78380 1.8300 1.8300 1.9406 1.9406 1.9406 1.9406 1.9406 1.9406 1.9406	1.0510 1.1598 1.2498 1.3430 1.14104 1.14104 1.1518 1.5518 1.5518 1.56170 1.56170 1.5781 1.6170 1.7851 1.785	0.6778 .95672 1.0668 1.2572 1.34906 1.4560 1.4560 1.4560 1.4560 1.55946 1.5692 1.5692 1.6892 1.6892 1.8612 1.8612 1.8612 1.8612 1.8624	.9618 1.0734 1.1734 1.2646 1.3074 1.3488 1.3488 1.4274 1.5010 1.5766 1.5766 1.5766 1.5766 1.6994 1.7596 1.7596	.8360 .9674 1.0709 1.1802 1.2270 1.2270 1.3558 1.3558 1.3588 1.4752 1.4752 1.5990 1.5800 1.58462 1.7702 1.8840	1.2792 1.3226 1.3642 1.4438 1.5184 1.5892 1.6560 1.7198 1.8390

TABLE II - Continued

VALUES OF  $2\left(\frac{\rho_1}{\rho_2}\right)^{1/2}\left(\frac{q_1}{q_0}\right)^{1/2}$ 

FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $[\mathbf{M}_{\mathbf{o}} = \mathbf{0.80}]$ 

P <sub>1</sub>	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
0.800 .700 .600 .500 .400 .250 .250 .150 .050 .025 .050 .050 .050 .075 .100 .200 .200 .250 .200 .250	1.6558 1.7268 1.7268 1.9150 1.8558 1.9160 1.9706 1.9706 2.024 2.0370 2.0524 2.0704 2.0704 2.0704 2.0704 2.11700 2.11700 2.11700 2.11700 2.2147 2.2577 2.2577 2.2538	1.9394 1.9682 1.99682 2.0236 2.0236 2.0564 2.0564 2.0636 2.0764 2.0691 2.1022 2.1146 2.1276 2.1756 2.1756 2.22266 6.2356 6.2356 6.2356 6.2356 6.2356 6.2356 6.2356 6.2356	1.6174 1.6960 1.76900 1.8374 1.96300 1.93300 1.9630 2.0624 2.0624 2.0624 2.0624 2.1024 2.1024 2.1158 2.1158 2.2158	1.57542 1.89246 1.9246 1.9246 1.9560 1.9560 1.9560 2.0154 2.0596 2.0734 2.1010 2.1146 2.1146 2.125 2.1368 2.230d 2.235 2.2368 2.2368 2.2368 2.2368	1.5578 1.6578 1.6578 1.88098 1.9140 1.9466 2.0098 2.0594 2.0594 2.0594 2.1245 2.1245 2.1245 2.1245 2.2822 2.3298 2.3298 2.3298 2.3298 2.3298 2.3298 2.3298 2.3298 2.3298	1.6282 1.7132 1.7020 1.8658 1.9010 1.9554 1.20008 2.0166 2.0322 2.0176 2.055 2.120 2	1.5990 1.6890 1.7718 1.8490 1.8718 1.82162 1.92162 1.92162 2.0526	1.5676 1.6688 1.6688 1.8296 1.8296 1.9052 1.9052 2.0108 2.0108 2.0266 2.0108 2.0266 2.1688	1.5312 1.6314 1.728 1.8076 1.8180 1.8876 1.9612 1.9792 1.9793 2.0144 2.052 2.0652	1.4924 1.5946 1.6940 1.7832 1.8254 1.80506 1.9050 1.9050 1.9050 2.0520 2.0520 2.0520 2.1514 2.1873 2.1873 2.7384 2.7384	1.448 1.5602 1.6620 1.7556 1.7958 1.8838 1.9432 2.0000 2.0184 2.0050 2.0184 2.0050 2.1764 2.1764 2.2066 2.2176 2.2	1.4006 1.5194 1.6268 1.7254 1.7716 1.8164 1.8594 1.9612 1.9612 1.9612 2.0377 2.0556 2.0916 2.1266 2.1936 2.2577 2.318	1.47474 1.4748 1.6920 1.7476 1.7874 1.8748 1.8976 1.9186 1.9594 1.9594 1.9594 1.9594 2.0184 2	1.284 1.4550 1.6550 1.7058 1.8058 1.8058 1.8924 1.9956

P <sub>1</sub> Δ <u>H</u> q <sub>o</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
0.800 .700 .500 .400 .250 .150 .150 .075 .075 .075 .075 .075 .075 .075 .0	1.6148 1.6690 1.7704 1.8184 1.8816 1.8870 1.9900 1.9752 1.9752 2.1118 2.1499 2.220 2.2882 2.3532	.9660 1.1500 1.3088 1.4472 1.52706 1.6276 1.6284 1.7452 1.8094 1.8700 1.9470 1.	8522 1.02498 1.5299249 1.5299249 1.5299249 1.5299249 1.5299249 1.6968 1.7494 1.8956 1.9145 2.06992 2.25908 2.25908 2.25908 2.25908 2.25908	72542 11.7440 11.74698 11.74698 11.76962 11.76962 11.76962 11.76838 11.88766 11.8876	0.5596 86596 1.8636 1.4618 1.41806 1.4518 1.4806 1.6072 1.6072 1.8022 1.8728	7,758 9,882 1,3482 1,4210 1,4210 1,4212 1,5768 1,6194 1,6784 1,7626 1,76	0.5670 1.2782 1.3582 1.5682 1.5682 1.6690 1.6610 1.6610 1.6710 1.7756 1.8554 1.9060 1.9548 2.0020 2.1766 2.2568	1.0016 1.2006 1.2072 1.3674 1.5126 1.5126 1.5122 1.6112 1.6121 1.6728 1.7028 1.7028 1.7028 1.7628 1.8688 1.7028 1.9200 1.9690	0.5744 .8890 1.1134 1.2958 1.3758 1.4522 1.4880 1.5250 1.5250 1.6540 1.728 1.8286 1.7288 1.9338 2.9338	1.0146 1.1210 1.2164 1.3044 1.3858 1.4618 1.4618 1.4618 1.55676 1.63546 1.63546 1.7852 1.8414 1.9954 2.0928 2.1824 2.2022	0.5816 .9012 1.0212 1.1244 1.3128 1.35146 1.35146 1.5436 1.5436 1.5436 1.5734 1.5764 1.5764 1.9610 2.0604 2.1536	7646 90670 1 92670 1 12324 1 2324 1 23212 1 3638 1 1430 1 5524 1 1680 1 680 1 9214 2 0252 2 2134	9114 1.142 1.1922 1.2928 1.2858 1.3296 1.3720 1.4128 1.4528 1.4528 1.4528 1.56388 1.56388 1.67614 1.8786 1.8786 1.8786

#### TABLE II - Continued

VALUES OF 2  $\left(\frac{\rho_1}{\rho_2}\right)^{1/2} \left(\frac{q_1}{q_0}\right)^{1/2}$ 

FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[\mathbf{w_o} = 0.85\right]$ 

Р <sub>1</sub>	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
.700 .600 .500 .400 .250 .250 .150 .075 .025 .025 .050 .025 050 150 150 250 250 250 250 250	1.6316 6.6976886 6.697688746648 8.7466484 9.92684 9.92	1984644460000468000064876480000648704648000064967044648000064967044648000064967044648000064967068800006496800006496800000649680000649680000649680000649680000649680000649680000649680000064968000064968000006496800006496800006496800000649680000064968000000649680000000000	11111111111111111111111111111111111111	11111111111111111111111111111111111111	11111111111111111111111111111111111111	1.4382 1.52566 1.70562 1.70562 1.70562 1.70562 1.705012 1.85860 1.95012 1.9501	11111111111111111111111111111111111111	6468648668868868868868868868868868868868	111788464 111788866 1117888866 11178888866 11178888866 1117888886 1117888 111788 11788	11111111111111111111111111111111111111	11111111111111111111111111111111111111	1.2970 1.1573 1.	1.2758 2758 2758 2758 2758 2758 2758 2759 2759 2759 2759 2759 2759 2759 2759	111166 17166 1716767514 172767514 172767514 1727677514 1727677516 17276774

<u>АН</u>	0 <b>.20</b>	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
0.800 .700 .400 .300 .2500 .150 .075 .050 .025 .0575 .150 .250 .250 .250 .250 .250 .250 .250 .2	1 .0666 1 .2604 1 .4016 1 .5288 1 .6914 1 .7902 1 .8366 1 .89590 1 .98590 1 .9862 1 .9456 2 .02662 2 .0214 2 .2270 2 .23556	1.0114 1.1924 1.3452 1.45984 1.5984 1.7578 1.8538 1.8766 1.8766 1.8940 2.1042 2.0660 2.1042 2.24478 2.24478	1.1156 1.2826 1.45538 1.5538 1.6686 1.7238 1.7986 1.8470 1.84706	11111111111111111111111111111111111111	0.748487468 0.74842744487468 11.3457844874588 11.3457866777755680 11.345787488857049698468 11.1111111111111111111111111111111111	1.04626 239062966 1.239062966 239062966966 2590629669696 2590674066 2674206 26742066 26742066 2674206 2674206 2674206 2674206 267	0.11.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.	8260428646666866666666666666666666666666	0.9160450 1.294264688 1.29426588 1.29426588 1.29426588 1.2942658 1.294268 1.29	8318 1.0754 1.2754 1.2548 1.4338 1.4712 1.5776 1.5776 1.57666 1.6136 1.67666 1.67666 1.8792 2.1262 2.21262 2.21470	0.6799 -968984490 -968984490 -968984490 -968984490 -968984490 -968984490 -968984490 -96898499 -96898499 -96898499 -96898499 -96898 -96898 -968	.8432 .9758 1.0914 1.1944 1.37140 1.45296 1.45296 1.6066 1.6066 1.6066 1.6330 1.7932 1.9634 2.1598	0.6878 .8488 .9824 1.0982 1.2022 1.2966 1.3408 1.3434 1.4614 1.5032 1.5404 1.6120 1.6798 1.7444 1.8058 1.7244 2.0282 2.1278

TABLE II - Continued

VALUES OF  $2\left(\frac{\rho_1}{\rho_2}\right)^{1/2}\left(\frac{q_1}{q_0}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[\mathbf{M}_{0} = 0.90\right]$ 

						[wc	, = 0.90	נ'				_		
P <sub>1</sub>	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20		-0.10	-0.05	0	0.05	0.10	0.15
0.808 .700 .600 .500 .400 .250 .075 .075	1.5354 1.6022 1.6644 1.7778 1.8294 1.8784 1.920 1.9362 1.9474 1.9584 1.9584 1.9584 2.012 2.0116 2.0522 2.0720 2.0720 2.1284 2.12	1.5254 1.5966 1.6624 1.7622 1.7820 1.8862 1.9128 1.9128 1.9128 1.9722 1.9834 1.9950 2.0062 2.0172 2.0282 2.0172 2.01282 2.01704 2.1504 2.1504 2.1504 2.2590 2.2590	1.5120 1.5876 1.6576 1.78234 1.8168 1.8222 1.9450 1.95714 1.95714 1.99	1.4946 1.5748 1.5748 1.7178 1.7820 1.8714 1.9268 1.95666 1.99666 1.99666 1.99666 1.9962 2.00944 2.00944 2.00946 2.0536 2.11248 2.2660 2.3426	1.4732 1.5586 1.7700 1.7780 1.7780 1.9578 1.9758 1.9758 1.9781 1.9984 2.0214 2.0524 2.0524 2.1576 2.224 2.2466 2.3664 2.3664	1.4474 1.52186 1.52188 1.6990 1.7704 1.8592 1.99304 1.9738 2.0022 2.0160 2.022 2.0160 2.0562 2.0562 2.144 2.154 2.260 2.3028 2.344 2.3838	1.4176 1.5146 1.5030 1.6804 1.7600 1.8306 1.8306 1.89282 1.9584 1.9738 2.0032 2.0176 2.0202 2.0460 2.0602 2.0736 2.1526 2.1526 2.2726 2.3772 2.3760 2.4010	1.48264 1.58064 1.66644 1.8260 1.82560 1.92526 1.92526 2.0318 2.0318 2.0318 2.0318 2.0318 2.0318 2.0318 2.0318 2.0318 2.0318 2.0318 2.1518 2.1518 2.1518 2.1518 2.1518 2.1518 2.1518 2.1518 2.1518 2.1518 2.1518 2.1518 2.1518	1.428 1.4542 1.4542 1.6458 1.8052 1.8052 1.89156 1.99156 1.9982 1	1.2972 1.1,170 1.6276 1.7098 1.7098 1.7920 1.83688 1.9580 1.9580 1.9580 1.9580 1.9580 2.00144 2.0028 2.0028 2.1528 2.1528 2.1528 2.1528 2.1428 2.1428 2.1428 2.1428	1.2454 1.3746 1.3746 1.4892 1.6864 1.7732 1.8142 1.85322 1.9452 1.9452 1.9628 2.0340 2.05668 2.05668 2.1506 2.1506 2.1506 2.1506 2.1506 2.1506 2.1506	1.3268 1.4496 1.5594 1.6594 1.7512 1.7942 1.79528 1.9762 1.9528 1.9710 2.05906 2.05906 2.1250 2.1250 2.1250 2.3472 2.3406 2.3473 2.34666	1.198 1.2726 1.4050 1.5228 1.6288 1.6288 1.7256 1.7712 1.8150 1.8572 1.8980 1.9376 1.9376 1.9376 2.0128 2.0428 2.0488 2.0838 2.1510 2.1510 2.1510 2.21488 2.3042 2.41468	1.2118 1.3552 1.4810 1.59408 1.7448 1.79054 1.87892 1.8998
AH qo		0.25	0.30	0.35	0.40	0.45	0,.50	0.55	0.60	0.65	0.70	0.75	0.80	
0.800 .700 .600	0 1.1421 0 1.2990 0 1.4340	6 0.850b 1.0638 1.0638 1.2388 1.3288 1.5118 1.7818 1.7818 1.7818 1.7828 1.8757 281.8972 281.8972 281.8972 281.9910 1.9928 281.9928	1.1646 1.324	.8656 1.0832 1.2592	0.298 .7366 .7368 1.1849 1.1849 1.1849 1.1849 1.1856 1.6166 1.6756 1.7567 1.8035 1	0.5676 8806 1.1022	0.3038 1.006 1	2 1.120	1.054714 1.314547		4 0.31211 1.0.31211 1.0.3121 1.0.	1	0.3168 7828 8.9296 1.1644 2.1.3110 6.1.3110 6.1.31310 6.1.3	

TABLE II - Continued

VALUES OF 2  $\left(\frac{\rho_1}{\rho_2}\right)^{1/2} \left(\frac{q_1}{q_0}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[\mathbf{M}_{\mathbf{0}} = 0.95\right]$ 

AH Qo	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	-0.05	0	0.05	0.10	0.15
700 600 500 100 150 150 150 100 150 100 10	1.6818 1.7528 1.7608 1.8038 1.8038 1.8049 1.8694 1.88900 1.8900 1.9900 1	11.726 11.726 12.725 13.725	11111111111111111111111111111111111111	100 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	1.7588 1.8176 1.84764 1.92928 1.945478 1.9679	11111111111111111111111111111111111111	1920644424418888600476061864468 193068794418988600576186959888 19307689718476000114195888 19307689718476000114195888	11111111111111111111111111111111111111	11111111111111111111111111111111111111	11111111111111111111111111111111111111	11111111111111111111111111111111111111	11111111111111111111111111111111111111	11111111111111122222222222222222222222	1.15.06660020000000000000000000000000000000

P <sub>1</sub>	0.20	0.25	0.30	0.35	0.40	0.45	0.50	0.55	0.60	0.65	0.70	0.75	0.80
0.800 .700 .600 .500 .250 .250 .150 .075 .075 .025 .025 .025 .025 .100 .200 .200 .200 .400 .700 .700	1.1892 1.7386 1.5850 1.6858 1.7360 1.8762 1.8762 1.9376 1.9577 1.9577 1.9577 2.0352 2.0152 2.01630 2.1170 2.3630 2.1636	1.164742 1.24456472 1.154574 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545744 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.154574 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.154574 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545744 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545742 1.1545744 1.1545744 1.154574	1.5012 1.6256 1.7288 1.7788 1.8746 1.8746 1.8946 1.9462 1.	9.3575.18875.18975.18975.18975.18975.18975.18975.18975.18975.19975	81924 1.02395842889595484 0.23958428895842745688917242745688917242749004904082 0.0490408121111111111111111111111111111111111	916190 91	1.0726 20726	0.684 -97032 1.37088 1.5774 1.45088 1.5774 1.7628 1	1.0822 1.28702 1.44824 1.5214 1.5214 1.6250 1.6250 1.7504 1.7504 1.83504 1.83504 1.9928 2.13506 2.2138 2.37016	97406 9804628 9804628 11.2386828 11.566378444 11.56637844 11.77950786 11.77950786 11.77950786 11.9918 11.9918 12.22222	8626 1.1108 1.21480 1.39404 1.59562 1.549562 1.549562 1.549562 1.7504 1.82562 1.7798 1.8192626 2.07988 2.22626	0.703488 0.11984 11.2384888 11.448626 11.5563964 14.566166 11.56616 11.56616	1.0112 1.1286 1.237292 1.4172 1.4586 1.5580 1.5580 1.5752 1.6476 1.7158 1.7806 1.7158 1.7806 1.7158 1.7806 1.7158 1.7806 1.7158 1.7806 1.7158

TABLE II - Concluded

values of 2  $\left(\frac{\rho_1}{\rho_2}\right)^{1/2} \left(\frac{q_1}{q_0}\right)^{1/2}$ 

FOR DETERMINING POINT DRAG COEFFICIENT - Concluded

M<sub>o</sub> = 1.00

								_						
P <sub>1</sub> Δ <u>H</u> q <sub>o</sub>	-0.50	-0.45	-0.40	-0.35	-0.30	-0.25	-0.20	-0.15	-0.10	<b>-</b> 0. <b>0</b> 5	0	0.05	0.10	0.15
.700 .600 .1400 .250 .250 .250 .025 .050 .025 .025 .0	1.5862 1.5862 1.6836 1.7498 1.7698 1.7698 1.8096 1.8096 1.8381 1.8381 1.8381 1.8381 1.8744 1.8748 1.935186	1.4802 1.5418 1.5992 1.6524 1.7024 1.7024 1.7724 1.8368 1.7724 1.8372 1.8572 1.8572 1.8572 1.8572 1.8572 1.8572 1.8572 1.9662 1.9702 1.9876 1.9342 1.9702 1.9876 1.95342 1.9702 1.9876 1.9534 1.9702 1.9876 1.9534 1.9702 1.9876 1.9534 1.9702 1.9876 1	1.5454 1.6046 1.7180 1.7684 1.7521 1.8156 1.8752 1.8502 1.8928 1.8928 1.9342 1.9342 1.9342 1.9342 1.9342 1.9342 2.0026	1.6728 1.6728 1.7298 1.7850 1.8350 1.8350 1.8576 1.8521 1.9058 1.9150 1.9592 1.9450 1.9586 1.9582 1.9582 1.9582 2.0102 2.0102 2.0102 2.0102	1.6128 1.6728 1.6738 1.73946 1.8438 1.8438 1.8438 1.8438 1.9348 1.9348 1.9348 1.9348 1.93568 2.03568 2.03568 2.01568	1.60784 1.7426 1.8588 1.8588 1.8588 1.9522 1.9576 1.9628 1.9688 1.9888 1	1.7434 1.8670 1.8668 1.8954 1.9366 1.9366 1.9760 1.9760 2.0264 2.0744 2.0744 2.0794 2.1268	1.5698 1.66862 1.66862 1.849602 1.88716 1.88716 1.8973 1.97866 0.03595 0.03596	1.65748 1.65748 1.85788 1.85784 1.85798 1.95500 1.95500 1.95500 1.95940 2.02620 2.02620 2.10306 2.10306 2.10306 2.10306	1.6432 1.7248 1.8360 1.83604 1.9542 1.9582 1.9683 2.0282 2.0282 2.0282 2.0574 2.1130 2.1130 2.1130 2.2160	1.6244 1.7104 1.7904 1.8288 1.8652 1.9906 1.9574 1.9549 2.0000 2.0152 2.0466 2.0616 2.0616 2.0912 2.1754 2.2384	1.6012; 1.6934 1.7180 1.8168 1.8938 1.9948 1.9948 1.9948 1.9918 1.9914 2.0174 2.03174 2.03174 2.03174 2.1246 2.2378	1.5736 1.5718 1.7618 1.8038 1.8046 1.8946 1.9940 2.0286 2.0154 2.0286 2.0454 2.0546 2.1540 2.1540 2.1540 2.1540 2.1540 2.1540	1.2944 1.5412 1.5412 1.6456 1.7458 1.8288 1.8704 1.9480 1.9480 1.9480 2.0428 2.0404 2.0518 2.0518 2.1588 2.2468
P <sub>1</sub>	0.20	0.25	0.30	0.35	0-40	0.45	0.50	0,55	0,60	0.65	0,70	0.75	0.80	
0.800 .700 .600 .500 .250 .250 .150 .075 .056 .025 .025 .025 .025 .025 .025 .036 .036 .036 .036 .036 .036 .036 .036	1.236 1.379 1.616 1.76 1.617 1.899 1.899 1.899 1.995 1	8 1.1706 0 1.3262 0 1.4610 4 1.5806 6 1.6886 2 1.7388	1.9452 1.2652 1.5659 1.7660 1.7660 1.7660 1.8578 1.8578 1.1.9686 1.9245 1.9245 1.9686	0 1.675 0 1.729 0 1.729 0 8 1.830 0 1.878 0 1.878 0 1.921 0 1.921 0 1.921 0 2.23 0 2	900 1.294 1.1576 1.	0 1.226 0 1.222 0 1.386 0 1.528 0 1.528 0 1.528 0 1.528 1.738 1.748 1.877 0 1.799 0 1.999 0 1.968 0 1.	10 0.608 1 1.14 1.14 1.14 1.14 1.14 1.14 1.14 1.	4 1.48 1.566 6 1.68 8 1.71 1.80	194 1.34 1.	12   1.44 1.51 1.52 1.62 1.62 1.62 1.62 1.62 1.62 1.62 1.62 1.62 1.62 1.62 1.62 1.63 1	22 - 95,202	1. 08 1. 19 1. 19 1. 1. 28 1. 1. 28 1. 1. 46 1. 46 1	46   1.099 34   1.200 34   1.399 36   1.399 36   1.42 37   1.42 37   1.599 38   1.599 38   1.670 38   1.6	90 450 450 450 450 450 450 450 450 450 45

## TABLE III

values of 
$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}$$
 for determining point drag coefficient

## FOR FLOWS WHEREIN ENERGY IS ADDED

 $[\mathbf{u}_{\mathbf{o}} = \mathbf{o}]$ 

						o_ oj						•
AH R	0	0.01	0.02	0 <b>.0</b> 3	0.05	0.10	0.15	0,20	0.25	0.30	0.35	о•40
0.800 .700 .600 .500 .250 .250 .200 .150 .025 .025 075 100 150 250 300 400 500 800	0.1528 -15275	0.5478 -4473 -36279 -28204 -1583 -1290 -1003 -00570 -01977 -04188 -0704 -1882 -1882 -2698 -3466	0.5430 .4425 .5577 .2831 .2156 .1535 .1242 .0682 .0415 .0284 .0155 .0028	0.5381 .43728 .25782 .2107 .1486 .1193 .0933 .0335 .0106 .0021 .0394 .0515 .0870 .1101 .1549 .19794 .23985 .3563	0.52872 34884 260132 10999 0011512 0011512 0011513 007099 0070994 1164738 1282757 1164738 1282757 1282757	0.5063 4058 32164 1168 21788 32164 1168 05315 0083 -003168 -003168 -007133 -0071333 -1184	0.48840 43840 21575851 001692495924551 001692495924551 001692495924551 0016924551 0016924551 0016924551 0016924551 0016924551 0016924551 0016924551 0016924551 0016924551 0016924551 0016924551 0016924551 0016924551 0016924551 0016924551 0016924551 0016924551 001692451 00169251	0.4658 26568 20588 20588 20588 207695 2018918	0 4467 26193 181977 1052 1052 1053 1054 1054 1055 1055 1055 1055 1055 1055	0 3295 3295 3295 3295 3295 3295 3295 3295	0.41291 34291 158550 02554 108659 108659 108659 108659 108659	0.3980 .2975 .2127 .1381 .0706 .0085 .0208 .0492 .0735 .1166 .1295 .1166 .1295 .1166 .1295 .11916 .22702 .25728 .25728 .23780 .37795 .41964

TABLE III - Continued

VALUES OF 
$$\left(\frac{\rho_2}{\rho_o}\right)^{1/2}$$
 -  $\left(\frac{q_2}{q_o}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[M_o = 0.05\right]$ 

					•		7					
$\frac{d^{O}}{\nabla H}$ , k	0	0.01	0.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
0.800 .700 .500 .400 .3250 .200 .150 .075 .0525 .0525 .0525 .0550 .1500	0	0.5466 36870 22889 10026330750 100307570 1005770 101370 101370 101	0.5417 35774 215746 2153466 0.04285 0.04285 0.0092244666 0.00922446666 0.00922446666 0.00922446666 0.00922446666 0.00922446666 0.00922446666 0.00922446666 0.00922446666 0.00922446666 0.009224466666 0.009224466666 0.009224466666 0.009224466666 0.009224466666 0.009224466666 0.0092244666666 0.0092244666666666666666666666666666666666	0.5372 .4369 .357753 .21892 .09365 .010217 .014713 .00535 .005359 .01255 .01355	0.5278 .4275 .34291 .2689 .13998 .0814 .052721 .00112 -01155 -0487 -06029 -072657 -06029 -072657 -06029 -072659 -07269 -0	0.5051 0.50575 0.50575 0.50575 0.50575 0.50575 0.50575 0.50571 0.50	0.4841 292475774 0.43899775774 0.684997757774 0.6849979914 0.684999999999999999999999999999999999999	0.1464981 1.64981 1.64981 1.64981 1.64995898 1.6499589 1.649991 1.649999 1.649999 1.6499999 1.64999999999999999999999999999999999999	0.4464 .3460 .2614 .1867 .1976 .0283 .0001 .0277 .08029 .1079 .1578 .1578 .2884 .3699 .44664 .34664	0.1290 3287 2141 1693 1021 0109 -01751 -0716 -08476 -11229 -13253 -14757 -17951 -2407 -3057 -3473 -14638	0.4126 -31237 .15237 .08579 .00538 .00514 .0858 .06140 .10167 .1267 .15179 .1761 .1881 .2115 .25771 .25791 .25791 .3636 .4032 .4032 .4032	0.3971 .2968 .2122 .1374 .0702 .0084 0209 0193 0769 1166 1295 11548 1672 1548 1672 1916 2036 2701 2726 2946 3794 4192 41957

# TABLE III - Continued

VALUES OF  $\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $[M_0 = 0.10]$ 

ΔH q <sub>o</sub>	0	0.01	0.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
0.800 .700 .600 .500 .250 .250 .150 .075 .055 .055 .055 .055 .055 .055 .0	0.54955 -2949551 -2949551 -2949551 -2949551 -2949551 -2949551 -2949551 -2949551 -2949551 -294951 -294951 -294951 -294	0 54446061 5446061	0.5357 535752 2.152350 0.1023557 0.0023577 0.0023577 0.0002577 0.0002577 0.0002577 0.0002577 0.0002577 0.0002577 0.0002577 0.	0.544 1.45	0.5253 -42542 -26000 -1390 -0833 -1090 -0833 -00115 -014636 -0759 -14635 -16357 -16359 -12869 -24869 -336	0.531,466 0.531,475,68 0.531,465,539 0.531,465,539,470 0.531,465,68 0.023,568,135,68 0.023,568,135,68 0.031,165,68,69,75,59,75,59,75,75,79,75,75,75,75,75,75,75,75,75,75,75,75,75,	0.4822 977777777777777777777777777777777777	0.4626 -2626	0.1442 2.6657 2.65677 1.05275 0.002747 0.002747 0.002747 0.002747 0.002747 0.002747 0.002747 0.002747 0.002747 0.002747 0.002747 0.002744 0.00274 0.00274 0.00274 0.00274 0.00274 0.00274 0.00274 0.00	0.4268 32687 16813 001781 0017	0.41052 .15052 .15052 .15049 .000445 .00080	0 3948 29508 13666 0078 00215 - 0497 - 1036 - 11246 - 1669 - 17911 - 12065 - 24938 - 27738 - 2775 - 44536

VALUES OF 
$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}$$
 FOR DETERMINING POINT DRAG COEFFICIENT - Continued

$$\left[\mathbf{M_o} = 0.15\right]$$

K AH qo	o	0.01	0.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
.700 .600 .500 .400 .250 .250 .150 .055 .025 .055 .025 .0575 .025 .0575 .050 .050 .050 .050 .050 .050 .0	0.5450 .4460 .3624 .2888 .2222 .1610 .1319 .1040 .0768 .0566 .0249 .0124 .0243 .0485	0.1 -1411 -38176791 -12877679196 -12877579950 -1287757950 -1287750	0.53260 536260 5	0.5179.246.743.20.00.00.00.00.00.00.00.00.00.00.00.00.	0.5211 -12285 -12385 -1	0.49962599777953749524456766779530481424416676677	0.478957088897667388257666741131130578875921593466996924477913513130578875360	0.459521 0.459521 0.459521 0.459521 0.459521 0.4579621 0.003461 0.0034691 1.00346866 1.0034686 1.0034686 1.0034686 1.0034686 1.0034686 1.003	0 1737668137566184773766813756602577913769137665366 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0.4237014865 -223701469865 -223701469865 -223701469865 -223701469865 -223701469865 -223701469865 -22370146980 -22370146980 -2237014680 -2237014680 -2237014680 -2	0.4448 0.473015326 0.0203268 0	0.3920 -3920 -3920 -3480 -00250747 -10366 -00250747 -11499 -11499 -11499 -11902553 -2293754 -2293754 -14589

#### TABLE III - Continued

values of 
$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}$$
 for determining point drag coefficient - Continued

 $\left[\mathbf{M}_{0}=0.20\right]$ 

$\begin{array}{c ccccccccccccccccccccccccccccccccccc$													
.700		0	0.01	0.02	0.03	0.05	0.10	0.15	0,20	0.25	0.30	0.35	0.40
250114612461247125618661867201221902563256227462901400177718261874192220162240244826432827300031633318500217822272276232324172640284930443229340235653719600256426132662271028033027323634313615378839524106700294129903039308731813404361338083992416643294484800330333543402345035443768397741724556452946934848	.700 .600 .500 .400 .300 .250 .250 .150 .075 .050 .025 0 025 050 075 100 150 250 250 300 400 500 600 700	.4412 .3585 .2856 .2197 .1592 .1305 .1028 .0760 .0499 .0372 .0246 .0122 0 -0121 -0240 -0358 -0475 -0703 -0927 -1146 -1361 -1777 -2178 -2564 -2941	.4364 .3537 .2807 .2148 .1543 .1256 .0979 .0711 .0450 .0323 .0197 .0049 .0170 .0289 .0407 .0524 .0752 .0977 .1196 .1410 .1826 .2227 .2613 .2990	.4315 .3488 .2759 .2100 .1494 .1207 .0930 .0662 .0401 .0274 .0149 .0025 .0029 .0219 .0338 .0456 .0573 .0801 .1025 .1244 .1458 .1874 .2276 .2662 .3039	.4268 .3441 .2711 .2052 .1447 .1160 .0883 .0614 .0353 .0226 .0100 -0024 -0147 -0267 -0386 -0504 -0621 -0848 -1073 -1292 -1506 -1922 -2323 -2710 -3087	.4174 .3347 .2618 .1959 .1353 .1066 .0789 .0521 .0260 .0133 .0007 -0117 -0239 -0397 -0714 -0997 -0714 -1166 -1386 -1600 -2016 .2417 -2803 -3181	.3952 .3125 .2395 .1736 .1131 .0843 .0566 .0299 .0037 .0090 .0216 .0339 .0462 .0583 .0702 .0820 .0937 .1165 .1389 .1608 .1823 .2240 .26440 .3027 .3404	.3744 .2917 .2188 .1529 .0923 .0635 .0359 .0090 .0171 .0298 .0424 .0548 .0670 .0791 .1028 .1145 .1374 .1374 .1598 .1817 .2031 .2448 .2849 .3236 .3613	.3550 .2722 .1993 .1334 .0728 .0441 .0163 .0105 .0366 .0493 .0619 .0745 .0865 .1105 .1224 .1341 .1569 .1224 .1341 .1569 .2012 .2227 .2643 .3044 .3044 .3808	.3367 .2539 .1810 .0545 .0258 -0020 -0288 -0549 -0676 -0802 -0927 -1048 -1170 -1289 -1407 -1524 -1752 -2196 -2410 -2827 -3615 -3992	.3194 .2367 .1638 .0978 .0085 -0192 -0461 -0722 -0849 -0975 -1099 -1222 -1342 -1462 -1580 -1697 -22583 -3000 -3402 -3402 -3406	.3032 .2204 .1475 .0815 .0209 .0078 .0355 .0623 .0885 .1012 .1138 .1262 .1384 .1506 .1625 .1743 .1860 .2313 .2532 .2746 .3163 .3565 .3565 .3952 .4329	.2878 .2051 .1321 .0661 .0055 .0232 .0509 .0778 .1039 .1166 .1293 .1417 .1539 .1660 .1779 .1898 .2014 .2243 .2467 .2687 .2901 .3318 .3719 .4106 .4484

VALUES OF  $\left(\frac{p_2}{p_o}\right)^{1/2} - \left(\frac{q_2}{q_o}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[\mathbf{M}_{\mathbf{0}} = 0.25\right]$ 

K ΔH q <sub>o</sub>	0 -	0.01	0.02	0.03	0.05	0.10	0.15	0,20	0.25	0.30	0.35	0.40
0.800 .700 .600 .500 .400 .250 .250 .250 .025 .025 .025 .025 .0	0.5314 -4352 -3537 -28166 -1567 -1286 -1012 -0748 -0366 -0243 -0120 -0120 -0120 -0120 -0120 -0120 -0120 -0120 -120 -	0.5266 43488 27119 12373 06943 03194 001685 001685 001685 001685 001687 001687 001687 001687 001687 001687 001687 001687 001687 001687 001687 001687 001687 001687	0.5217 -42540 -2769 -2769 -2769 -2769 -2769 -2769 -2765 -276	0.5170 -4208 -32672 -2672 -2021 -1141 -0867 -0347 -0098 -0025 -0145 -0389 -0837 -1483 -12866 -3033 -2666 -33389	0.5175 0.5175	0 148978879972992466799166799122698496 0 00000000000000000000000000000000000	0.4650 36882 215002 063451 109620 003451 -005467 -007804 -007804 -115816 -1158	0.4457 3494 26757 19506 0707 0426 01513 -0369 -0495 -0741 -0862 -0988 -1229 -1555 -1775 -122611 -3385 -35753 -419	0.43196 -112193 -0021937 -0021937 -0021937 -0021937 -0021937 -0021937 -0021937 -002193 -00219 -00219 -00219 -002193 -002193 -002193 -00219 -00219 -00219 -00219 -00219 -00219	0.41041 23254 232042 232042 232042 232042 232042 232042 232042 232042 232042 23204 20204 2	0.3941 29763 2163 10789 20191 -00366 -00387 -1036 -103	0.789 2789 2088 2088 2088 2088 2088 2088 2088 20

# TABLE III - Continued

VALUES OF 
$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}$$
 FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[\mathbf{M}_{\mathbf{O}} = 0.30\right]$ 

$\frac{\Delta H}{q_0}$ K	0	0.01	0.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
0.800 .700 .600 .300 .250 .250 .150 .150 .025 .050 .025 .050 .025 .050 .150 .150 .250 .250 .250 .250 .250 .250 .250 .2	0.524 -528	0.51239 12329 12329 1208	0.5128 .41813.73.44.668 .0326.4122.74.668 .0326.4122.74.669.000.009.14.009.14.009.14.009.14.009.14.009.14.009.14.009.14.009.16.0	0.516 .036	0.4995 432599478 13075996 13075996 13075996 13075996 100123596 100123596 100123596 100123596 100123596 100123596 100123596 100123596 100123596 100123596 100123596 100123596 100123596 100123596 10012359	0	0.45620 	0.435 72245 72245 72245 7226 7226 7226 7226 7226 7226 7226 722	0 1344 179902443 179902443 1000030580 1000030580 10000000000000000000000000000000000	0.436 0.40772 0.40772 0.40772 0.52172 0.521728 0.00765 0.00	0.3860 29111 14600 0010768 107708 1077076 1013574 114579 1	0.3707 .2762 .1958 .1249 .0607 .0017 -005791 -1168 -1168 -1169 -1169 -1160 -1896 -18

VALUES OF  $\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}$ 

FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[ M_{O} = 0.35 \right]$ 

	_											
$\frac{do}{d}$	0	0.01	0.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
0.800 .700 .600 .500 .400 .300 .250 .150 .100 .025 .025 050 025 075 100 250 250 300 400 500 600 800	0.5118 .4199 .270859 .15036 .0717512 .0135316 0.02338 .0112288 .02338 .0468699 .127590 .127590	0.5070 .4161 .26657 .14885 .06420 .01885 .0044 .001649 .001645 .001649 .0016475 .001649 .0016475 .001649 .001649 .001649 .001645 .001649 .0016	0.5028 .3618 .19318 .1147 .0627 .00325 .00325 .00910 .00910 .00917 .0091	0.4975 .49752 .49677 .257436 .13694 .05730 .00827 .00827 .009477 .02579 .00827 .009497 .009497 .009497 .01224 .121886 .29232 .29235	0.4861 .39180 .186178	0.467 .2926235530 .0052618 .0052618 .0052618 .0052618 .00526890 .00526	0 	0.42482 -18466668 -25466668 -031237990 -04237 -0423	0.4098 .31886 .1686.10420594 .00031610 .00031610 .000891379 .112690168 .112690168 .112708438 .122708438 .122708438 .122708438 .122708438 .122708438 .122708438 .122708438	0.3999 .29912 .158868 .0303284 .000873579 .000873579 .10090859789 .115457068 .11867768 .11867768 .11867768 .11867768 .11867768 .11867768 .11867768 .11867768 .11867768 .11867768 .11867768	0.22455557460 0.22455557460 0.0000000000000000000000000000000000	0.3612 .2687 .1900 .10573 .05079 .002498 -11165 -1124019 -1124019 -118588 -126805 -12686 -126

## TABLE III - Continued

values of 
$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2}$$
 -  $\left(\frac{q_2}{q_0}\right)^{1/2}$  for determining point drag coefficient - continued

 $[M_0 = 0.40]$ 

ΔH K	0	0.01	0.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
9 0.800 .700 .600 .500 .400 .250 .250 .050 .025 .050 .055 .050 .055 .050 .055 .050 .055 .050 .055 .050 .055 .050 .055 .050 .055 .050 .055 .050 .055 .050 .055 .0	0.4998 .4100 .3331 .2652 .2038 .1472 .1206 .0950 .0701 .0460 .0343 .0227 .0112 -0330 -0436 -0851 -1051 -1625 -1989 -2338 -2674 -3000	0.4951 .4052 .3283 .2604 .1990 .1425 .1159 .0693 .0412 .0269 .00648 .00648 .0069 .0378 .0484 .0699 .1295 .1673 .2038 .2387 .2723 .2723	0.4904 4905 32577 19477 10856 0365 0348 01327 00907 - 03125 - 07946 - 1147 - 1342 - 1728 - 24351 - 2797	0.4858 .79599 .1895190 .1895191 .1895191 .0805918 .09518	0.4767 .3868 .30999 .18099 .18099 .18099 .09717 .09717 .0110 .01214 .034553 .05669 .08885 .10885 .10881 .22574 .22574 .22327	0.45683399266889944941 	0.45458 35581005995674428 35458159946568763338 00191245687699908 00191245687699908 00191245687699908 00191246909 001919226909 001919226909 001919226909 001919226909 001919226909 001919226909 001919226909 001919226909 001919226909 001919226909 00191922699909 001919226909 001919269 001919269 0019192690 001919269 001919269 001919269 001919269 001919269 001919269 0	0.4160 324910 119282 119282 03605 0014803 0014	0.38332 -39832 -101499 -0018753688 -0018753688 -0018753688 -001875367 -1125678 -112678	0.3815 - 22144 - 21462 - 0024936 - 0024936 - 00397858 - 1039785 - 11919 - 11845 - 1184	0.046594 0.046594 0.046594 0.046590 0.046590 0.046590 0.046590 0.046590 0.04690 0.04690 0.04	0.3506 .2605 .1833 .1552 .0536 .0299 .0556 .0806 .1166 .1282 .1398 .1511 .1623 .1511 .1623 .1511 .1645 .2764 .2764 .2764 .2764 .2764 .3873 .3873 .4208

VALUES OF 
$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}$$
 FOR DETERMINING POINT DRAG COEFFICIENT - Continued

$$\left[\mathbf{M}_{\mathbf{0}} = 0.45\right]$$

<u>ΔH</u> <u>Q</u> <sub>0</sub>	0.4868	0.01	0.02	0.03	0.05	0.10 . 0.4426	0.15	0.20	0.25	0.30	0.35	0.40
0.800 .700 .600 .500 .250 .250 .150 .075 .055 .055 -055 -055 -150 -250 -300 -500 -500 -800	0.4666 .3993 .3215 .2582 .1983 .11724 .0682 .0448 .03321 .0109 00320 0423 0626 1020 1210 1256 19365 25903		.4775 .39951 .24889 .1889 .10830 .05854 .005553 .00215 .00215 .00515 .00	.4729 .38129 .38143 .1843 .10784 .05407 .005407 .005461 .00561 .00568 .0	- 1249 - 1249 - 1256 - 1257 - 1201 -	2550 2801 2138 1538 0777 00147 001427 -0339 -04554 -07676 -12770 -1668 -23818 -27048	250 2350 2350 2350 2350 2637 27 20527 2020		2985 29235 15770 04158 01583 -0586 -0586 -05910 -1037 -13446 -1845 -1845 -226642 -29693 -33638	2068 2068 2068 2069 2010 2010 2010 2010 2010 2010 2010 201	2650 2650 2650 1245 .0043 .0091 0169 0664 0804 1027 1240 1350 1459 1566 1676 2375 2367 2367 2367 3664 23767 2367 2367	2511 1760 1095 0493 0060 0320 0571 0815 1050 1165 1278 1391 1501 1611 1718 1927 2132 2527 2132 2527 2719 3088 3446 34109 4425

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TABLE III - Continued

VALUES OF 
$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}$$
 FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[\mathbf{M}_{\mathbf{0}} = 0.50\right]$ 

			r					1				
K												
AH qo	0	0.01	0.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
10			<u> </u>									
0.800	0.4725	0.4679	0.4633	0.4588	0.4499	0.4289	0.4091	0.3905	0.3730	0.3566	0.3409	0.3261
.700	0.4725 .3878	.3832	.3786	•3741	-3652	•3440	.3243	.3056	.2881	.2716	·2559	.2411 .1681
.600	.3152	-3105	-3059	.3013	.2925	.2713	.2514	.2328	.2152	.1986	.1829	1001
.500	.2507	·2460	2414	.2369	.2280	.2069	.1869	.1681	.1505	•1339	.1182	.1033
.400	.1925	.1878	.1831	.1786	.1697	.1484	.1284	.1097	.0911	.0753	.0596	<b>أ</b> بليان.
•300	.1390	.1343	1297	.1251	.1162	-0949	.0749	.0561	0384	.0217	.0060	0090
.250	.1139	.1092	.1046	.1001	.0911	.0697	.0497	.0309	.0132	0035	0193	0343
.200	.0896	-0850	.0803	•0757	.0668	·0454	.0254	.0065	0112	0279 0516	0438 0675	0588 0825
.150	.0661	.0614	.0567	.0522	.0432	.0218 0010	.0017	0171 0399	0349 0577	0744	0903	1053
.100	·0434	-0387	.0340	0294	.0204		0211	0510	0688	0856	1015	1165
.075	.0323	.0276	.0229	.0184	.0093	0121	0322	0620	0797	0965	1124	1274
.050	.0214	.0167	.0121	-0075	0015	0230	0430	0728	0906	1073	1232	1383
.025	.0107	.0059	0013 0094	0033 0140	0123 0230	0338 0445	0539 0646	0835	1013	1181	1340	-1491
0 005	0 03.05	0047	0199	0245	0335	0550	0751	- 0000	1118	1286	1445	1596
025	0105 0208	0152 0255	0301	- 0347	0438	0652	0854	0940 1043	1221	1390	1549	1699
050	0310	0357	0403	- 0449	0540	0755	0956	1146	1324	1492	1651	1862
075 100	0412	0460	0506	0552	0643	0858	1059	- 1249	- 1627	1596	1755	+.1905
150	0608	0655	0702	0748	0838	1058	1256	1445	1623	1792	1952	2103
200	0799	0847	0894	0939	1030	1246	1448	1638	1816	1985	2ílili	<b>*.2296</b>
250	0986	1034	1081	1127	1217	1433	1635	1826	2004	2173	2333	- 2484
300	1170	1218	1265	1311	1401	1617	1820	2010	2189	2358	2518	2669
400	1524	1572	1619	1665	1756	1972	2175	2366	2545	2714	2874	5026
500	1863	1911	1958	2004	2095	2312	2515	2706	2886	3055	3216	3568
600	2188	2236	2283	2330	2421	-,2638	2842	3033	3213	3380	3544	3697
700	2501	2548	2596	2642	2734	2952	3156	3347	3528	3698	- • 3859	4012
800	2802	2851	2898	- 2945	3036	3254	3459	3651	3832	4002	4164	4317
		L	<u> </u>				L					

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values of 
$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}$$

FOR DETERMINING POINT DRAG COEFFICIENT - Continued

$$\left[\mathbf{M}_{o} = 0.55\right]$$

Δ <u>H</u> κ	0	0.01	C.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
0.800 .700 .600 .500 .400 .300 .250 .200 .150 .025 0 .025 0 .025 075 050 075 100 250 250 250 300 300 250 250 250 050 025 050 025 050 025 050 -	0.4572 .37551 .3051 .1862 .1346 .1101 .0862 .0639 .0419 .0206 .0103 0 0101 0300 0395 0772 09586 0772 1128 1468 1468 2106 2107 2696	0.4527 .37095 .3816 .1899 .108199 .108192 .03265 .01656 .00148 -03447 -04436 -0819 -11541 -18455 -1841 -22455	0.4481 .3664 .29536 .1771 .1253 .1009 .0776 .0326 .0219 .0114 .0010 0093 0194 0394 0489 0480 0865 1222 1562 1888 2501 2790	0.4437 .3619 .2914 .2291 .1726 .1208 .0963 .0725 .00500 .0281 .00738 0038 00339 01339 	0.4350 .35327 .2827 .2203 .1637 .1119 .0636 .0412 .00125 .00228 .0329 .0429 .0429 .0524 .0815 .1000 .1180 .1357 .1698 .203377 .26388 .2938	0.4141 .33237 .2617 .1992 .0908 .06634 .0200 0128 03411 05412 07437 15144 15741 1594 1594 15914 15914 15914 15914 15914 15914 15914 15914 15914 15915 2553 2855 3144	0.314216 .324216 .324216 .12216 .07655 .00202273370 .004421 .00784414799552 .00784414799552 .115771445578 .12245578	0.3762 .29437 .1043 .05279 .01407 .00567 .00407 .005150 .005224 .005228 .005229 .11225 .11217 .16034 .17684 .123631 .22645 .3248 .3539	0.3589 .2769 .2062 .1436 .0848 .01037 0583 0691 07901 1106 1206 1402 1781 1781 1962 2182 2182 31428 3718	0:34605 .26058 .12704 .07083 07085 07528 075458 075458 075458 075458 075458 112773 112773 112773 112773 112773 1226579 226579 23297 3389	0.3271 .2451 .1742 .1715 .0547 .0026 0460 0685 09015 11226 13229 14331 1528 1420 2107 22886 2107 21886 21810 31558 3458	0.3125 .2304 .1595 .0967 .0368 0368 0609 1164 1270 1375 1481 1870 1870 2257 2439 2961 3606 3606 3910 420

values of 
$$\left(\frac{\rho_2}{\rho_o}\right)^{1/2} - \left(\frac{q_2}{q_o}\right)^{1/2}$$
 for determining point drag coefficient - continued

 $\left[\mathbf{M}_{\mathbf{0}} = 0.60\right]$ 

K ΔH q <sub>o</sub>	0	0.01	0.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
0.800 .700 .600 .500 .400 .250 .250 .150 .025 .025 .025 .025 .025 .025 .025 .0	0.4425 2342 172624 17262 10836 00400 00199 001988 00199 0019880 001999 0019880 00199 0019880 00199 0019880 00199 0019880 00199 0019880 00199 0019880 00199 0019880 00199 0019880 00199 0019880 00199 0019880 001980 0019880 0019880 001980 001980 001980 001980 001980 001980 001980	0.4366 .3579 .28997 .172516 .075751 .107870 .0325553 .00145 .00145 .00145 .00762 .00762 .00762 .00762 .00762 .00762 .00762	0.4324 -32555 -225066 -127066 -127066 -127067 -127067 -127067 -12706 -12	0.4291 2811 2811 2811 2808 16661 00699 001664 001664 0003375 0013315 001621 001622 00162 00162 001622 00162 001622 001622 00162 001622 001622 001622 001622 001622 001622 001622 001622 001622	0.44240 44240 44240 44240 45778 66698 6000 6000 6000 6000 6000 6000 60	0.331313333333211572 9.911136659 9.91113862012 9.9113862012 9.911333333332110 9.9113111111111111111111111111111111111	0.379538 1176632 1176632 11766330 1176632 1000233333332 10002333333352 10002333353 112083552 112083552 11208552 11208552 11208552 11208552 11208552 11208552 11208552 11208552	22005748 62215388490037000095502870117365542 0000056722211208670117365542 00000000000000000000000000000000000	0.000 100 100 100 100 100 100 100 100 10	0.32488 1805990 0.014909190 0.00009190 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0000 0.0	0 16551 13551 104959 104990 10004 10004 10004 10004 10004 10009 10004 10	0.2980 .2190 .1505 .0898 .0347 -0623 -1058 -11655 -12666 -17657 -18566 -17657 -18566 -1857 -2217 -2291 -25630 -3208 -320

NACA ARR No. L5H27

VALUES OF  $\left(\frac{\rho_2}{\rho_0}\right)^{1/2}$  -  $\left(\frac{q_2}{q_0}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[\mathbf{M}_{0} = 0.65\right]$ 

$\frac{\overline{d}^{0}}{\nabla H}$	0	0.01	0.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
0.800 .700 .600 .500 .400 .250 .250 .075 .025 -0750 -0	0.4212 3488 .2834 .22729 .1248 .1021 .0803 .0588 .0191 .0095 0.0095 0187 02766 02712 08712 0	0.4197 .3443 .2788 .1602 .09758 .05442 .0145 .0046 .03412 .0145 .0046 .023212 .05555 .0758	0.4153 .3399 .27163 .1658 .0931 .0797 .01997 .0190 .0091 .0095 .0091 .03458 .0803 .09134553 .09134553 .12033 .22566	0.4110 .3566 .27119 .1595 .11886 .06697 .02553 .00555 .00400 .01359 .04102 .06778 .04102 .06778 .117908 .117970 .14998 .20348 .20348 .20348 .20348	0.4025 3271 2616 21509 1029 0582 03766 00328 003216 003210 04990 07930 012658 10268	0.3822 30411 1829 1082974 01622 -00142 -00142 -00567 -007974 -1799 -1799 -1146 -1799 -1146 -122351 -1469 -122351 -122351 -122351 -122351 -122351 -122351	0.326 2.263	0.34536 20458 109446 -002122 -0042122 -0042122 -0042122 -006195 -0099990 -1155296 -1185296 -1	0.32548 1272478945 12724789999999999999999999999999999999999	0.3124 .2368 .1722 .05110 .01120 01355589 0075559 1055447 115478 1202016 1202016 2251264 235365	0.2974 -15569 -00474 -002794 -002794 -002794 -111209 -1209 -11309 -11586 -1201577 -1201577 -1201577 -1201577 -1201577 -1201577 -1201577 -12015	0.2829 2070 14111 08295 -001910 -00640 -10662 -112573 -115472 -11742 -11742 -11742 -11742 -11742 -11742 -11742 -11742 -11742 -11742 -11742 -11742 -11742 -1174

VALUES OF 
$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}$$
 FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[\mathbf{u}_{0}=0.70\right]$ 

			•	_		LMo = 0	.70]					
$\frac{\Delta H}{q_0}$ K.	o	0.01	0.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
0.800 .700 .500 .200 .200 .200 .200 .200 .200 .2	0.4068 .3347 .2720 .2163 .1657 .0769	0.4024 -3303 -2676 -2119 -1613 -1724 -0724 -0724 -0724 -0724 -0724 -0724 -0724 -0736	0.3981 .3260 .2633 .20759 .1569 .0889 .0680 .0478 .0282 .0187 .0090 -02675 -0408 -0735 -0408 -0735 -1088 -1384 -1668 -1941 -12452	0.3938 .3217 .2590 .20326 .1563 .0846 .0636 .0438 .0044 .00314 .00311 .0399 .0455 .09730 .1128 .1986 .1128 .1986 .1138 .1986 .1997	0.3855 .3133 .2505 .19470 .0760 .0750 .0348 .00577 -001320 -0311 -0398 -0487 -05739 -1217 -1217 -1217 -12586	0.3655 -29304 -17457 -05545 -0545 -01434 -023327 -066779 -066779 -1270 -1270 -1270 -1270 -12846 -2798	0.3467 .2744 .1555 .10582 .0582 .0584 .0153 .0049 .0246 .03436 .0529 .0799 .0886 .0799 .0886 .0711 .0799 .0886 .0711 .0799 .0886 .09141 .1305 .1466 .1621 .2288 .27444 .2996	0.3290 .2566 .1936 .0866 .0402 .0183 .0028 .0231 .0618 .0711 .0894 .0981 .10697 .11524 .1805 .21063 .23668 .2368 .23183	0.3123 .2399 .1767 .0697 .0231 .0013 .0199 .0402 .0600 .0690 .0790 .0884 .0975 .1066 .1154 .1242 .1330 .1823 .1979 .22808 .1823 .1979 .2568 .28147 .3360	0.2966 .2240 .1608 .1047 .0537 .0070 -0149 -0564 -0762 -0857 -1047 -1138 -1229 -1317 -1405 -1495 -1826 -1987 -2144 -3010 -3527	0.2816 .2090 .1458 .0895 .0383 -0383 -0514 -0718 -0917 -1012 -1106 -1200 -1293 -1471 -1560 -1648 -1817 -2300 -2602 -2891 -3168	0.2674 1947 1314 0752 0227 - 0227 - 0447 - 0660 - 1062 - 1158 - 1253 - 13438 - 1530 - 1618 - 1707 - 17954 - 2291 - 2291 - 2291 - 235836

VALUES OF  $\left(\frac{\rho_2}{\rho_6}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[M_{o} = 0.75\right]$ 

$\frac{\Delta H}{\overline{q}_{O}}$	0	0.01	0 02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
0.800 .700 .600 .500 .400 .300 .250 .050 .025 0 .0550 .025 .0550 .025 .0550 .025 .0550 .025 .0500 .025 .025	0.3888 .3297 .2068 .1585 .1147 .0935 .0735 .0355 .0174 .0086 0.0170 .0086 01253 .0496 .0949 .1204 .1201 .1201 .1201 .1201	0.3815524 15524 15524 1108990 004910 004910 00000 00131980 0012198	0.3802 .3815 .1911 .1989 .10547 .06457 .00267 .00089 .00159 .00584 .00583 .0058	0.3760 370768 19553 10101 108043 10804 108043 10804 108043 10804 108043 108043 108043 108043 108043 108043 108043 108043 108043	0 23855 1379209 0 203855 1379209 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	0.31786 .1656 .1656 .11788 .0512 .0512 .001566 .001566 .0015976 .009751 .009758 .1022758 .10333 .103	0 326 26006828 320066828 0010532267 00000000000000000000000000000000000	0.3121 .1824 .1891 .0805 .0360 .0150 .0053 .0247 .0435 .0707 .0707 .0707 .070880 .0965 .1049 .1131 .1250 .1602 .1751 .2573 .2824 .3067	0.29566 .16247.10637.100192.004058.007877.40995.50.13202.4622.14622.4999.2222.27479.22222.27479.2222.27479.22222.27479.22222.27479.22222.27479.22222.27479.22222.27479.22222.27479.22222.27479.22222.27479.22222.27479.22222.27479.22222.27479.22222.27479.22222.27479.22222.27479.222222.27479.22222.274222.274222.274222.27422.27422.27422.27422.27422.27422.27422.27422.274222.27422	0.2801 .2109 .1500 .094782 .0179 .0357766 .09488 .10325 .1297 .1297 .13824 .1297	0.2650 19516 0.18128 0.0119 0.053329 09111 1127651 -	0.2513 18199 0.16785 0.0167788 0.016778 0.016778 0.016778 0.016778 0.016778 0.016778 0.0167788 0.016778 0.016778 0.016778 0.016778 0.016778 0.016778 0.0167788 0.016778 0.016778 0.016778 0.016778 0.016778 0.016778 0.0167788 0.016778 0.016778 0.016778 0.016778 0.016778 0.016778 0.0167788 0.016778 0.016778 0.016778 0.016778 0.016778 0.016778 0.0167788 0.016778 0.016778 0.016778 0.016778 0.016778 0.016778 0.0167788 0.016778 0.016778 0.016778 0.016778 0.016778 0.016778 0.0167788 0.016778 0.016778 0.016778 0.016778 0.016778 0.016778 0.0167788 0.016778 0.016778 0.016778 0.016778 0.016778 0.016778 0.0167788 0.016778 0.016778 0.016778 0.016778 0.016778 0.016778 0.0167788 0.016778 0.016778 0.016778 0.016778 0.016778 0.016778 0.0167788 0.016778 0.016778 0.016778 0.016778 0.016778 0.016778 0.0167

VALUES OF 
$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2}$$
 -  $\left(\frac{q_2}{q_0}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[M_{o} = 0.80\right]$ 

$\frac{\Delta H}{q_o}$	0	0.01	0.02	0.03	0.05	0:10	0.15	0.20	0.25	0.30	0.35	o•fto
0.800 .700 .600 .500 .100 .250 .150 .075 .055 -055 -150 -150 -250 -250 -300 -500 -700 -800	0.3706 .3053 .2483 .1973 .1091 .0893 .0702 .0516 .0352 .0166 .0082 0.0160 .0240 .0470 .0418 .0470 .0418 .0762 .0902 .1170 .1676 .1910 .1916	0.3664 .3010 .2440 .1930 .1470 .1047 .0849 .0458 .0429 .0208 .00138 .00126 .002862 .00365 .002862 .00565 .009417 .1956 .1956 .2181	0.3622 29988 14005 0.6159 0.00057 0.00087 0.00	0.3580 29356 1385 09764 05772 0389 00130 -00130 -00130 -00148 -0130 -00148 -0130 -00191 -0791 -0794 -1305 -1305 -13063 -12047	0.3500 2845 2764 1762 0879 0680 0430 0132 0047 0132 0047 0132 0047 0135 0047 0135 0047 0135 0047 0135 0047 0135 0047 0135 0047 0135 0047 0135 0047 0135 0047 0135 0047 0135 0047 0135 0047 0135 0047 0135 0047 0135 0047 0135 0047 0047 0047 0047 0047 0047 0047 004	0.3306 20767 1104 0680 01076 001076 001076 001076 001076 001049 001076 0	0.3124 21468 18982 18982 199994 100085 100085 100085 10008 1	0.2952 22955 17207 1727 1727 1727 1727 1727 1727 172	0.2789 21556 15542 15542 0577 01550 -002430 -00698 -007869 -007869 -009355 -1115729 -112729 -112729 -112729 -112729 -12889 -2889 -2889 -2889	0.2636 1977 1901 0886 0420 -0007 -0208 -0401 -0588 -0769 -1029 -1112 -1275 -1275 -1275 -1275 -1275 -1286 -23064 -2058 -23064 -2058 -23064 -230	0.2490 185538 1275777 100755519 100755519 100950 11	0.2351 .1691 .0597 .0130 -0299 -0500 -0694 -1052 -11258 -11258 -11408 -11491 -1577 -16731 -12038 -12038 -22603 -22603 -23587 -35587

VALUES OF 
$$\left(\frac{\rho_2}{\rho_o}\right)^{1/2} - \left(\frac{q_2}{q_o}\right)^{1/2}$$

FOR DETERMINING POINT DRAG COEFFICIENT - Continued

$$\left[\mathbf{M}_{\mathbf{o}} = 0.85\right]$$

М q <sub>o</sub>	0	0.01	0.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
0.800 -700 -600 -500 -1400 -250 -050 -050 -050 -075 -050 -075 -050 -075 -100 -1500 -1000 -	0.3521 .29361 .18378 .108467 .08467 .0932378 .004922 .0055729 .005729 .005729 .005729 .0057258	0.3486 183996 1839996 1839996 00427913541 001972461 001972461 0079570 1146359 11463590	0.3499 28179523 281795631 0.75523 0.75	0.3798 37736 127512 0.2753112 0.0753694 0.013508 0.001208	0.3369 226959 126959 12890 064581 0001328 0001	0.31564 1062886 106333 1064460 1000887 10033108 10009	9.61.428995226888762728193545700753319762004932626282 9.22.728.45700753319762004932626282 9.22.728.45700753319762004932626282 9.22.7289545576220049326262824	0 216129440788244429508268710078394	0.2619 1948 0.9516 0.00865 0.00865 - 0.0699 - 0.07861 - 0.09149 - 1091718 - 112494 - 112494 - 112494 - 112494 - 1225584 - 225584 - 2297	0.2474 1844 0.304616 - 0.00467 - 0.00677568 - 0.0097568 - 0.0097568 - 100978 - 1125976 - 1125976 - 1126977 - 1126977 - 1126977 - 112697 -	0 114519861191688998861748883320851141111111111111111111111111111111111	0.15570 1551

TABLE III - Continued

VALUES OF 
$$\left(\frac{\rho_2}{\rho_o}\right)^{1/2}$$
 -  $\left(\frac{q_2}{q_o}\right)^{1/2}$  FOR DETERMINING POINT DRAG COEFFICIENT - Continued

 $\left[\mathbf{M}_{\mathrm{O}} = 0.90\right]$ 

			[							i		
ΔH K	0	0.01	0.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
AH qo  0.800 .700 .600 .500 .100 .250 .100 .055 .055 .055 .055 .055 .055 .0	0 362 375299 173804455 1738054665 1738054665 1738054665 17380586665 17467 1747 174	0.01 0.3296 .2198 .17382 .17382 .07591 .0163 .01897 .001189 .005930 .01899 .0189990 .0189990 .0189990 .0189990 .0189990	0.02 0.3256 .2670 .2157 .1697 .1281 .0900 .07550 .0321 .0142 .0065 .00159 .0159 .0231 .0372 .0542 .0773 .08992 .0773 .18992 .13594	0.03 0.215 0.2216 0.2216 0.2216 0.2216 0.2216 0.2216 0.2216 0.2217 0.221	0.05 0.3137 .20377 .1168 .0779 .04259 .00459 .00534 .005352 .0499 .005355 .0497 .02855 .0497 .08924 .08924 .08924 .08924 .08924 .08924 .08924 .08924 .08924 .08924 .09	0.10  0.2950 .2950 .1847 .1386 .0585 .0434 .0097 .01754 .00550 .00479 .015522 .08864 .095622 .08864 .1095 .11467 .17921 .1913	0.15 0.2774 .2185 .1669 .1206 .0783 .0223 .0251 .0119 .0281 .05189 .0589 .0589 .06644 .0737 .08877 .08877 .08878 .11530 .11530 .115888 .115888 .12324	0.20 0.2606 .2017 .1500 .1036 .0618 .00292 .04514 .0688 .07638 .09814 .09854 .11326 .11326 .11326 .11326 .128689 .22803	0.25 1.346 1.13846 1.13845	0.30 0.2299 .1708 .1188 .0723 .03085 -0266 -0140 -0611 -0774 -0832 -1160 -1234 -11577 -11651 -11912 -21395 -2835	0.35 0.2157 .1565 .1044 .05776 .015776 .02314 .05879 .07522 .10818 .123081	0.40  0.2022 1428 0907 0441 0017 -0553 -0727 -0899 -1063 -1143 -1222 -1299 -1375 -1451 -1568 -1809 -1944 -2077 -2206 -2456 -2692 -2917 -3133

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VALUES OF 
$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}$$

FOR DETERMINING POINT DRAG COEFFICIENT - Continued

$$\left[\mathbf{M}_{\mathbf{o}} = 0.95\right]$$

K AH qo	0	0.01	0.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
0.800 .700 .600 .500 .400 .300 .250 .200 .025 .025 .025 .075 .100 .250 .250 .250 .250 .250 .250 .250 .2	0.3152 2603 2118 1685 1291 0762 0762 0215 00141 0070 0 0070 - 0138 - 0205 - 0270 - 05270 - 05270 - 05270 - 0768 - 1124 - 1622 - 1813	0.312 .2562 .25677 .16449 .08820 .075599 .075599 .012473 .01080 .0014180 .01449 .056911 .12665 .1856	0.3073 .2073 .2064 .1209 .0849 .0615 .0055	0.304 304838 196698 1110803747 1110803747 1000529657 10005296577 10005296577 11179 111794 111794 111794 111794 111794 111794	0 2950 2940 2940 2940 2940 2940 2940 2940 294	0.211481 722729039045822672272729903904582672888551299257023332 222729039045825702257023332 222729039045888551299257023332 100000000000000000000000000000000	0 2055	0.2435 18891 0.955941 0.05591 0.01499 0453091 0453091 066552350 06652350 0665230 0665230 0665230 066523	0.2279 1723 1723 1723 1723 1723 1723 1723 1723	0.2132 1575 10844 064433 -012952 -024621 -076526 -08526 -09970 -11211 -12796 -11211 -12796 -11211 -12796 -11211 -12796 -1	0 1941 0 1941 0 0500 0 0768 0 07623 0 07623 0 0998 1 114188 1 1477 1 1675 1 1202 1 120	89964858662852952952525252525252525252525252525252

## TABLE III - Concluded

VALUES OF 
$$\left(\frac{\rho_2}{\rho_0}\right)^{1/2} - \left(\frac{q_2}{q_0}\right)^{1/2}$$
 FOR DETERMINING POINT DRAG COEFFICIENT - Concluded

 $M_0 = 1.00$ 

$\frac{\Delta H}{\overline{Q}_{O}}$ K	o	0.01	0.02	0.03	0.05	0.10	0.15	0.20	0.25	0.30	0.35	0.40
0.800 .700 .600 .500 .400 .300 .250 .025 .050 .025 .050 050 150 150 250 300 400 500 600	0.2971 .2454 .1999 .1590 .1219 .0889 .0719 .0275 .0275 .02034 .0065 0130 0130 0376 0376 0948 0948 1344 1531 1711	0.2931 -2914 -1959 -19578 -067274 -053732 -001925 -00192918 -07684 -07684 -1138774 -1138774	0.2892 .23750 .1510 .1510 .1519 .0798 .0637 .0191 .00555 00827 0217 02779 04581 0697 08099 1626 1616 1796	0.2854 .1880 .1471 .10999 .07597 .0151 .0081 .00156 0188 0188 031790 05622 05622 0867 1478 1858	0.2778 .2261 .1804 .1394 .1022 .0681 .0519 .0325 .0002 -00675 .0267 -0267 -0333 -03459 -0580 -07818 -1359 -1148 -1359 -11920	0.2597 .25978 .16209 .08355 .04934 .01276 .012575 .0188 .05588 .05588 .05588 .05588 .05588 .056502 .0794 .11242 .11242 .11936 .211938	0.24 1906 1140607 1001596	0.1784 108794 0.128794 0.0153	0.112 15898 1171378 0017281 -0017281 -004629665 -0099350 -1016466777638 -1148108 -11	0.1966 .1943 .0981 .0985 .01587 .01523 .04737 .0919 .0919 .09052 .01052	0.1828 .180423 .08423 .09425 .09423 .09425 .099933 .12663 .12663 .12663 .12663 .1267 .1267 .178463 .178467 .178467 .19	0.1696 1171 .0706 .0289 .0090 .0438 .0603 .0760 .1060 .1131 .1202 .1271 .13406 .1473 .1539 .16036 .1851 .1971 .2086 .2520 .2720 .2915 .3101

TABLE IV  $F_{c} \quad \text{AS A FUNCTION OF } \quad \frac{H - p}{p}$ 

<u>н - р</u>	Fc	<u>н - р</u> р	Fc	н <b>-</b> р	Fc	<u>Н - р</u>	Fc	<u>H - p</u>	Fc	<u>н - р</u> р	F <sub>C</sub>	<u>H - p</u>	Fc
0.01 .023.045.0678.090.1123.145.678.900.102.090.1123.145.678.222.222.223.333.333.333.333.333.333.33	1.00356 1.001648 1.01791 1.01479 1.024836 1.024836 1.0355838882 1.035838882 1.04589 1.04589 1.0558588 1.06881936 1.06881936 1.077837 1.08914782 1.094782 1.094782 1.10488 1.10488 1.11698 1.11698 1.11699 1.11692 1.11692 1.11692	444444444555555555566666666666666666666	1.12311 1.12316 1.123456 1.12946 1.132555 1.132555 1.132555 1.132555 1.132555 1.132555 1.13255 1.14489 1.155380 1.155380 1.155380 1.155380 1.16829 1.16829 1.17498 1.1893775 1.18938 1.1893775 1.199575 1.20432 1.21587 1.221438 1.221438 1.221438 1.221438 1.221438	1.04 1.05 1.06 1.07 1.08	1.235.86 2.35.86 2.35.86 1.224.696 1.225.86 1.225.86 1.225.86 1.226.692.77 1.228.88 1.226.692.77 1.228.88 1.229.991.86 1.332.88	1.12 1.12 1.12 1.12 1.12 1.12 1.12 1.12	1.331636 1.333627 1.333627 1.333627 1.3349691 1.3349691 1.33491 1.3359693 1.336266 1.3369692 1.3369692 1.3377491 1.338995 1.38995 1.38995 1.39750 1.39750 1.39750 1.40798 1.40798 1.40798 1.41987	111111111111111111111111111111111111111	1.42230 1.42478 1.42721 1.42964 1.43209 1.43452 1.43452 1.43697 1.43939 1.444909 1.45151 1.45390 1.45629 1.46352 1.463	1.82 8.84 8.85 8.89 9.99 9.99 9.99 9.99 9.99 9.99	1.5061 1.5061 1.5061 1.51061 1.51061 1.51061 1.51061 1.5510995 1.5510995 1.5510995 1.5510995 1.5510995 1.5510995 1.5510995 1.5510995 1.5510995 1.5510995 1.55109995 1.55109995 1.55109995 1.55109995 1.55109995 1.55109995 1.551099995 1.551099995 1.55109999999999999999999999999999999999	2.16 16 118 19 10 12 12 12 12 12 12 12 12 12 12 12 12 12	1.5844 1.58708 1.558708 1.55890298 1.5599675 1.5599675 1.6605554 1.6605774 1.6605774 1.6659150 1.66591400 1.6712309 1.7723337 1.77533839 1.775338399 1.775338399 1.775338399

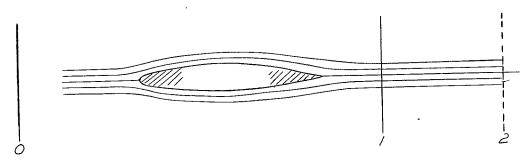
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TABLE V M AS A FUNCTION OF p/H

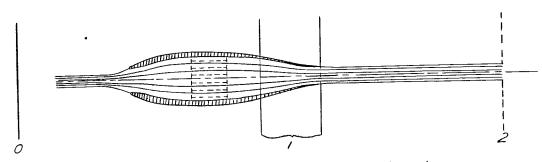
M	H\q	М	p/H	M	p/H	
0 003450675 0125 1125 115675 12012245678 9012	1 9999891405440602344307157845551923555 999999999999999999999999999999999	25 3333333344445678901224567890122456 6666666666666666666666666666666666	0.991 991 991 991 991 991 991 991 991 991	7890122456789012245678901224567890 666777777777788888888888899999999999999	0 0 0 0 0 0 0 0 0 0 0 0 0 0	

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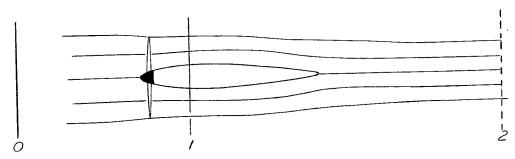


(a) For measurement of profile drag.



(b) For measurement of internal drag or thrust.

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(c) For measurement of propeller thrust.

Figure 1. - Station designations.

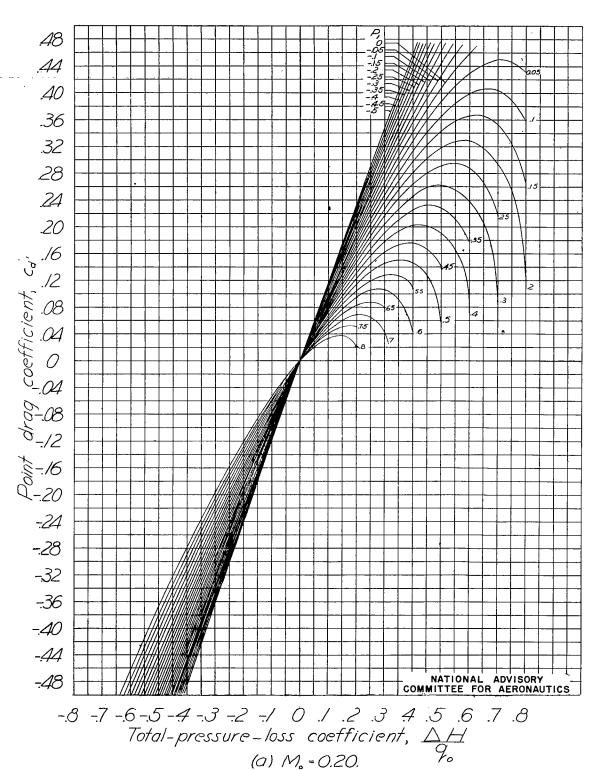
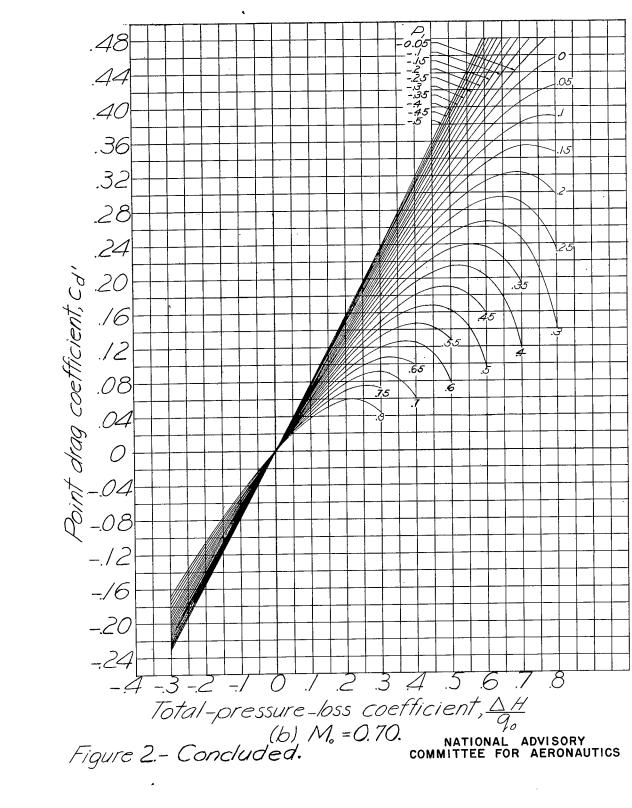


Figure 2.—Point drag coefficient from table I for isoenergic flow.



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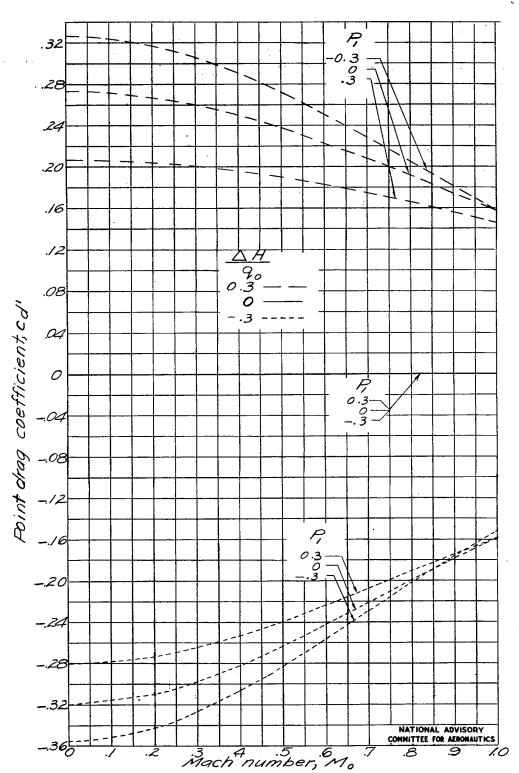


Figure 3. - Variation of point drag coefficient with Mach number for isoenergic flow.

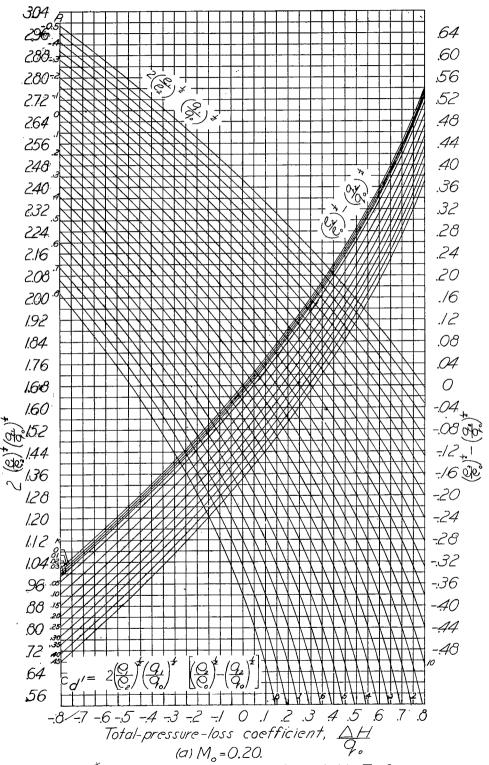


Figure 4.—Point drag coefficient from Table II for
flows wherein energy is added. NATIONAL ADVISORY
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